NASA MEMO 2-24-59L

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MEMORANDUM

BASIC PRESSURE MEASUREMENTS AT TRANSONIC SPEEDS ON A THIN 45° SWEPTBACK HIGHLY TAPERED WING WITH SYSTEMATIC SPANWISE TWIST VARIATIONS

WING WITH QUADRATIC SPANWISE TWIST VARIATION

By John P. Mugler, Jr.

Langley Research Center
Langley Field, Va.

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

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SUMMARY

Pressure distributions obtained in the Langley 8-foot transonic pressure tunnel on a thin highly tapered twisted 45° sweptback wing-body combination are presented. The wing has a quadratic spanwise twist variation from 0° at 10 percent of the semispan to 6° at the tip. The tip is at a lower angle of attack than the root. Tests were made at stagnation pressures of both 0.5 and 1.0 atmosphere at Mach numbers from 0.800 to 1.200 through an angle-of-attack range from -4° to 20°.

INTRODUCTION

A research program is currently being conducted at the Langley Research Center to determine the loads due to wing twist at transonic and supersonic speeds. As part of this program, tests have been made in the Langley 8-foot transonic pressure tunnel on four wings; an untwisted wing to serve as a reference, and wings with linear, quadratic, and cubic variations of twist across the span. References 1 and 2 present the basic pressure measurements on the untwisted and linearly twisted wings, respectively. The present paper presents the basic pressure measurements on the wing with a quadratic variation of twist across the span. These data are being presented without analysis.

SYMBOLS

b	wing span
$b'/2$	unsupported semispan (distance from outer face of wing mounting block to tip)

c airfoil section chord, measured parallel to plane of symmetry

$\bar{c}$ wing mean aerodynamic chord

c_m wing section pitching-moment coefficient about 0.25c,

$$\int_0^1 \left(C_{p,L} - C_{p,U}\right)\left(0.25 - \frac{x}{c}\right)d\left(\frac{x}{c}\right)$$

c_n wing section normal-force coefficient, $\int_0^1 \left(C_{p,L} - C_{p,U}\right)d\left(\frac{x}{c}\right)$

C_p pressure coefficient

$C_{p,sonic}$ pressure coefficient corresponding to local Mach number of 1.0

D diameter

l body length

M Mach number

q free-stream dynamic pressure

R Reynolds number based on $\bar{c}$

x distance measured from leading edge of wing or from nose of body, positive rearward

y spanwise distance measured from body center line

y' spanwise distance measured from outer face of wing mounting block

$\frac{\partial \Delta\alpha}{\partial n}$ wing-twist influence coefficient due to normal load at c/4 point

$\frac{\partial \Delta\alpha}{\partial m}$ wing-twist influence coefficient due to moment about c/4 point

α angle of attack of wing-body center line

$\Delta\alpha$ angle of attack of wing station minus angle of attack of wing-body center line

ϕ built-in twist angle

Subscripts:

L lower surface

U upper surface

APPARATUS

Tunnel

The investigation was conducted in the Langley 8-foot transonic pressure tunnel. The test section of this facility is rectangular in cross section. The upper and lower walls are slotted longitudinally to allow continuous operation through the transonic speed range with negligible effects of choking and blockage. During this investigation, the tunnel was operated at stagnation pressures of approximately 0.5 and 1.0 atmosphere. The dewpoint of the tunnel air was controlled and kept constant at approximately 0° F. The stagnation temperature of the tunnel air was automatically controlled and was kept constant and uniform across the tunnel at 123° F. Control of both dewpoint and stagnation temperature in this manner minimized humidity effects. Details of the test section have been presented in reference 3.

Models

The wing tested has the same plan form, thickness, and camber distribution as the untwisted wing described in reference 1. However, the wing of the present investigation had twist built into each wing panel from 10 percent of the semispan to the tip. The sections were twisted about their leading edge in planes parallel to the model plane of symmetry with the trailing edges up. Therefore, the tips are at a lower angle of attack than the wing-body center line. The twist varied quadratically from 0° at the 10-percent-semispan station to 6° at the tip. The wing was constructed of steel and was tested as a midwing configuration. The wing was tested in combination with the basic body of reference 1. Details of the wing-body combination are presented in figure 1 and the wing twist characteristics are presented in table I.

TESTS

The wing-body combination was tested at Mach numbers from 0.800 to 1.200 at tunnel stagnation pressures of 0.5 and 1.0 atmosphere. At the stagnation pressure of 0.5 atmosphere, the angle-of-attack range extended from -4° to 20°. At the stagnation pressure of 1.0 atmosphere, the angle-of-attack range extended from -4° to 20° only at the Mach numbers of 0.800, 0.900, and 0.940. For the remainder of the test Mach numbers at this staganation pressure, the angle-of-attack range extended from -4° to 12°.

Transition strips were fixed on the model during all the tests. The strips were about 0.10 inch wide and were formed by sprinkling No. 120 carborundum grains on a plastic adhesive. The strips extended from the wing-body juncture to the wing tip at 10 percent of the local chord on the upper and lower wing surfaces and formed a ring around the body at 10 percent of the body length.

The Reynolds number based on the wing mean aerodynamic chord varied over the Mach number range from about 1.3×10^6 to 1.5×10^6 during tests at 0.5 atmosphere and from about 2.6×10^6 to 2.9×10^6 during tests at 1.0 atmosphere. (See fig. 2.)

MEASUREMENTS AND ACCURACY

Measurements of the local static pressures on the model were made using flush-mounted orifices distributed over the upper and lower wing surfaces and along longitudinal body rows. Figure 3 shows the location of the six stations on the wing and the five rows on the body where the orifices were located. Pressure coefficients determined from these measurements are estimated to be accurate within ± 0.006.

The angle of attack of the model was measured with a strain-gage attitude transmitter mounted in the nose of the model and is estimated to be accurate within $\pm 0.1^\circ$. Calibrations of the test section of the Langley 8-foot transonic pressure tunnel indicate that local deviations from the average free-stream Mach number are of the order of ± 0.005 at subsonic speeds. With increases in Mach number, these deviations increased but did not exceed ± 0.010 in the region of the wing at $M = 1.2$. Several representative Mach number distributions at the center of the test section have been presented in reference 3. The average stream Mach number was held to within ± 0.003 of the nominal values shown in the figures.

The stagnation pressures of 1,058 and 2,116 pounds per square foot have been designated 0.5 and 1.0 atmosphere, respectively, throughout this paper. The stagnation pressure was generally held to within ±10 pounds per square foot during tests at 0.5 atmosphere and to within ±20 pounds per square foot during tests at 1.0 atmosphere.

Influence coefficients were obtained for this wing from a static calibration and are presented in table II. Wing-twist angles, computed by using the experimental wing section data in conjunction with the influence coefficients of table II, are estimated to be accurate to within one-quarter of a degree.

RESULTS

The pressure coefficients for the wing in the presence of the body are presented in tables III and IV for stagnation pressures of 0.5 and 1.0 atmosphere, respectively. Pressure coefficients for the body in the presence of the wing are presented in tables V and VI for stagnation pressures of 0.5 and 1.0 atmosphere, respectively. The values of the free-stream dynamic pressure shown in the tables is the average value over the angle-of-attack range. The pressure coefficients have been plotted to show the pressure-coefficient distributions over the surfaces and are presented in figure 4 for the wing in the presence of the body and in figure 5 for the body in the presence of the wing. The distributions over the wing (fig. 4) have been numerically integrated for section normal-force and section pitching-moment coefficients about 0.25c and the results are presented in table VII. The section data were used in conjunction with the influence coefficients of table II to calculate the change in angle at several wing stations and these values are also presented in table VII.

In figures 4 and 5, data have been presented for both stagnation pressures in the same figure. Fixing transition during the tests tended to minimize the effects of Reynolds number on the pressure coefficients. This fact is evident from figures 4 and 5, which show that in all cases changing the stagnation pressure from 0.5 to 1.0 atmosphere had no significant effects on the pressure coefficients over the body or over the inboard wing stations. Aeroelastic effects caused the wing to twist over the outboard regions. The results in table VII show that the outboard wing sections are generally operating at a lesser angle of attack at a stagnation pressure of 1.0 atmosphere than at 0.5 atmosphere due to the differences in dynamic pressure. Therefore, the differences in the pressure distributions over the outboard wing sections at the two

different stagnation pressures in figure 4 should be attributed to the differences in local angle of attack and not to Reynolds number effects.

Langley Research Center,
National Aeronautics and Space Administration,
Langley Field, Va., December 9, 1958.

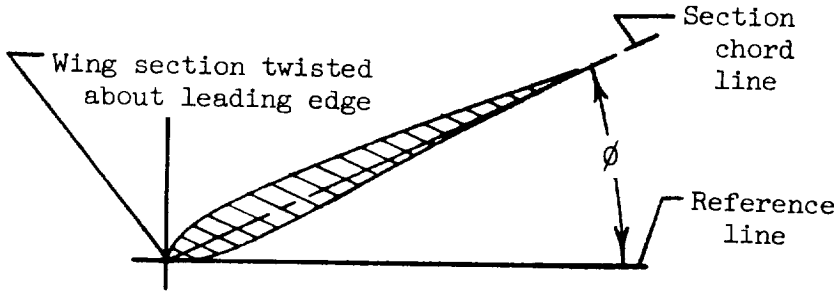
REFERENCES

1. Mugler, John P., Jr.: Basic Pressure Measurements at Transonic Speeds on a Thin 45° Sweptback Highly Tapered Wing With Systematic Spanwise Twist Variations - Untwisted Wing. NASA MEMO 10-20-58L, 1958.

2. Mugler, John P., Jr.: Basic Pressure Measurements at Transonic Speeds on a Thin 45° Sweptback Highly Tapered Wing With Systematic Spanwise Twist Variations - Wing With Linear Spanwise Twist Variation. NASA MEMO 12-28-58L, 1959.

3. Mugler, John P., Jr.: Transonic Wind-Tunnel Investigation of the Aerodynamic Loading Characteristics of a 60° Delta Wing in the Presence of a Body With and Without Indentation. NACA RM L55G11, 1955.

TABLE I.- WING TWIST CHARACTERISTICS



Typical Section

$\frac{y}{b/2}$	ϕ, deg
0	0
.10	0
.12	.003
.25	.167
.40	.667
.60	1.852
.80	3.630
.95	5.352
1.00	6.000

TABLE II.- WING DEFLECTION CHARACTERISTICS

Twist measurement station, $\frac{y}{b/2}$	Rate of change in twist angle due to a load at section quarter chord, $\frac{\partial \Delta\alpha}{\partial n}$, deg/lb, at -				
	$\frac{y'}{b'/2} = 0.185$	$\frac{y'}{b'/2} = 0.348$	$\frac{y'}{b'/2} = 0.565$	$\frac{y'}{b'/2} = 0.795$	$\frac{y'}{b'/2} = 0.948$
0.25	0	-0.0002	-0.0007	-0.0010	-0.0006
.40	0	-.0002	-.0018	-.0029	-.0044
.60	0	-.0001	-.0014	-.0108	-.0175
.80	0	-.0001	-.0010	-.0116	-.0481
.95	0	-.0001	-.0008	-.0108	-.0640

Twist measurement station, $\frac{y}{b/2}$	Rate of change in twist angle due to a pitching moment about section quarter chord, $\frac{\partial \Delta\alpha}{\partial m}$, deg/in-lb, at -				
	$\frac{y'}{b'/2} = 0.185$	$\frac{y'}{b'/2} = 0.348$	$\frac{y'}{b'/2} = 0.565$	$\frac{y'}{b'/2} = 0.795$	$\frac{y'}{b'/2} = 0.948$
0.25	0.0001	0.0001	0.0001	0.0001	0.0005
.40	.0002	.0003	.0002	.0004	.0008
.60	.0002	.0006	.0024	.0027	.0021
.80	.0002	.0007	.0037	.0155	.0122
.95	.0002	.0007	.0037	.0245	.0970

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE

OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY

(a) 12-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.000	.197	.407	.515	.525	.507	.282	-.042	-.342	-.579	.000
	.032	.316	.208	.098	-.015	-.157	-.472	-.810	-1.553	-1.337	.032
	.078	.198	.105	.020	-.063	-.165	-.356	-.606	-.879	-1.240	.078
	.150	.099	.018	-.060	-.137	-.227	-.404	-.593	-.771	-1.102	.150
	.250	.018	-.054	-.127	-.191	-.272	-.444	-.612	-.605	-.901	.250
	.350	-.001	-.070	-.132	-.188	-.256	-.411	-.553	-.588	-.766	.350
	.450	-.048	-.111	-.173	-.220	-.291	-.433	-.521	-.613	-.701	.450
	.550	-.084	-.140	-.196	-.246	-.307	-.443	-.466	-.629	-.666	.550
	.650	-.057	-.108	-.155	-.195	-.248	-.338	-.387	-.554	-.605	.650
	.760	-.050	-.093	-.129	-.157	-.193	-.254	-.347	-.485	-.568	.760
	.840	-.052	-.083	-.110	-.127	-.153	-.189	-.271	-.424	-.566	.840
	.926	-.029	-.050	-.066	-.077	-.090	-.109	-.169	-.317	-.516	.926
Lower surface	.035	-.373	-.262	-.109	.022	.146	.355	.535	.695	.813	.035
	.082										.082
	.150	-.236	-.154	-.075	-.000	.078	.219	.354	.434	.574	.150
	.250										.250
	.350	-.222	-.159	-.091	-.033	.029	.144	.249	.357	.428	.350
	.450	-.240	-.183	-.119	-.062	-.006	.099	.197	.287	.364	.450
	.550	-.228	-.177	-.121	-.068	-.018	.075	.162	.245	.315	.550
	.650	-.205	-.161	-.110	-.064	-.019	.061	.135	.201	.259	.650
	.750	-.138	-.108	-.070	-.031	.004	.074	.130	.179	.220	.750
	.850	-.090	-.069	-.038	-.010	.020	.075	.105	.135	.156	.850
	.900	-.065	-.050	-.029	-.005	.020	.060	.077	.095	.101	.900
	M = 0.900; q = 358 lb/sq ft										
Upper surface	.000	.318	.467	.548	.560	.544	.406	.164	-.073	-.299	.000
	.032	.326	.221	.122	.013	-.109	-.390	-.733	-1.297	-1.286	.032
	.078	.210	.123	.038	-.041	-.122	-.281	-.471	-.771	-1.231	.078
	.150	.104	.022	-.049	-.130	-.209	-.354	-.496	-.631	-.908	.150
	.250	.017	-.060	-.134	-.203	-.279	-.392	-.525	-.632	-.709	.250
	.350	-.008	-.078	-.143	-.206	-.266	-.397	-.521	-.615	-.668	.350
	.450	-.065	-.133	-.197	-.258	-.316	-.435	-.552	-.470	-.683	.450
	.550	-.115	-.180	-.259	-.335	-.392	-.500	-.608	-.539	-.692	.550
	.650	-.086	-.140	-.203	-.289	-.381	-.487	-.582	-.568	-.619	.650
	.760	-.073	-.121	-.169	-.221	-.345	-.497	-.513	-.587	-.601	.760
	.840	-.077	-.110	-.143	-.174	-.211	-.508	-.426	-.620	-.631	.840
	.926	-.050	-.069	-.088	-.101	-.100	-.217	-.240	-.512	-.577	.926
Lower surface	.035	-.348	-.235	-.094	.034	.146	.359	.550	.712	.839	.035
	.082										.082
	.150	-.226	-.156	-.072	-.004	.074	.231	.373	.501	.603	.150
	.250										.250
	.350	-.247	-.181	-.107	-.047	.015	.146	.265	.375	.465	.350
	.450	-.271	-.221	-.149	-.089	-.025	.099	.205	.306	.395	.450
	.550	-.325	-.246	-.167	-.108	-.045	.068	.166	.259	.346	.550
	.650	-.331	-.226	-.153	-.098	-.048	.053	.134	.216	.296	.650
	.750	-.208	-.145	-.094	-.053	-.013	.068	.127	.194	.262	.750
	.850	-.118	-.091	-.056	-.025	.007	.065	.105	.145	.206	.850
	.900	-.081	-.066	-.043	-.016	.008	.051	.075	.104	.153	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 368 lb/sq ft										
Upper surface	.000	.373	.492	.569	.581	.569	.451	.229	.010	-.221	.000
	.032	.336	.239	.136	.036	-.079	-.343	-.658	-1.238	-1.383	.032
	.078	.218	.139	.056	-.016	-.092	-.252	-.404	-.689	-1.357	.078
	.150	.110	.036	-.037	-.111	-.182	-.313	-.442	-.556	-.719	.150
	.250	.016	-.052	-.128	-.192	-.246	-.361	-.477	-.574	-.560	.250
	.350	-.010	-.074	-.138	-.194	-.259	-.375	-.486	-.581	-.557	.350
	.450	-.072	-.135	-.198	-.247	-.299	-.409	-.509	-.601	-.638	.450
	.550	-.132	-.209	-.277	-.324	-.374	-.470	-.573	-.629	-.696	.550
	.650	-.095	-.159	-.261	-.316	-.368	-.472	-.562	-.455	-.692	.650
	.760	-.087	-.143	-.236	-.317	-.374	-.481	-.550	-.505	-.640	.760
	.840	-.092	-.128	-.179	-.330	-.403	-.531	-.548	-.535	-.671	.840
	.926	-.057	-.074	-.084	-.161	-.306	-.486	-.436	-.526	-.625	.926
Lower surface	.035	-.310	-.220	-.073	.036	.149	.358	.555	.722	.857	.035
	.082										.082
	.150	-.207	-.146	-.068	-.005	.076	.227	.375	.513	.625	.150
	.250										.250
	.350	-.240	-.186	-.106	-.057	.013	.138	.263	.382	.487	.350
	.450	-.268	-.211	-.155	-.102	-.034	.086	.201	.318	.420	.450
	.550	-.316	-.272	-.204	-.135	-.057	.055	.168	.276	.368	.550
	.650	-.338	-.302	-.207	-.127	-.064	.035	.132	.231	.318	.650
	.750	-.317	-.258	-.121	-.074	-.031	.051	.128	.212	.286	.750
	.850	-.285	-.140	-.067	-.037	-.007	.043	.104	.167	.239	.850
	.900	-.198	-.082	-.044	-.025	-.008	.022	.070	.127	.186	.900
	M = 0.980; q = 385 lb/sq ft										
Upper surface	.000	.426	.536	.595	.609	.594	.497	.295	.095	-.127	.000
	.032	.358	.261	.164	.068	-.039	-.292	-.596	-1.090	-1.230	.032
	.078	.244	.163	.088	.019	-.053	-.202	-.352	-.567	-1.115	.078
	.150	.129	.057	-.010	-.079	-.147	-.261	-.381	-.477	-.574	.150
	.250	.034	-.037	-.109	-.159	-.213	-.318	-.416	-.510	-.607	.250
	.350	.006	-.055	-.115	-.179	-.226	-.327	-.429	-.518	-.624	.350
	.450	-.061	-.117	-.170	-.219	-.272	-.363	-.453	-.542	-.649	.450
	.550	-.147	-.202	-.247	-.293	-.337	-.422	-.516	-.600	-.699	.550
	.650	-.119	-.191	-.245	-.292	-.342	-.432	-.514	-.609	-.689	.650
	.760	-.116	-.198	-.258	-.308	-.354	-.443	-.528	-.621	-.493	.760
	.840	-.168	-.242	-.298	-.349	-.401	-.495	-.575	-.658	-.472	.840
	.926	-.165	-.231	-.293	-.334	-.383	-.478	-.559	-.602	-.596	.926
Lower surface	.035	-.269	-.165	-.043	.070	.176	.384	.576	.740	.885	.035
	.082										.082
	.150	-.165	-.106	-.043	.025	.094	.251	.403	.535	.653	.150
	.250										.250
	.350	-.221	-.162	-.092	-.038	.022	.161	.290	.412	.516	.350
	.450	-.246	-.187	-.128	-.083	-.033	.101	.229	.344	.450	.450
	.550	-.296	-.239	-.193	-.141	-.079	.064	.189	.304	.404	.550
	.650	-.324	-.279	-.222	-.175	-.104	.040	.159	.262	.356	.650
	.750	-.312	-.262	-.208	-.151	-.070	.053	.151	.243	.326	.750
	.850	-.307	-.248	-.195	-.136	-.053	.041	.124	.205	.279	.850
	.900	-.315	-.257	-.205	-.149	-.067	.015	.091	.164	.232	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.000	.495	.588	.638	.647	.642	.561	.372	.176	-.031	.000
	.032	.402	.307	.213	.117	.017	-.203	-.507	-1.006	-1.111	.032
	.078	.295	.214	.138	.073	.008	-.117	-.281	-.507	-1.011	.078
	.150	.183	.112	.043	-.022	-.085	-.176	-.295	-.387	-.476	.150
	.250	.086	.016	-.052	-.100	-.149	-.235	-.333	-.421	-.511	.250
	.350	.064	.007	-.065	-.118	-.165	-.247	-.346	-.433	-.527	.350
	.450	-.001	-.053	-.117	-.161	-.207	-.279	-.377	-.458	-.551	.450
	.550	-.085	-.133	-.185	-.226	-.268	-.339	-.430	-.510	-.602	.550
	.650	-.069	-.124	-.185	-.230	-.276	-.347	-.433	-.521	-.607	.650
	.760	-.074	-.130	-.197	-.242	-.286	-.364	-.449	-.534	-.604	.760
	.840	-.120	-.178	-.242	-.290	-.333	-.414	-.495	-.575	-.651	.840
	.926	-.118	-.170	-.232	-.274	-.317	-.401	-.484	-.568	-.632	.926
Lower surface	.035	-.188	-.106	.009	.115	.218	.431	.625	.788	.922	.035
	.082										.082
	.150	-.095	-.047	.011	.070	.139	.301	.452	.589	.697	.150
	.250										.250
	.350	-.147	-.111	-.047	.004	.075	.207	.342	.463	.566	.350
	.450	-.169	-.135	-.074	-.025	.024	.154	.281	.401	.501	.450
	.550	-.220	-.180	-.130	-.086	-.026	.116	.243	.358	.462	.550
	.650	-.247	-.216	-.161	-.119	-.055	.087	.210	.321	.412	.650
	.750	-.235	-.203	-.153	-.103	-.028	.103	.207	.305	.386	.750
	.850	-.234	-.189	-.139	-.086	-.021	.091	.181	.261	.342	.850
	.900	-.241	-.197	-.150	-.098	-.034	.071	.149	.225	.295	.900
	M = 1.125; q = 421 lb/sq ft										
Upper surface	.000	.529	.582	.623	.628	.614	.581	.495	.451	.139	.000
	.032	.355	.259	.178	.075	-.025	-.249	-.471	-.858	-.920	.032
	.078	.270	.193	.135	.060	-.007	-.156	-.282	-.461	-.843	.078
	.150	.176	.111	.047	-.034	-.080	-.174	-.258	-.321	-.362	.150
	.250	.081	.024	-.016	-.076	-.126	-.206	-.279	-.337	-.378	.250
	.350	.071	.009	-.038	-.084	-.141	-.227	-.287	-.351	-.402	.350
	.450	.018	-.033	-.077	-.123	-.171	-.243	-.310	-.372	-.423	.450
	.550	-.045	-.097	-.131	-.177	-.223	-.291	-.364	-.416	-.466	.550
	.650	-.036	-.089	-.133	-.184	-.230	-.301	-.356	-.425	-.470	.650
	.760	-.053	-.105	-.147	-.197	-.242	-.307	-.370	-.431	-.463	.760
	.840	-.103	-.157	-.193	-.244	-.284	-.349	-.405	-.470	-.502	.840
	.926	-.081	-.138	-.174	-.224	-.268	-.337	-.399	-.463	-.494	.926
Lower surface	.035	-.209	-.117	-.008	.068	.170	.356	.574	.818	.985	.035
	.082										.082
	.150	-.091	-.040	.022	.072	.134	.272	.441	.614	.753	.150
	.250										.250
	.350	-.127	-.085	-.019	.007	.068	.208	.345	.504	.631	.350
	.450	-.144	-.101	-.036	-.000	.059	.162	.294	.446	.575	.450
	.550	-.185	-.151	-.087	-.050	.001	.117	.258	.413	.537	.550
	.650	-.209	-.174	-.116	-.079	-.031	.095	.233	.382	.498	.650
	.750	-.201	-.165	-.102	-.065	-.016	.122	.245	.374	.475	.750
	.850	-.183	-.146	-.086	-.050	.003	.125	.234	.343	.438	.850
	.900	-.189	-.151	-.095	-.056	-.004	.108	.209	.314	.395	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 436 lb/sq ft										
Upper surface	.000	.557	.605	.636	.641	.634	.602	.509	.429	.496	.000
	.032	.344	.256	.164	.079	-.023	-.219	-.424	-.701	-.753	.032
	.078	.268	.200	.133	.061	.011	-.148	-.273	-.380	-.738	.078
	.150	.165	.098	.040	.000	-.060	-.138	-.237	-.298	-.379	.150
	.250	.117	.061	-.003	-.064	-.098	-.178	-.248	-.296	-.339	.250
	.350	.096	.044	-.006	-.045	-.100	-.185	-.251	-.307	-.359	.350
	.450	.055	.004	-.040	-.077	-.123	-.199	-.270	-.332	-.380	.450
	.550	-.001	-.051	-.094	-.135	-.175	-.242	-.308	-.367	-.421	.550
	.650	-.001	-.055	-.105	-.144	-.184	-.250	-.317	-.373	-.435	.650
	.760	-.025	-.069	-.113	-.148	-.184	-.254	-.325	-.376	-.443	.760
	.840	-.084	-.128	-.167	-.205	-.236	-.302	-.364	-.416	-.476	.840
	.926	-.068	-.115	-.158	-.188	-.228	-.299	-.357	-.421	-.478	.926
Lower surface	.035	-.187	-.100	-.008	.081	.175	.348	.528	.702	.876	.035
	.082										.082
	.150	-.064	-.020	.038	.094	.137	.259	.425	.581	.732	.150
	.250										.250
	.350	-.093	-.052	-.008	.039	.095	.221	.353	.476	.605	.350
	.450	-.107	-.062	-.016	.032	.082	.192	.302	.424	.550	.450
	.550	-.149	-.109	-.070	-.023	.028	.145	.256	.390	.516	.550
	.650	-.175	-.132	-.090	-.044	.013	.110	.223	.355	.478	.650
	.750	-.167	-.125	-.085	-.037	.010	.111	.236	.362	.467	.750
	.850	-.145	-.114	-.083	-.038	.006	.113	.245	.347	.436	.850
	.900	-.162	-.118	-.088	-.055	-.013	.099	.232	.321	.400	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.000	-.380	-.009	.323	.417	.364	-.163	-.674	-1.041	-1.169	.000
	.025	.313	.212	.075	-.102	-.349	-1.053	-1.359	-1.264	-1.092	.025
	.076	.160	.070	-.050	-.174	-.333	-.630	-1.118	-1.259	-1.111	.076
	.150										.150
	.250	.002	-.075	-.164	-.244	-.335	-.546	-.900	-1.183	-1.029	.250
	.350	-.050	-.120	-.197	-.262	-.346	-.528	-.689	-1.064	-.987	.350
	.450	-.074	-.138	-.198	-.261	-.326	-.477	-.503	-.939	-.916	.450
	.550	-.070	-.121	-.174	-.220	-.278	-.384	-.449	-.785	-.845	.550
	.650	-.065	-.111	-.151	-.188	-.234	-.303	-.410	-.663	-.783	.650
	.750	-.047	-.078	-.118	-.132	-.167	-.209	-.321	-.580	-.747	.750
	.850	-.018	-.043	-.064	-.075	-.091	-.119	-.220	-.477	-.704	.850
	.926	.011	-.002	-.007	-.011	-.022	-.035	-.117	-.373	-.650	.926
Lower surface	.023	-.658	-.512	-.219	-.004	.170	.385	.490	.558	.595	.023
	.072	-.516	-.296	-.124	-.016	.102	.277	.402	.510	.579	.072
	.150	-.431	-.232	-.096	-.011	.073	.219	.337	.442	.520	.150
	.250	-.302	-.200	-.109	-.035	.036	.164	.268	.369	.444	.250
	.350	-.252	-.182	-.109	-.043	.018	.133	.231	.316	.386	.350
	.450	-.225	-.174	-.111	-.059	.003	.099	.185	.264	.324	.450
	.550	-.191	-.146	-.096	-.047	.002	.087	.160	.225	.279	.550
	.650	-.143	-.110	-.070	-.030	.012	.087	.139	.185	.230	.650
	.750	-.085	-.063	-.031	.003	.033	.092	.129	.153	.183	.750
	.860	-.029	-.019	-.001	.023	.052	.085	.101	.090	.093	.860
	.900	-.006	.003	.015	.036	.052	.079	.079	.045	.025	.900
	M = 0.900; q = 358 lb/sq ft										
Upper surface	.000	-.250	.072	.331	.417	.394	.057	-.384	-.754	-.842	.000
	.025	.306	.207	.083	-.079	-.291	-.939	-1.208	-1.203	-.932	.025
	.076	.159	.060	-.052	-.171	-.313	-.549	-1.003	-1.183	-.958	.076
	.150										.150
	.250	-.013	-.092	-.180	-.260	-.344	-.521	-.711	-1.010	-.969	.250
	.350	-.069	-.149	-.231	-.317	-.376	-.536	-.681	-.938	-.942	.350
	.450	-.102	-.170	-.256	-.372	-.442	-.577	-.661	-.874	-.893	.450
	.550	-.094	-.160	-.224	-.301	-.417	-.558	-.622	-.796	-.843	.550
	.650	-.091	-.143	-.200	-.263	-.417	-.570	-.590	-.715	-.794	.650
	.750	-.065	-.107	-.144	-.184	-.216	-.524	-.521	-.666	-.755	.750
	.850	-.035	-.060	-.082	-.098	-.092	-.189	-.361	-.608	-.726	.850
	.926	.005	-.010	-.016	-.020	-.012	-.055	-.218	-.522	-.681	.926
Lower surface	.023	-.673	-.506	-.243	-.024	.147	.375	.503	.583	.638	.023
	.072	-.512	-.286	-.141	-.030	.083	.271	.412	.522	.605	.072
	.150	-.404	-.250	-.117	-.029	.056	.218	.348	.455	.547	.150
	.250	-.360	-.238	-.136	-.058	.016	.157	.277	.387	.473	.250
	.350	-.328	-.230	-.143	-.071	-.004	.125	.233	.333	.420	.350
	.450	-.338	-.239	-.154	-.086	-.025	.091	.187	.282	.361	.450
	.550	-.325	-.204	-.132	-.076	-.021	.077	.159	.241	.316	.550
	.650	-.220	-.152	-.097	-.049	-.008	.077	.136	.206	.269	.650
	.750	-.114	-.084	-.050	-.013	.020	.083	.124	.171	.224	.750
	.860	-.040	-.028	-.008	.018	.039	.076	.087	.110	.145	.860
	.900	-.008	-.004	.010	.033	.048	.065	.055	.054	.082	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 368 lb/sq ft										
Upper surface	.000	-.184	.095	.341	.426	.411	.133	-.281	-.648	-.900	.000
	.025	.304	.211	.090	-.058	-.258	-.782	-1.086	-1.213	-1.169	.025
	.076	.158	.070	-.041	-.151	-.289	-.508	-.873	-1.192	-1.142	.076
	.150										.150
	.250	-.016	-.088	-.175	-.242	-.339	-.494	-.642	-1.190	-1.069	.250
	.350	-.080	-.155	-.235	-.292	-.363	-.499	-.642	-.952	-1.044	.350
	.450	-.114	-.200	-.307	-.364	-.418	-.550	-.662	-.784	-.968	.450
	.550	-.112	-.182	-.276	-.343	-.407	-.536	-.630	-.717	-.888	.550
	.650	-.109	-.172	-.282	-.368	-.432	-.552	-.616	-.683	-.826	.650
	.750	-.083	-.122	-.180	-.331	-.404	-.522	-.571	-.667	-.766	.750
	.850	-.045	-.065	-.079	-.153	-.349	-.488	-.503	-.629	-.743	.850
	.926	.004	-.004	-.008	-.000	-.075	-.200	-.288	-.528	-.669	.926
Lower surface	.023	-.624	-.512	-.243	-.040	.138	.370	.509	.597	.655	.023
	.072	-.460	-.302	-.136	-.044	.073	.261	.409	.531	.628	.072
	.150	-.366	-.259	-.119	-.040	.050	.211	.346	.465	.571	.150
	.250	-.342	-.247	-.146	-.077	.006	.145	.277	.394	.495	.250
	.350	-.321	-.241	-.160	-.092	-.015	.113	.232	.344	.442	.350
	.450	-.331	-.274	-.196	-.119	-.042	.074	.182	.290	.385	.450
	.550	-.333	-.287	-.183	-.107	-.039	.059	.159	.253	.338	.550
	.650	-.332	-.280	-.126	-.076	-.023	.052	.134	.218	.295	.650
	.750	-.295	-.133	-.057	-.026	.004	.061	.120	.190	.255	.750
	.860	-.089	-.028	-.006	.011	.025	.042	.080	.131	.180	.860
	.900	-.017	.005	.015	.031	.033	.021	.048	.075	.122	.900
	M = 0.980; q = 385 lb/sq ft										
Upper surface	.000	-.111	.159	.363	.442	.436	.209	-.194	-.534	-.801	.000
	.025	.319	.225	.114	-.023	-.209	-.701	-1.019	-1.166	-1.206	.025
	.076	.173	.082	-.018	-.120	-.247	-.476	-.827	-1.077	-1.182	.076
	.150										.150
	.250	-.004	-.075	-.146	-.232	-.306	-.447	-.576	-1.079	-1.215	.250
	.350	-.076	-.143	-.207	-.269	-.333	-.452	-.580	-.996	-1.209	.350
	.450	-.137	-.217	-.281	-.334	-.388	-.504	-.610	-.759	-1.130	.450
	.550	-.130	-.205	-.269	-.322	-.382	-.495	-.611	-.582	-.796	.550
	.650	-.156	-.239	-.301	-.355	-.410	-.513	-.617	-.625	-.435	.650
	.750	-.147	-.220	-.289	-.332	-.386	-.486	-.597	-.571	-.474	.750
	.850	-.148	-.225	-.286	-.339	-.389	-.484	-.587	-.475	-.665	.850
	.926	-.129	-.195	-.241	-.282	-.334	-.388	-.452	-.417	-.657	.926
Lower surface	.023	-.589	-.462	-.222	-.020	.145	.38[illegible]	.532	.620	.681	.023
	.072	-.426	-.237	-.119	-.025	.086	.28[illegible]	.435	.553	.654	.072
	.150	-.324	-.213	-.110	-.023	.058	.23[illegible]	.368	.488	.599	.150
	.250	-.319	-.216	-.123	-.061	.011	.16[illegible]	.296	.417	.526	.250
	.350	-.304	-.221	-.138	-.083	-.024	.12[illegible]	.255	.371	.477	.350
	.450	-.317	-.251	-.182	-.136	-.065	.08[illegible]	.208	.314	.419	.450
	.550	-.324	-.262	-.204	-.156	-.086	.06[illegible]	.179	.283	.376	.550
	.650	-.332	-.275	-.215	-.163	-.076	.05[illegible]	.156	.250	.336	.650
	.750	-.309	-.255	-.198	-.138	-.045	.05[illegible]	.142	.224	.300	.750
	.860	-.293	-.246	-.189	-.124	-.038	.03[illegible]	.097	.167	.227	.860
	.900	-.267	-.220	-.168	-.100	-.048	.00[illegible]	.058	.117	.173	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.000	-.014	.220	.404	.482	.481	.290	-.093	-.436	-.682	.000
	.025	.365	.273	.166	.037	-.135	-.583	-1.013	-1.047	-1.081	.025
	.076	.220	.135	.037	-.059	-.177	-.394	-.890	-.991	-1.078	.076
	.150										.150
	.250	.054	-.017	-.101	-.171	-.241	-.359	-.476	-.968	-1.079	.250
	.350	-.018	-.072	-.149	-.207	-.267	-.367	-.485	-.927	-1.079	.350
	.450	-.085	-.147	-.212	-.268	-.319	-.413	-.516	-.750	-1.049	.450
	.550	-.079	-.138	-.210	-.259	-.312	-.412	-.517	-.520	-.778	.550
	.650	-.113	-.170	-.239	-.294	-.338	-.426	-.529	-.560	-.669	.650
	.750	-.105	-.153	-.226	-.270	-.317	-.404	-.511	-.558	-.627	.750
	.850	-.111	-.156	-.224	-.277	-.318	-.408	-.512	-.565	-.574	.850
	.926	-.111	-.148	-.213	-.256	-.293	-.340	-.465	-.509	-.354	.926
Lower surface	.023	-.479	-.394	-.168	.025	.189	.428	.577	.666	.732	.023
	.072	-.322	-.190	-.066	.017	.127	.327	.482	.606	.701	.072
	.150	-.235	-.152	-.061	.022	.107	.276	.421	.540	.646	.150
	.250	-.235	-.160	-.074	-.015	.060	.212	.352	.470	.575	.250
	.350	-.226	-.169	-.093	-.029	.033	.172	.305	.424	.527	.350
	.450	-.238	-.191	-.127	-.073	-.014	.129	.258	.375	.472	.450
	.550	-.242	-.203	-.145	-.096	-.030	.107	.231	.340	.433	.550
	.650	-.248	-.213	-.157	-.107	-.040	.099	.208	.309	.395	.650
	.750	-.232	-.192	-.141	-.088	-.020	.106	.197	.281	.358	.750
	.860	-.217	-.184	-.138	-.085	-.011	.085	.157	.233	.291	.860
	.900	-.215	-.177	-.131	-.075	-.011	.063	.117	.186	.240	.900
	M = 1.125; q = 421 lb/sq ft										
Upper surface	.000	.071	.277	.440	.498	.501	.362	.069	-.268	-.506	.000
	.025	.369	.284	.197	.052	-.094	-.519	-.792	-.877	-.900	.025
	.076	.231	.152	.070	-.030	-.136	-.411	-.743	-.861	-.892	.076
	.150										.150
	.250	.072	-.004	-.063	-.133	-.202	-.302	-.378	-.696	-.884	.250
	.350	.013	-.051	-.105	-.161	-.221	-.314	-.384	-.629	-.869	.350
	.450	-.049	-.109	-.158	-.215	-.267	-.351	-.421	-.633	-.871	.450
	.550	-.050	-.113	-.159	-.215	-.271	-.348	-.428	-.576	-.666	.550
	.650	-.075	-.132	-.176	-.232	-.283	-.371	-.439	-.542	-.540	.650
	.750	-.064	-.116	-.161	-.209	-.263	-.347	-.420	-.513	-.496	.750
	.850	-.073	-.123	-.167	-.216	-.264	-.346	-.412	-.484	-.413	.850
	.926	-.078	-.129	-.163	-.213	-.257	-.304	-.389	-.417	-.198	.926
Lower surface	.023	-.462	-.372	-.137	.020	.187	.407	.571	.704	.795	.023
	.072	-.274	-.156	-.047	.029	.132	.312	.479	.641	.764	.072
	.150	-.176	-.104	-.015	.044	.113	.266	.420	.579	.713	.150
	.250	-.185	-.116	-.027	.032	.086	.220	.362	.516	.646	.250
	.350	-.175	-.116	-.053	-.003	.061	.190	.325	.473	.601	.350
	.450	-.186	-.140	-.076	-.038	.021	.147	.283	.436	.552	.450
	.550	-.199	-.157	-.093	-.052	.008	.124	.260	.404	.519	.550
	.650	-.204	-.163	-.103	-.064	-.006	.115	.247	.384	.484	.650
	.750	-.193	-.153	-.091	-.049	.005	.129	.250	.363	.457	.750
	.860	-.176	-.144	-.085	-.050	.000	.125	.221	.324	.393	.860
	.900	-.185	-.147	-.093	-.054	.000	.116	.195	.282	.347	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 436 lb/sq ft										
Upper surface	.000	.182	.344	.466	.532	.529	.423	.180	-.117	-.362	.000
	.025	.374	.296	.199	.080	-.049	-.439	-.670	-.779	-.753	.025
	.076	.242	.164	.077	-.005	-.101	-.353	-.615	-.742	-.723	.076
	.150										.150
	.250	.081	.018	-.044	-.108	-.157	-.258	-.335	-.550	-.753	.250
	.350	.037	-.017	-.072	-.120	-.179	-.281	-.347	-.452	-.748	.350
	.450	-.021	-.075	-.133	-.178	-.232	-.310	-.371	-.467	-.778	.450
	.550	-.029	-.084	-.144	-.192	-.242	-.309	-.380	-.478	-.687	.550
	.650	-.061	-.109	-.159	-.198	-.241	-.326	-.400	-.468	-.535	.650
	.750	-.045	-.092	-.144	-.179	-.223	-.302	-.377	-.456	-.469	.750
	.850	-.054	-.102	-.152	-.186	-.230	-.307	-.377	-.463	-.474	.850
	.926	-.058	-.099	-.147	-.185	-.223	-.290	-.356	-.445	-.467	.926
Lower surface	.023	-.454	-.336	-.114	.060	.185	.400	.563	.679	.782	.023
	.072	-.244	-.105	-.035	.043	.144	.315	.469	.605	.737	.072
	.150	-.154	-.083	-.017	.050	.123	.269	.418	.547	.687	.150
	.250	-.154	-.092	-.030	.037	.106	.224	.360	.485	.621	.250
	.350	-.149	-.085	-.032	.016	.071	.198	.325	.446	.577	.350
	.450	-.166	-.114	-.063	-.015	.039	.162	.279	.407	.535	.450
	.550	-.183	-.134	-.086	-.033	.028	.140	.249	.381	.503	.550
	.650	-.183	-.134	-.091	-.041	.010	.118	.235	.369	.476	.650
	.750	-.158	-.123	-.087	-.035	.017	.123	.260	.364	.458	.750
	.860	-.140	-.100	-.059	-.012	.028	.124	.254	.340	.410	.860
	.900	-.144	-.104	-.064	-.019	.028	.128	.241	.310	.374	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.000	.206	.257	.333	.267	-.419	-1.193	-1.295	-.934	-.872	.000
	.020	.360	.276	.128	-.093	-.458	-1.188	-1.280	-.940	-.864	.020
	.073	.185	.077	-.054	-.212	-.430	-1.096	-1.209	-.971	-.857	.073
	.150	.098	.008	-.105	-.212	-.370	-.993	-1.136	-.979	-.833	.150
	.250	.030	-.048	-.142	-.233	-.349	-.711	-1.103	-.922	-.813	.250
	.350	-.019	-.088	-.168	-.245	-.340	-.497	-.993	-.875	-.794	.350
	.440	-.041	-.107	-.173	-.237	-.320	-.421	-.939	-.845	-.772	.440
	.550	-.049	-.102	-.156	-.207	-.271	-.344	-.807	-.811	-.757	.550
	.650	-.051	-.094	-.138	-.172	-.222	-.270	-.702	-.769	-.743	.650
	.750	-.039	-.075	-.103	-.127	-.160	-.192	-.586	-.731	-.723	.750
	.850	-.012	-.039	-.058	-.071	-.092	-.112	-.453	-.689	-.707	.850
	.923	.030	.009	.002	-.002	-.014	-.033	-.304	-.636	-.681	.923
Lower surface	.024	-.745	-.619	-.361	-.038	.187	.385	.471	.510	.526	.024
	.073	-.683	-.430	-.153	-.025	.108	.284	.396	.475	.525	.073
	.150	-.723	-.351	-.138	-.038	.065	.211	.317	.403	.469	.150
	.250	-.629	-.251	-.124	-.043	.037	.162	.259	.336	.395	.250
	.350	-.267	-.189	-.107	-.040	.029	.138	.219	.291	.348	.350
	.450	-.140	-.154	-.091	-.029	.029	.119	.186	.246	.293	.450
	.550	-.099	-.108	-.060	-.012	.035	.110	.162	.208	.247	.550
	.650	-.069	-.068	-.028	.012	.046	.109	.144	.167	.197	.650
	.750	-.020	-.022	.009	.041	.068	.116	.129	.130	.141	.750
	.850	.024	.018	.034	.061	.080	.112	.101	.064	.048	.850
	.900	.036	.033	.042	.064	.082	.103	.076	.000	-.018	.900
	M = 0.900; q = 358 lb/sq ft										
Upper surface	.000	.220	.274	.333	.282	-.197	-1.231	-1.196	-.785	-.897	.000
	.020	.342	.252	.121	-.087	-.408	-1.265	-1.203	-.810	-.934	.020
	.073	.162	.057	-.073	-.235	-.426	-.977	-1.166	-.819	-.849	.073
	.150	.076	-.015	-.133	-.259	-.397	-.716	-1.091	-.848	-.836	.150
	.250	.009	-.077	-.181	-.290	-.417	-.643	-1.069	-.858	-.824	.250
	.350	-.042	-.121	-.210	-.338	-.422	-.631	-1.004	-.848	-.802	.350
	.440	-.066	-.138	-.221	-.313	-.476	-.638	-.896	-.821	-.780	.440
	.550	-.072	-.131	-.199	-.279	-.465	-.652	-.817	-.795	-.769	.550
	.650	-.069	-.119	-.170	-.220	-.260	-.632	-.740	-.766	-.758	.650
	.750	-.053	-.093	-.125	-.159	-.154	-.303	-.670	-.745	-.739	.750
	.850	-.023	-.046	-.069	-.086	-.085	-.108	-.610	-.714	-.726	.850
	.923	.023	.008	.000	-.006	-.005	-.022	-.515	-.683	-.701	.923
Lower surface	.024	-.805	-.621	-.417	-.075	.144	.370	.473	.534	.560	.024
	.073	-.678	-.467	-.183	-.055	.077	.267	.389	.486	.552	.073
	.150	-.647	-.356	-.182	-.067	.037	.194	.314	.417	.493	.150
	.250	-.691	-.336	-.174	-.077	.009	.143	.249	.345	.425	.250
	.350	-.482	-.268	-.143	-.069	.002	.124	.212	.303	.377	.350
	.450	-.333	-.216	-.125	-.057	.005	.104	.183	.257	.328	.450
	.550	-.205	-.151	-.084	-.028	.018	.099	.159	.221	.281	.550
	.650	-.120	-.090	-.046	-.001	.036	.099	.138	.181	.237	.650
	.750	-.048	-.032	-.001	.035	.064	.110	.119	.149	.193	.750
	.850	.006	.014	.035	.060	.080	.106	.083	.086	.114	.850
	.900	.024	.032	.048	.067	.083	.097	.050	.033	.055	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 368 lb/sq ft										
Upper surface	.000	.238	.280	.338	.302	-.090	-1.050	-1.130	-1.070	-.861	.000
	.020	.330	.251	.120	-.058	-.352	-1.061	-1.135	-1.030	-.891	.020
	.073	.151	.051	-.075	-.216	-.401	-.992	-1.129	-.983	-.882	.073
	.150	.065	-.028	-.142	-.243	-.379	-.610	-1.014	-.916	-.902	.150
	.250	-.006	-.094	-.199	-.300	-.395	-.586	-.970	-.852	-.893	.250
	.350	-.060	-.134	-.265	-.325	-.416	-.594	-.956	-.806	-.860	.350
	.440	-.082	-.162	-.286	-.376	-.455	-.609	-.918	-.794	-.836	.440
	.550	-.088	-.155	-.262	-.388	-.467	-.615	-.819	-.805	-.830	.550
	.650	-.086	-.131	-.189	-.392	-.473	-.627	-.758	-.796	-.819	.650
	.750	-.064	-.095	-.122	-.227	-.462	-.614	-.693	-.773	-.798	.750
	.850	-.024	-.044	-.066	-.047	-.161	-.383	-.609	-.756	-.782	.850
	.923	.028	.014	.006	.020	-.002	-.147	-.506	-.718	-.754	.923
Lower surface	.024	-.763	-.634	-.421	-.113	.124	.355	.472	.546	.580	.024
	.073	-.612	-.493	-.186	-.085	.058	.248	.384	.495	.569	.073
	.150	-.547	-.375	-.204	-.102	.014	.178	.309	.424	.512	.150
	.250	-.533	-.372	-.226	-.111	-.012	.127	.248	.354	.445	.250
	.350	-.469	-.346	-.184	-.101	-.016	.102	.209	.312	.396	.350
	.450	-.416	-.344	-.159	-.090	-.016	.080	.175	.268	.348	.450
	.550	-.372	-.270	-.094	-.051	.001	.074	.153	.235	.308	.550
	.650	-.305	-.111	-.047	-.011	.023	.075	.135	.195	.263	.650
	.750	-.115	-.028	.002	.030	.054	.081	.117	.167	.219	.750
	.850	.003	.023	.038	.063	.076	.074	.084	.108	.149	.850
	.900	.032	.040	.052	.073	.083	.061	.054	.062	.095	.900
	M = 0.980; q = 385 lb/sq ft										
Upper surface	.000	.260	.295	.356	.336	.026	-1.020	-1.080	-1.161	-1.197	.000
	.020	.336	.257	.144	-.022	-.288	-1.009	-1.103	-1.178	-1.181	.020
	.073	.150	.053	-.052	-.191	-.356	-.904	-1.027	-1.130	-1.161	.073
	.150	.060	-.026	-.119	-.219	-.359	-.538	-.913	-1.141	-1.120	.150
	.250	-.024	-.101	-.187	-.268	-.368	-.551	-.866	-1.136	-1.069	.250
	.350	-.077	-.172	-.236	-.310	-.392	-.540	-.851	-1.148	-1.031	.350
	.440	-.115	-.204	-.286	-.354	-.422	-.562	-.842	-1.158	-.983	.440
	.550	-.140	-.227	-.310	-.371	-.442	-.569	-.821	-1.156	-.961	.550
	.650	-.157	-.240	-.323	-.385	-.454	-.584	-.774	-1.123	-.928	.650
	.750	-.166	-.256	-.324	-.390	-.452	-.576	-.727	-.870	-.876	.750
	.850	-.128	-.222	-.308	-.371	-.435	-.554	-.669	-.802	-.854	.850
	.923	-.021	-.074	-.132	-.180	-.283	-.374	-.446	-.742	-.816	.923
Lower surface	.024	-.745	-.591	-.394	-.113	.107	.363	.494	.564	.605	.024
	.073	-.592	-.454	-.180	-.079	.052	.259	.402	.512	.598	.073
	.150	-.516	-.321	-.176	-.098	-.002	.189	.326	.444	.541	.150
	.250	-.447	-.322	-.215	-.137	-.045	.131	.265	.380	.480	.250
	.350	-.434	-.303	-.205	-.141	-.059	.104	.229	.337	.434	.350
	.450	-.418	-.319	-.232	-.169	-.065	.078	.194	.298	.387	.450
	.550	-.384	-.304	-.219	-.149	-.052	.071	.174	.266	.347	.550
	.650	-.359	-.279	-.205	-.128	-.031	.061	.152	.234	.306	.650
	.750	-.311	-.240	-.166	-.085	-.009	.061	.134	.208	.265	.750
	.850	-.220	-.165	-.088	-.007	.007	.036	.094	.164	.200	.850
	.900	-.139	-.089	-.032	.016	.006	.013	.065	.125	.144	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.000	.318	.352	.398	.389	.118	-.834	-.975	-1.054	-1.094	.000
	.020	.381	.308	.194	.046	-.205	-.832	-.993	-1.068	-1.102	.020
	.073	.202	.113	-.001	-.125	-.282	-.760	-.940	-1.024	-1.089	.073
	.150	.116	.033	-.066	-.156	-.279	-.453	-.815	-1.032	-1.095	.150
	.250	.029	-.033	-.127	-.204	-.295	-.457	-.741	-1.042	-1.115	.250
	.350	-.038	-.104	-.175	-.242	-.323	-.453	-.702	-1.046	-1.118	.350
	.440	-.067	-.139	-.220	-.282	-.350	-.471	-.667	-1.045	-1.112	.440
	.550	-.102	-.161	-.240	-.303	-.369	-.482	-.640	-1.057	-1.106	.550
	.650	-.111	-.176	-.254	-.316	-.381	-.496	-.629	-1.058	-1.079	.650
	.750	-.132	-.188	-.261	-.324	-.381	-.494	-.619	-1.031	-1.033	.750
	.850	-.119	-.174	-.253	-.308	-.366	-.474	-.602	-.936	-.746	.850
	.923	-.086	-.121	-.181	-.230	-.292	-.346	-.437	-.876	-.775	.923
Lower surface	.024	-.637	-.537	-.352	-.063	.152	.403	.537	.613	.654	.024
	.073	-.477	-.391	-.139	-.027	.103	.304	.448	.562	.647	.073
	.150	-.400	-.264	-.121	-.039	.052	.229	.375	.497	.589	.150
	.250	-.332	-.253	-.149	-.080	.004	.178	.314	.428	.527	.250
	.350	-.330	-.239	-.149	-.084	-.006	.152	.279	.392	.486	.350
	.450	-.326	-.252	-.175	-.113	-.027	.126	.246	.352	.442	.450
	.550	-.306	-.240	-.168	-.100	-.018	.119	.223	.320	.404	.550
	.650	-.287	-.221	-.152	-.085	-.010	.115	.202	.291	.366	.650
	.750	-.242	-.183	-.118	-.051	.020	.116	.187	.263	.330	.750
	.850	-.192	-.141	-.079	-.021	.035	.094	.148	.214	.269	.850
	.900	-.148	-.104	-.052	-.005	.029	.068	.112	.176	.224	.900
	M = 1.125; q = 421 lb/sq ft										
Upper surface	.000	.358	.379	.434	.429	.230	-.665	-.894	-.896	-.914	.000
	.020	.389	.308	.222	.097	-.128	-.681	-.885	-.911	-.920	.020
	.073	.212	.122	.035	-.079	-.221	-.630	-.867	-.889	-.916	.073
	.150	.133	.055	-.028	-.115	-.228	-.524	-.821	-.864	-.915	.150
	.250	.058	-.012	-.081	-.162	-.253	-.385	-.759	-.871	-.922	.250
	.350	.000	-.064	-.121	-.197	-.274	-.405	-.662	-.865	-.915	.350
	.440	-.040	-.108	-.161	-.226	-.292	-.414	-.522	-.858	-.914	.440
	.550	-.066	-.132	-.187	-.246	-.309	-.406	-.482	-.873	-.913	.550
	.650	-.083	-.149	-.201	-.267	-.329	-.427	-.492	-.866	-.889	.650
	.750	-.099	-.160	-.210	-.274	-.334	-.434	-.495	-.843	-.841	.750
	.850	-.093	-.155	-.201	-.262	-.321	-.409	-.480	-.748	-.634	.850
	.923	-.063	-.125	-.173	-.231	-.283	-.343	-.405	-.669	-.615	.923
Lower surface	.024	-.613	-.545	-.331	-.082	.151	.398	.543	.656	.726	.024
	.073	-.430	-.355	-.119	-.019	.117	.299	.459	.604	.716	.073
	.150	-.349	-.220	-.085	-.015	.085	.234	.385	.546	.666	.150
	.250	-.278	-.207	-.092	-.035	.030	.178	.333	.485	.608	.250
	.350	-.248	-.187	-.092	-.048	.019	.164	.307	.458	.573	.350
	.450	-.261	-.193	-.117	-.070	-.008	.141	.280	.424	.532	.450
	.550	-.259	-.193	-.118	-.068	-.003	.143	.268	.402	.498	.550
	.650	-.247	-.181	-.106	-.056	.010	.146	.256	.371	.463	.650
	.750	-.205	-.150	-.073	-.023	.037	.162	.252	.352	.429	.750
	.850	-.161	-.107	-.041	.001	.060	.158	.223	.312	.373	.850
	.900	-.140	-.090	-.029	.012	.065	.141	.195	.275	.328	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Concluded

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.200; q = 436 lb/sq ft										
Upper surface	.000	.405	.432	.481	.467	.304	-.479	-.760	-.792	-.759	.000
	.020	.420	.351	.264	.124	-.056	-.541	-.747	-.793	-.746	.020
	.073	.242	.165	.073	-.040	-.168	-.482	-.723	-.828	-.762	.073
	.150	.157	.086	.003	-.083	-.180	-.430	-.691	-.782	-.766	.150
	.250	.083	.017	-.051	-.116	-.195	-.366	-.649	-.724	-.763	.250
	.350	.031	-.032	-.098	-.150	-.222	-.340	-.630	-.681	-.763	.350
	.440	-.012	-.070	-.131	-.180	-.248	-.364	-.587	-.637	-.770	.440
	.550	-.045	-.101	-.153	-.207	-.255	-.364	-.422	-.609	-.775	.550
	.650	-.069	-.122	-.186	-.230	-.281	-.367	-.418	-.614	-.769	.650
	.750	-.085	-.136	-.188	-.239	-.293	-.372	-.437	-.599	-.780	.750
	.850	-.082	-.128	-.183	-.226	-.272	-.359	-.436	-.588	-.765	.850
	.923	-.065	-.111	-.158	-.196	-.239	-.318	-.397	-.467	-.731	.923
Lower surface	.024	-.581	-.521	-.295	-.047	.160	.422	.553	.647	.721	.024
	.073	-.399	-.318	-.108	.019	.140	.327	.457	.582	.697	.073
	.150	-.301	-.162	-.061	.023	.105	.260	.383	.518	.640	.150
	.250	-.224	-.154	-.077	-.001	.071	.189	.322	.459	.585	.250
	.350	-.203	-.144	-.075	-.016	.046	.172	.296	.433	.553	.350
	.450	-.223	-.157	-.096	-.043	.014	.133	.267	.403	.514	.450
	.550	-.233	-.171	-.111	-.053	.008	.127	.271	.386	.488	.550
	.650	-.212	-.158	-.110	-.051	.003	.119	.273	.370	.460	.650
	.750	-.177	-.116	-.073	-.024	.022	.151	.278	.360	.433	.750
	.850	-.142	-.084	-.032	.017	.067	.187	.267	.326	.389	.850
	.900	-.130	-.074	-.027	.026	.074	.181	.245	.298	.352	.900

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.000	.116	.336	.476	.365	-.323	-.981	-.781	-.821	-.787	.000
	.025	.392	.313	.149	-.112	-.676	-.949	-.773	-.711	-.725	.025
	.075	.260	.164	.032	-.163	-.406	-.929	-.744	-.686	-.689	.075
	.150	.153	.060	-.060	-.205	-.371	-.909	-.768	-.686	-.699	.150
	.250	.093	.010	-.091	-.187	-.323	-.896	-.756	-.669	-.687	.250
	.350	.047	-.030	-.109	-.194	-.303	-.878	-.735	-.649	-.675	.350
	.450										.450
	.550	-.011	-.061	-.128	-.179	-.248	-.717	-.672	-.624	-.664	.550
	.650	-.027	-.074	-.120	-.165	-.212	-.575	-.626	-.605	-.645	.650
	.750	-.024	-.059	-.095	-.123	-.161	-.377	-.595	-.586	-.630	.750
	.850	-.004	-.033	-.056	-.070	-.095	-.194	-.541	-.568	-.620	.850
	.900	.015	-.013	-.032	-.040	-.055	-.120	-.533	-.561	-.607	.900
	.925										.925
Lower surface	.040	-.642	-.553	-.255	-.006	.196	.366	.436	.469	.484	.040
	.091	-.604	-.518	-.169	-.022	.130	.284	.367	.417	.453	.091
	.150	-.574	-.519	-.128	-.029	.105	.238	.313	.372	.410	.150
	.250	-.607	-.482	-.100	-.010	.082	.192	.257	.308	.346	.250
	.360	-.618	-.281	-.080	-.003	.070	.164	.208	.250	.292	.360
	.450	-.555	-.117	-.053	.006	.065	.143	.178	.207	.237	.450
	.550	-.419	-.043	-.032	.020	.068	.128	.143	.161	.183	.550
	.650	-.210	-.006	.001	.042	.078	.128	.118	.118	.134	.650
	.800	.057	.037	.039	.069	.091	.114	.054	.028	.033	.800
	.874	.100	.049	.043	.069	.086	.096	-.009	-.047	-.047	.874
	M = 0.900; q = 358 lb/sq ft										
Upper surface	.000	.124	.342	.462	.354	-.174	-1.277	-.871	-.846	-.837	.000
	.025	.381	.295	.148	-.155	-.824	-1.400	-.800	-.764	-.756	.025
	.075	.249	.146	.009	-.211	-.460	-1.326	-.726	-.708	-.714	.075
	.150	.138	.040	-.088	-.272	-.473	-1.251	-.726	-.710	-.723	.150
	.250	.080	-.009	-.118	-.256	-.465	-.937	-.714	-.697	-.715	.250
	.350	.033	-.049	-.138	-.254	-.482	-.747	-.703	-.686	-.708	.350
	.450										.450
	.550	-.022	-.082	-.148	-.223	-.240	-.539	-.672	-.667	-.693	.550
	.650	-.039	-.093	-.143	-.198	-.240	-.363	-.657	-.649	-.680	.650
	.750	-.033	-.070	-.115	-.149	-.173	-.248	-.631	-.637	-.670	.750
	.850	-.009	-.040	-.068	-.088	-.098	-.155	-.596	-.622	-.665	.850
	.900	.013	-.018	-.037	-.050	-.055	-.108	-.577	-.610	-.656	.900
	.925										.925
Lower surface	.040	-.773	-.590	-.351	-.031	.161	.339	.422	.472	.504	.040
	.091	-.730	-.536	-.208	-.038	.102	.262	.356	.422	.474	.091
	.150	-.706	-.539	-.169	-.030	.080	.220	.302	.378	.434	.150
	.250	-.713	-.544	-.131	-.019	.066	.173	.247	.313	.371	.250
	.360	-.713	-.389	-.098	-.019	.056	.147	.201	.260	.320	.360
	.450	-.684	-.204	-.069	-.003	.055	.131	.167	.221	.266	.450
	.550	-.545	-.078	-.038	.016	.062	.122	.134	.177	.218	.550
	.650	-.256	-.020	-.005	.040	.076	.120	.103	.134	.174	.650
	.800	.114	.037	.040	.073	.096	.114	.042	.059	.084	.800
	.874	.138	.050	.052	.072	.089	.093	-.016	-.006	.013	.874

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 368 lb/sq ft										
Upper surface	.000	.151	.327	.443	.362	-.049	-.989	-1.079	-.904	-.950	.000
	.025	.355	.282	.108	-.116	-.660	-1.100	-1.070	-.834	-.797	.025
	.075	.227	.126	-.022	-.210	-.412	-1.116	-.778	-.735	-.757	.075
	.150	.118	.023	-.127	-.295	-.440	-1.069	-.809	-.744	-.772	.150
	.250	.063	-.018	-.155	-.320	-.439	-.914	-.785	-.733	-.768	.250
	.350	.013	-.058	-.170	-.342	-.477	-.627	-.738	-.724	-.755	.350
	.450										.450
	.550	-.034	-.081	-.157	-.333	-.488	-.666	-.691	-.704	-.739	.550
	.650	-.047	-.091	-.146	-.155	-.506	-.642	-.670	-.689	-.723	.650
	.750	-.035	-.072	-.113	-.113	-.222	-.546	-.649	-.680	-.712	.750
	.850	-.014	-.039	-.065	-.072	-.054	-.404	-.605	-.665	-.709	.850
	.900	.006	-.017	-.033	-.038	-.013	-.365	-.602	-.654	-.697	.900
	.925										.925
Lower surface	.040	-.752	-.660	-.403	-.088	.128	.313	.418	.477	.520	.040
	.091	-.681	-.580	-.249	-.096	.070	.231	.346	.430	.495	.091
	.150	-.643	-.559	-.186	-.088	.049	.186	.289	.385	.453	.150
	.250	-.684	-.528	-.150	-.055	.042	.149	.242	.323	.393	.250
	.360	-.681	-.456	-.107	-.037	.036	.122	.197	.270	.339	.360
	.450	-.593	-.323	-.076	-.022	.039	.103	.166	.229	.294	.450
	.550	-.472	-.204	-.039	.005	.049	.090	.134	.190	.247	.550
	.650	-.323	-.097	-.002	.041	.070	.093	.110	.147	.201	.650
	.800	-.054	.002	.045	.075	.097	.081	.059	.083	.120	.800
	.874	.033	.031	.053	.083	.094	.055	.007	.023	.048	.874
	M = 0.980; q = 385 lb/sq ft										
Upper surface	.000	.192	.343	.428	.381	.094	-.856	-1.129	-1.102	-.831	.000
	.025	.336	.246	.115	-.070	-.469	-1.060	-1.144	-1.052	-.824	.025
	.075	.193	.088	-.025	-.177	-.383	-1.039	-1.030	-.896	-.776	.075
	.150	.079	-.023	-.149	-.268	-.427	-1.004	-1.054	-.939	-.806	.150
	.250	.018	-.079	-.199	-.296	-.403	-.930	-1.011	-.912	-.808	.250
	.350	-.041	-.150	-.233	-.343	-.431	-.827	-.963	-.880	-.803	.350
	.450										.450
	.550	-.114	-.202	-.307	-.378	-.483	-.597	-.851	-.815	-.783	.550
	.650	-.138	-.247	-.340	-.409	-.490	-.643	-.833	-.778	-.774	.650
	.750	-.064	-.246	-.361	-.450	-.524	-.665	-.786	-.755	-.778	.750
	.850	-.041	-.028	-.184	-.380	-.515	-.666	-.730	-.743	-.773	.850
	.900	-.021	.007	-.039	-.157	-.376	-.630	-.709	-.735	-.773	.900
	.925										.925
Lower surface	.040	-.728	-.615	-.441	-.159	.063	.303	.429	.504	.551	.040
	.091	-.658	-.537	-.298	-.174	.009	.222	.358	.450	.526	.091
	.150	-.638	-.522	-.247	-.165	-.005	.182	.304	.406	.485	.150
	.250	-.681	-.492	-.254	-.157	-.016	.134	.256	.350	.429	.250
	.360	-.719	-.488	-.277	-.178	-.020	.102	.211	.301	.376	.360
	.450	-.697	-.458	-.253	-.156	-.027	.079	.181	.265	.331	.450
	.550	-.627	-.392	-.221	-.108	-.018	.063	.150	.225	.285	.550
	.650	-.508	-.294	-.154	-.007	.007	.052	.127	.188	.242	.650
	.800	-.233	-.095	.020	.053	.034	.034	.084	.123	.166	.800
	.874	-.130	-.020	.055	.064	.029	.004	.038	.069	.102	.874

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.000	.262	.394	.468	.435	.168	-.665	-1.070	-1.030	-1.020	.000
	.025	.370	.301	.183	-.009	-.366	-.885	-1.095	-.997	-.996	.025
	.075	.234	.150	.036	-.112	-.308	-.873	-1.089	-.973	-.972	.075
	.150	.120	.039	-.080	-.197	-.351	-.841	-1.078	-.984	-.966	.150
	.250	.053	-.018	-.134	-.227	-.339	-.782	-1.046	-.944	-.927	.250
	.350	-.013	-.077	-.176	-.269	-.363	-.718	-1.028	-.912	-.888	.350
	.450										.450
	.550	-.081	-.146	-.230	-.308	-.409	-.509	-.935	-.856	-.839	.550
	.650	-.114	-.182	-.264	-.332	-.423	-.550	-.922	-.828	-.833	.650
	.750	-.139	-.212	-.293	-.376	-.442	-.576	-.871	-.794	-.816	.750
	.850	-.139	-.202	-.297	-.369	-.451	-.581	-.722	-.766	-.787	.850
	.900	-.097	-.163	-.263	-.348	-.432	-.574	-.610	-.752	-.779	.900
	.925										.925
Lower surface	.040	-.616	-.548	-.389	-.100	.102	.339	.467	.550	.596	.040
	.091	-.533	-.469	-.256	-.113	.046	.262	.398	.503	.573	.091
	.150	-.503	-.450	-.201	-.098	.029	.220	.347	.460	.535	.150
	.250	-.542	-.415	-.190	-.092	.007	.178	.296	.403	.482	.250
	.360	-.583	-.409	-.210	-.125	-.000	.149	.255	.355	.433	.360
	.450	-.564	-.394	-.196	-.107	.002	.129	.225	.322	.392	.450
	.550	-.520	-.342	-.175	-.091	.009	.113	.194	.287	.348	.550
	.650	-.447	-.269	-.137	-.054	.024	.107	.172	.253	.310	.650
	.800	-.253	-.157	-.061	.007	.044	.086	.130	.196	.236	.800
	.874	-.169	-.099	-.026	.020	.036	.059	.089	.145	.175	.874
	M = 1.125; q = 421 lb/sq ft										
Upper surface	.000	.323	.438	.499	.481	.253	-.405	-.822	-.919	-.889	.000
	.025	.398	.332	.231	.059	-.241	-.730	-.891	-.865	-.851	.025
	.075	.260	.183	.088	-.047	-.226	-.699	-.873	-.769	-.784	.075
	.150	.149	.066	-.022	-.133	-.282	-.683	-.865	-.813	-.816	.150
	.250	.087	.001	-.066	-.163	-.272	-.643	-.841	-.801	-.788	.250
	.350	.027	-.052	-.119	-.202	-.299	-.612	-.829	-.782	-.758	.350
	.450										.450
	.550	-.044	-.109	-.179	-.261	-.338	-.412	-.755	-.736	-.723	.550
	.650	-.078	-.145	-.199	-.283	-.366	-.454	-.783	-.706	-.689	.650
	.750	-.101	-.176	-.244	-.308	-.388	-.485	-.763	-.668	-.655	.750
	.850	-.106	-.184	-.245	-.320	-.396	-.495	-.707	-.628	-.639	.850
	.900	-.099	-.180	-.244	-.316	-.394	-.499	-.689	-.606	-.620	.900
	.925										.925
Lower surface	.040	-.558	-.476	-.348	-.063	.131	.341	.496	.614	.682	.040
	.091	-.452	-.403	-.199	-.075	.070	.264	.427	.571	.657	.091
	.150	-.422	-.365	-.137	-.069	.051	.228	.381	.534	.620	.150
	.250	-.436	-.312	-.132	-.055	.034	.187	.338	.479	.573	.250
	.360	-.429	-.293	-.159	-.092	-.006	.175	.304	.437	.527	.360
	.450	-.421	-.295	-.146	-.084	-.002	.159	.277	.406	.490	.450
	.550	-.414	-.276	-.138	-.087	.002	.158	.254	.378	.450	.550
	.650	-.394	-.233	-.109	-.049	.026	.162	.239	.343	.414	.650
	.800	-.287	-.161	-.056	-.003	.067	.154	.202	.294	.346	.800
	.874	-.221	-.123	-.027	.017	.077	.133	.170	.249	.290	.874

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 436 lb/sq ft										
Upper surface	.000	.340	.438	.482	.490	.301	-.204	-.652	-.835	-.783	.000
	.025	.408	.341	.240	.090	-.146	-.586	-.758	-.791	-.709	.025
	.075	.273	.196	.105	-.016	-.167	-.557	-.726	-.788	-.731	.075
	.150	.157	.084	-.002	-.104	-.236	-.555	-.727	-.818	-.742	.150
	.250	.093	.026	-.050	-.127	-.228	-.529	-.708	-.814	-.744	.250
	.350	.043	-.027	-.091	-.165	-.252	-.514	-.705	-.815	-.753	.350
	.450										.450
	.550	-.013	-.081	-.151	-.215	-.275	-.458	-.675	-.790	-.738	.550
	.650	-.042	-.101	-.172	-.239	-.310	-.374	-.671	-.800	-.766	.650
	.750	-.075	-.143	-.201	-.270	-.336	-.419	-.673	-.748	-.741	.750
	.850	-.091	-.149	-.226	-.284	-.348	-.430	-.652	-.663	-.729	.850
	.900	-.085	-.155	-.219	-.287	-.353	-.438	-.664	-.619	-.741	.900
	.925										.925
Lower surface	.040	-.575	-.597	-.365	-.066	.132	.337	.486	.597	.671	.040
	.091	-.454	-.451	-.222	-.062	.074	.261	.422	.548	.641	.091
	.150	-.397	-.352	-.123	-.053	.060	.223	.382	.513	.609	.150
	.250	-.369	-.253	-.123	-.033	.048	.187	.350	.464	.563	.250
	.360	-.337	-.211	-.133	-.070	-.002	.157	.327	.427	.520	.360
	.450	-.301	-.196	-.120	-.052	.020	.185	.309	.399	.485	.450
	.550	-.286	-.185	-.108	-.039	.031	.176	.298	.374	.450	.550
	.650	-.283	-.153	-.076	-.007	.062	.191	.290	.347	.419	.650
	.800	-.230	-.103	-.031	.037	.103	.208	.264	.306	.361	.800
	.874	-.191	-.074	-.001	.063	.123	.195	.233	.268	.314	.874

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.022	.418	.361	.258	.013	-.662	-.563	-.504	-.504	-.558	.022
	.076	.307	.224	.106	-.081	-.340	-.546	-.490	-.501	-.555	.076
	.150	.222	.137	.030	-.121	-.322	-.524	-.478	-.491	-.550	.150
	.250	.144	.070	-.021	-.143	-.298	-.495	-.464	-.481	-.542	.250
	.370	.078	.016	-.062	-.158	-.289	-.463	-.452	-.473	-.538	.370
	.450	.036	-.015	-.086	-.167	-.267	-.423	-.431	-.457	-.529	.450
	.550	.000	-.038	-.098	-.162	-.245	-.402	-.429	-.457	-.529	.550
	.600										.600
	.650	-.030	-.055	-.108	-.155	-.213	-.380	-.420	-.455	-.529	.650
	.750	-.053	-.047	-.088	-.123	-.166	-.348	-.404	-.445	-.519	.750
	.850	-.097	-.038	-.069	-.089	-.114	-.330	-.399	-.441	-.513	.850
	.900										.900
Lower surface	.078	-.463	-.477	-.531	-.062	.139	.287	.333	.374	.392	.078
	.150	-.441	-.471	-.351	-.052	.086	.210	.255	.303	.343	.150
	.250	-.424	-.476	-.152	-.015	.093	.184	.209	.251	.284	.250
	.360	-.421	-.499	-.070	-.007	.070	.141	.157	.189	.218	.360
	.450										.450
	.560	-.427	-.471	-.019	.013	.070	.088	.076	.095	.111	.560
	.610	-.438	-.432	-.007	.025	.069	.075	.058	.071	.087	.610
	.700	-.425	-.346	.023	.046	.087	.064	.034	.041	.050	.700
	.800	-.412	-.175	.045	.073	.095	.030	-.006	-.005	-.008	.800
	M = 0.900; q = 358 lb/sq ft										
Upper surface	.022	.416	.353	.246	-.022	-.765	-1.146	-.529	-.572	-.622	.022
	.076	.308	.219	.099	-.118	-.577	-.882	-.519	-.564	-.617	.076
	.150	.223	.134	.021	-.160	-.367	-.868	-.509	-.558	-.609	.150
	.250	.146	.062	-.037	-.184	-.349	-.820	-.494	-.548	-.600	.250
	.370	.082	.008	-.081	-.210	-.360	-.754	-.483	-.538	-.593	.370
	.450	.042	-.025	-.106	-.215	-.342	-.682	-.468	-.527	-.583	.450
	.550	.009	-.048	-.119	-.204	-.293	-.627	-.472	-.525	-.582	.550
	.600										.600
	.650	-.026	-.070	-.127	-.194	-.254	-.569	-.468	-.524	-.577	.650
	.750	-.042	-.061	-.106	-.152	-.183	-.511	-.465	-.519	-.571	.750
	.850	-.079	-.050	-.083	-.109	-.116	-.455	-.464	-.513	-.565	.850
	.900										.900
Lower surface	.078	-.508	-.486	-.542	-.057	.142	.282	.324	.369	.401	.078
	.150	-.498	-.478	-.404	-.052	.085	.209	.251	.304	.356	.150
	.250	-.488	-.479	-.222	-.021	.093	.183	.209	.254	.302	.250
	.360	-.495	-.501	-.099	-.007	.071	.145	.152	.198	.242	.360
	.450										.450
	.560	-.539	-.497	-.023	.021	.072	.106	.075	.105	.138	.560
	.610	-.546	-.481	-.011	.030	.072	.095	.055	.082	.115	.610
	.700	-.539	-.427	.024	.057	.089	.089	.036	.059	.082	.700
	.800	-.509	-.307	.053	.080	.102	.063	-.003	.009	.031	.800

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
		M = 0.940; q = 368 lb/sq ft									
Upper surface	.022	.411	.361	.243	-.079	-.685	-1.231	-.620	-.641	-.687	.022
	.076	.304	.228	.093	-.174	-.598	-1.131	-.618	-.627	-.677	.076
	.150	.219	.143	.012	-.210	-.464	-1.067	-.607	-.618	-.667	.150
	.250	.144	.072	-.039	-.204	-.456	-1.046	-.573	-.605	-.659	.250
	.370	.082	.016	-.087	-.206	-.492	-.997	-.564	-.598	-.652	.370
	.450	.041	-.015	-.115	-.220	-.528	-.903	-.551	-.584	-.643	.450
	.550	.006	-.044	-.125	-.216	-.549	-.765	-.527	-.582	-.636	.550
	.600										.600
	.650	-.024	-.065	-.137	-.198	-.243	-.584	-.516	-.579	-.632	.650
	.750	-.040	-.059	-.114	-.161	-.100	-.482	-.503	-.567	-.622	.750
	.850	-.070	-.047	-.089	-.108	-.061	-.432	-.496	-.563	-.615	.850
	.900										.900
Lower surface	.078	-.529	-.504	-.524	-.071	.107	.247	.313	.371	.415	.078
	.150	-.490	-.491	-.390	-.064	.059	.177	.240	.308	.373	.150
	.250	-.465	-.488	-.228	-.020	.079	.159	.205	.264	.321	.250
	.360	-.443	-.511	-.110	-.009	.061	.127	.151	.207	.263	.360
	.450										.450
	.560	-.492	-.530	-.023	.013	.070	.098	.079	.120	.170	.560
	.610	-.517	-.532	-.011	.026	.071	.091	.062	.099	.143	.610
	.700	-.485	-.500	.025	.047	.091	.096	.050	.077	.114	.700
	.800	-.447	-.404	.053	.073	.107	.085	.014	.036	.064	.800
		M = 0.980; q = 385 lb/sq ft									
Upper surface	.022	.367	.276	.175	.008	-.372	-1.140	-.978	-.650	-.750	.022
	.076	.255	.134	.018	-.136	-.407	-1.040	-.836	-.659	-.742	.076
	.150	.170	.047	-.073	-.208	-.422	-.999	-.837	-.668	-.743	.150
	.250	.098	-.034	-.137	-.254	-.407	-.970	-.816	-.671	-.744	.250
	.370	.040	-.105	-.212	-.310	-.441	-.952	-.793	-.675	-.741	.370
	.450	.007	-.146	-.268	-.351	-.480	-.927	-.765	-.671	-.729	.450
	.550	-.016	-.075	-.314	-.388	-.529	-.940	-.747	-.682	-.736	.550
	.600										.600
	.650	-.035	-.057	-.342	-.422	-.551	-.905	-.730	-.687	-.728	.650
	.750	-.036	-.050	-.083	-.442	-.540	-.891	-.706	-.684	-.719	.750
	.850	-.054	-.037	.001	-.255	-.520	-.889	-.686	-.677	-.711	.850
	.900										.900
Lower surface	.078	-.693	-.729	-.694	-.337	.009	.203	.327	.398	.441	.078
	.150	-.637	-.715	-.606	-.193	-.028	.132	.256	.337	.403	.150
	.250	-.592	-.712	-.485	-.065	.006	.111	.220	.294	.351	.250
	.360	-.562	-.694	-.314	-.061	-.010	.069	.171	.243	.298	.360
	.450										.450
	.560	-.511	-.595	-.038	-.008	.008	.038	.104	.159	.207	.560
	.610	-.521	-.563	.005	.009	.011	.028	.088	.137	.186	.610
	.700	-.545	-.455	.050	.037	.035	.035	.077	.119	.157	.700
	.800	-.552	-.200	.092	.081	.052	.023	.047	.080	.110	.800

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.022	.380	.321	.225	.075	-.254	-.962	-1.168	-.823	-.637	.022
	.076	.263	.187	.079	-.066	-.313	-.870	-1.113	-.766	-.636	.076
	.150	.173	.098	-.013	-.137	-.337	-.835	-1.075	-.760	-.642	.150
	.250	.091	.020	-.077	-.183	-.329	-.816	-1.049	-.754	-.648	.250
	.370	.013	-.049	-.141	-.243	-.364	-.803	-1.020	-.740	-.667	.370
	.450	-.039	-.104	-.188	-.288	-.397	-.781	-.983	-.719	-.667	.450
	.550	-.072	-.145	-.237	-.335	-.441	-.807	-.990	-.722	-.678	.550
	.600										.600
	.650	-.117	-.190	-.279	-.355	-.473	-.780	-.896	-.709	-.676	.650
	.750	-.136	-.222	-.312	-.389	-.489	-.770	-.799	-.690	-.672	.750
	.850	-.117	-.231	-.334	-.414	-.502	-.769	-.744	-.674	-.671	.850
	.900										.900
Lower surface	.078	-.714	-.681	-.653	-.336	.008	.238	.357	.449	.500	.078
	.150	-.692	-.679	-.539	-.190	-.021	.170	.289	.395	.461	.150
	.250	-.690	-.668	-.408	-.139	.011	.154	.253	.346	.415	.250
	.360	-.679	-.651	-.329	-.132	.002	.116	.206	.300	.362	.360
	.450										.450
	.560	-.626	-.574	-.202	-.065	.011	.083	.148	.226	.274	.560
	.610	-.622	-.545	-.163	-.046	.014	.074	.134	.207	.256	.610
	.700	-.564	-.483	-.094	-.017	.032	.078	.123	.188	.229	.700
	.800	-.531	-.381	-.042	.011	.043	.068	.100	.152	.184	.800
	M = 1.125; q = 421 lb/sq ft										
Upper surface	.022	.407	.339	.265	.120	-.136	-.779	-.956	-.699	-.535	.022
	.076	.293	.209	.126	-.010	-.225	-.693	-.897	-.607	-.537	.076
	.150	.208	.120	.041	-.079	-.267	-.671	-.868	-.609	-.539	.150
	.250	.137	.054	-.022	-.119	-.254	-.657	-.850	-.597	-.526	.250
	.370	.066	-.008	-.078	-.167	-.283	-.651	-.821	-.589	-.539	.370
	.450	.008	-.060	-.129	-.211	-.315	-.641	-.785	-.566	-.544	.450
	.550	-.023	-.106	-.171	-.262	-.357	-.670	-.810	-.570	-.539	.550
	.600										.600
	.650	-.061	-.137	-.207	-.284	-.383	-.646	-.792	-.559	-.549	.650
	.750	-.092	-.169	-.234	-.311	-.404	-.628	-.717	-.544	-.544	.750
	.850	-.111	-.193	-.258	-.337	-.423	-.624	-.665	-.532	-.541	.850
	.900										.900
Lower surface	.078	-.649	-.602	-.573	-.393	.007	.273	.405	.536	.595	.078
	.150	-.623	-.584	-.434	-.172	-.014	.213	.344	.485	.558	.150
	.250	-.629	-.555	-.307	-.141	.006	.204	.314	.445	.514	.250
	.360	-.611	-.540	-.259	-.128	.008	.177	.276	.401	.465	.360
	.450										.450
	.560	-.552	-.478	-.178	-.071	.046	.156	.232	.332	.382	.560
	.610	-.541	-.449	-.153	-.046	.057	.149	.221	.316	.363	.610
	.700	-.489	-.395	-.091	-.003	.087	.157	.213	.302	.339	.700
	.800	-.449	-.328	-.028	.047	.112	.156	.196	.269	.299	.800

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 436 lb/sq ft										
Upper surface	.022	.416	.363	.287	.164	-.035	-.590	-.799	-.818	-.734	.022
	.078	.309	.239	.154	.035	-.138	-.521	-.739	-.799	-.707	.078
	.150	.226	.156	.073	-.033	-.199	-.512	-.714	-.793	-.696	.150
	.250	.157	.092	.018	-.073	-.194	-.508	-.703	-.801	-.702	.250
	.370	.094	.027	-.038	-.114	-.222	-.516	-.695	-.792	-.692	.370
	.450	.043	-.014	-.087	-.152	-.252	-.516	-.673	-.761	-.660	.450
	.550	-.001	-.060	-.128	-.203	-.287	-.550	-.673	-.779	-.672	.550
	.600										.600
	.650	-.040	-.103	-.167	-.234	-.312	-.543	-.661	-.745	-.648	.650
	.750	-.072	-.136	-.196	-.257	-.332	-.513	-.654	-.710	-.630	.750
	.850	-.097	-.157	-.228	-.280	-.351	-.507	-.659	-.670	-.597	.850
	.900										.900
Lower surface	.078	-.694	-.686	-.521	-.357	.001	.263	.417	.503	.574	.078
	.150	-.665	-.663	-.441	-.130	-.008	.198	.358	.452	.540	.150
	.250	-.626	-.576	-.266	-.104	.000	.202	.339	.417	.502	.250
	.360	-.590	-.418	-.238	-.115	-.008	.178	.304	.380	.458	.360
	.450										.450
	.560	-.445	-.279	-.189	-.083	.018	.176	.266	.321	.388	.560
	.610	-.382	-.258	-.173	-.067	.029	.184	.260	.308	.372	.610
	.700	-.297	-.219	-.131	-.031	.074	.204	.259	.300	.356	.700
	.800	-.241	-.167	-.054	.050	.134	.213	.247	.277	.324	.800

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.073	.324	.285	.210	.050	-.227	-.395	-.328	-.370	-.440	.073
	.150	.214	.175	.095	-.047	-.267	-.370	-.326	-.371	-.438	.150
	.250	.118	.086	.017	-.087	-.253	-.344	-.325	-.368	-.436	.250
	.350	.048	.020	-.036	-.115	-.233	-.324	-.318	-.367	-.435	.350
	.460	-.020	-.035	-.074	-.135	-.215	-.303	-.317	-.367	-.435	.460
	.550	-.075	-.076	-.101	-.142	-.200	-.283	-.317	-.367	-.437	.550
	.660	-.109	-.097	-.095	-.118	-.161	-.264	-.312	-.367	-.441	.660
	.750	-.122	-.106	-.066	-.079	-.114	-.247	-.310	-.370	-.441	.750
	.800										.800
Lower surface	.100	-.242	-.282	-.438	-.150	.077	.169	.224	.268	.285	.100
	.200	-.217	-.254	-.393	-.097	.049	.108	.152	.195	.213	.200
	.250	-.208	-.249	-.392	-.068	.043	.087	.113	.151	.176	.250
	.350	-.207	-.230	-.346	-.046	.033	.037	.056	.089	.099	.350
	.450	-.192	-.213	-.280	-.032	.032	.005	.004	.019	.018	.450
	.550	-.171	-.202	-.178	.007	.026	-.019	-.033	-.032	-.025	.550
	.600	-.166	-.199	-.113	.019	.029	-.027	-.047	-.051	-.054	.600
	.690	-.156	-.191	-.016	.036	.039	-.049	-.078	-.092	-.103	.690
	M = 0.900; q = 358 lb/sq ft										
Upper surface	.073	.337	.294	.210	.018	-.480	-.453	-.391	-.436	-.493	.073
	.150	.229	.184	.092	-.076	-.295	-.442	-.392	-.433	-.490	.150
	.250	.131	.090	.010	-.124	-.306	-.422	-.391	-.434	-.487	.250
	.350	.056	.020	-.051	-.170	-.306	-.406	-.392	-.429	-.483	.350
	.460	-.028	-.044	-.105	-.190	-.245	-.386	-.395	-.427	-.480	.460
	.550	-.093	-.095	-.133	-.181	-.221	-.374	-.395	-.427	-.479	.550
	.660	-.141	-.117	-.116	-.149	-.179	-.360	-.398	-.430	-.480	.660
	.750	-.155	-.118	-.079	-.095	-.122	-.351	-.403	-.429	-.479	.750
	.800										.800
Lower surface	.100	-.240	-.296	-.463	-.150	.088	.204	.219	.268	.297	.100
	.200	-.219	-.271	-.412	-.105	.055	.139	.146	.205	.234	.200
	.250	-.220	-.267	-.406	-.067	.045	.106	.110	.162	.197	.250
	.350	-.212	-.249	-.364	-.041	.025	.049	.046	.094	.123	.350
	.450	-.199	-.229	-.305	-.021	.011	-.007	-.023	.022	.045	.450
	.550	-.193	-.222	-.223	.011	.021	-.038	-.073	-.046	-.017	.550
	.600	-.189	-.217	-.166	.021	.024	-.053	-.091	-.071	-.048	.600
	.690	-.184	-.213	-.057	.041	.034	-.080	-.134	-.123	-.109	.690

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 368 lb/sq ft										
Upper surface	.073	.347	.310	.211	.002	-.750	-.476	-.453	-.485	-.551	.073
	.150	.243	.200	.095	-.081	-.458	-.445	-.446	-.479	-.547	.150
	.250	.145	.107	.011	-.128	-.443	-.428	-.442	-.476	-.542	.250
	.350	.063	.036	-.057	-.177	-.302	-.406	-.440	-.471	-.534	.350
	.460	-.023	-.034	-.122	-.238	-.150	-.392	-.437	-.469	-.530	.460
	.550	-.098	-.085	-.149	-.189	-.172	-.385	-.440	-.469	-.527	.550
	.660	-.154	-.108	-.125	-.139	-.159	-.378	-.442	-.467	-.526	.660
	.750	-.173	-.107	-.079	-.088	-.110	-.367	-.442	-.469	-.521	.750
	.800										.800
Lower surface	.100	-.317	-.335	-.434	-.154	.072	.153	.213	.282	.319	.100
	.200	-.292	-.306	-.389	-.119	.045	.101	.145	.218	.261	.200
	.250	-.287	-.304	-.397	-.074	.049	.070	.114	.180	.223	.250
	.350	-.277	-.290	-.346	-.061	.020	.020	.050	.114	.161	.350
	.450	-.260	-.274	-.293	-.043	.002	-.035	-.018	.044	.086	.450
	.550	-.246	-.273	-.223	.011	.021	-.083	-.072	-.025	.013	.550
	.600	-.237	-.272	-.168	.021	.027	-.093	-.096	-.057	-.016	.600
	.690	-.224	-.281	-.060	.041	.038	-.121	-.142	-.116	-.082	.690
	M = 0.980; q = 385 lb/sq ft										
Upper surface	.073	.341	.301	.158	-.034	-.360	-1.044	-.626	-.608	-.656	.073
	.150	.237	.196	.064	-.133	-.413	-.985	-.619	-.602	-.651	.150
	.250	.148	.109	.005	-.196	-.435	-.949	-.616	-.597	-.644	.250
	.350	.071	.037	-.033	-.255	-.464	-.929	-.604	-.590	-.639	.350
	.460	-.014	-.035	-.084	-.311	-.488	-.901	-.595	-.587	-.636	.460
	.550	-.091	-.092	-.114	-.394	-.532	-.887	-.588	-.582	-.632	.550
	.660	-.155	-.111	-.095	-.392	-.565	-.818	-.587	-.578	-.627	.660
	.750	-.174	-.102	-.046	.010	-.525	-.715	-.578	-.572	-.622	.750
	.800										.800
Lower surface	.100	-.366	-.322	-.326	-.202	-.051	.123	.237	.309	.351	.100
	.200	-.339	-.282	-.280	-.175	-.052	.078	.171	.250	.296	.200
	.250	-.344	-.269	-.278	-.092	-.026	.063	.144	.216	.262	.250
	.350	-.333	-.262	-.277	-.064	-.038	.023	.086	.152	.198	.350
	.450	-.319	-.261	-.274	-.064	-.078	-.042	.013	.083	.127	.450
	.550	-.319	-.280	-.269	.007	-.089	-.095	-.042	.014	.062	.550
	.600	-.314	-.291	-.255	.022	-.078	-.117	-.066	-.016	.030	.600
	.690	-.303	-.298	-.207	.057	-.068	-.153	-.114	-.077	-.040	.690

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.073	.325	.269	.170	.027	-.232	-.861	-1.100	-.563	-.602	.073
	.150	.214	.161	.063	-.074	-.305	-.814	-1.068	-.563	-.599	.150
	.250	.127	.073	-.012	-.145	-.337	-.789	-1.038	-.572	-.603	.250
	.350	.054	-.001	-.082	-.196	-.368	-.788	-1.016	-.575	-.602	.350
	.460	-.032	-.092	-.155	-.241	-.397	-.794	-.993	-.583	-.603	.460
	.550	-.124	-.190	-.252	-.328	-.451	-.801	-.964	-.586	-.605	.550
	.660	-.199	-.283	-.351	-.412	-.496	-.772	-.894	-.592	-.608	.660
	.750	-.186	-.288	-.385	-.431	-.489	-.756	-.826	-.595	-.605	.750
	.800										.800
Lower surface	.100	-.558	-.528	-.687	-.386	-.087	.139	.275	.371	.407	.100
	.200	-.505	-.478	-.617	-.276	-.115	.093	.214	.314	.355	.200
	.250	-.489	-.468	-.599	-.235	-.083	.095	.191	.278	.328	.250
	.350	-.470	-.441	-.541	-.186	-.061	.058	.140	.224	.267	.350
	.450	-.447	-.416	-.491	-.157	-.072	.005	.075	.159	.198	.450
	.550	-.421	-.399	-.447	-.094	-.084	-.043	.017	.090	.139	.550
	.600	-.410	-.394	-.406	-.070	-.076	-.064	-.009	.061	.107	.600
	.690	-.386	-.397	-.326	-.057	-.085	-.098	-.067	-.003	.039	.690
	M = 1.125; q = 421 lb/sq ft										
Upper surface	.073	.365	.293	.209	.101	-.103	-.684	-.897	-.468	-.481	.073
	.150	.267	.197	.115	-.000	-.189	-.637	-.853	-.470	-.484	.150
	.250	.188	.123	.045	-.071	-.234	-.621	-.824	-.470	-.492	.250
	.350	.119	.058	-.022	-.122	-.262	-.618	-.815	-.469	-.493	.350
	.460	.043	-.007	-.066	-.166	-.302	-.627	-.808	-.466	-.489	.460
	.550	-.047	-.099	-.152	-.237	-.347	-.636	-.794	-.464	-.484	.550
	.660	-.134	-.199	-.248	-.324	-.399	-.618	-.777	-.464	-.491	.660
	.750	-.159	-.231	-.289	-.359	-.404	-.597	-.765	-.464	-.492	.750
	.800										.800
Lower surface	.100	-.612	-.617	-.693	-.563	-.099	.194	.349	.473	.509	.100
	.200	-.563	-.575	-.619	-.328	-.115	.147	.299	.423	.463	.200
	.250	-.543	-.576	-.571	-.245	-.086	.164	.279	.390	.433	.250
	.350	-.480	-.537	-.382	-.264	-.056	.138	.238	.340	.378	.350
	.450	-.439	-.509	-.300	-.247	-.020	.098	.179	.279	.313	.450
	.550	-.459	-.522	-.245	-.116	.012	.058	.129	.213	.255	.550
	.600	-.454	-.511	-.210	-.058	.020	.037	.103	.188	.228	.600
	.690	-.457	-.493	-.141	.010	.022	.005	.053	.126	.166	.690

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Concluded

(f) 95-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 436 lb/sq ft										
Upper surface	.073	.368	.304	.231	.139	-.027	-.526	-.755	-.710	-.470	.073
	.150	.274	.219	.143	.041	-.123	-.487	-.715	-.700	-.450	.150
	.250	.205	.142	.067	-.025	-.158	-.480	-.690	-.691	-.439	.250
	.350	.142	.083	.008	-.074	-.192	-.483	-.684	-.693	-.422	.350
	.460	.086	.035	-.038	-.117	-.228	-.496	-.684	-.695	-.402	.460
	.550	.005	-.038	-.100	-.172	-.270	-.510	-.681	-.691	-.402	.550
	.660	-.089	-.131	-.190	-.248	-.322	-.514	-.660	-.671	-.417	.660
	.750	-.128	-.172	-.235	-.287	-.341	-.504	-.652	-.654	-.438	.750
	.800										.800
Lower surface	.100	-.717	-.734	-.618	-.524	-.106	.220	.362	.447	.510	.100
	.200	-.676	-.681	-.570	-.434	-.135	.156	.315	.407	.473	.200
	.250	-.701	-.699	-.577	-.290	-.094	.182	.309	.381	.453	.250
	.350	-.661	-.666	-.530	-.244	-.084	.167	.276	.344	.404	.350
	.450	-.617	-.634	-.464	-.261	-.073	.145	.231	.294	.344	.450
	.550	-.644	-.642	-.343	-.180	.005	.131	.192	.242	.301	.550
	.600	-.620	-.621	-.188	-.137	.037	.112	.168	.219	.275	.600
	.690	-.615	-.511	-.101	.003	.084	.079	.119	.161	.216	.690

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE

OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY

(a) 12-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 627 lb/sq ft										
Upper surface	.000	.183	.408	.514	.529	.498	.249	-.087	-.360	-.607	.000
	.032	.309	.203	.089	-.031	-.163	-.474	-.895	-1.644	-1.590	.032
	.078	.195	.109	.017	-.070	-.167	-.366	-.573	-.813	-1.454	.078
	.150	.093	.015	-.060	-.143	-.226	-.415	-.615	-.767	-1.080	.150
	.250	.015	-.056	-.127	-.198	-.276	-.447	-.619	-.602	-.828	.250
	.350	-.003	-.065	-.129	-.192	-.262	-.413	-.552	-.582	-.733	.350
	.450	-.048	-.107	-.161	-.223	-.287	-.429	-.511	-.612	-.663	.450
	.550	-.085	-.138	-.191	-.247	-.307	-.436	-.460	-.607	-.648	.550
	.650	-.063	-.108	-.153	-.197	-.244	-.331	-.391	-.529	-.579	.650
	.760	-.050	-.089	-.123	-.159	-.192	-.247	-.343	-.464	-.538	.760
	.840	-.057	-.081	-.106	-.128	-.155	-.188	-.269	-.403	-.545	.840
	.926	-.033	-.053	-.063	-.079	-.092	-.110	-.168	-.305	-.508	.926
Lower surface	.035	-.381	-.243	-.100	.028	.141	.358	.543	.685	.809	.035
	.082										.082
	.150	-.241	-.147	-.066	.007	.079	.227	.368	.478	.577	.150
	.250										.250
	.350	-.229	-.153	-.086	-.028	.027	.147	.258	.349	.436	.350
	.450	-.242	-.174	-.113	-.060	-.008	.102	.203	.284	.366	.450
	.550	-.227	-.170	-.113	-.067	-.021	.080	.168	.239	.311	.550
	.650	-.207	-.156	-.106	-.066	-.026	.061	.135	.194	.258	.650
	.750	-.138	-.098	-.061	-.030	.002	.074	.132	.172	.222	.750
	.850	-.089	-.061	-.032	-.009	.016	.073	.111	.136	.164	.850
	.900	-.069	-.047	-.025	-.007	.012	.059	.084	.095	.110	.900
	M = 0.900; q = 713 lb/sq ft										
Upper surface	.000	.310	.471	.549	.560	.541	.380	.129	-.093	-.311	.000
	.032	.318	.213	.111	-.003	-.115	-.386	-.786	-1.338	-1.500	.032
	.078	.207	.119	.039	-.044	-.124	-.272	-.473	-.708	-1.464	.078
	.150	.097	.020	-.053	-.129	-.200	-.359	-.503	-.623	-.675	.150
	.250	.011	-.063	-.133	-.207	-.280	-.408	-.539	-.637	-.706	.250
	.350	-.010	-.076	-.140	-.205	-.264	-.399	-.530	-.622	-.685	.350
	.450	-.062	-.127	-.189	-.256	-.315	-.433	-.552	-.471	-.696	.450
	.550	-.114	-.179	-.250	-.332	-.388	-.498	-.618	-.553	-.688	.550
	.650	-.086	-.140	-.196	-.273	-.381	-.488	-.598	-.560	-.611	.650
	.760	-.074	-.118	-.163	-.213	-.316	-.496	-.506	-.554	-.584	.760
	.840	-.079	-.109	-.138	-.170	-.191	-.490	-.397	-.599	-.606	.840
	.926	-.050	-.067	-.082	-.095	-.095	-.182	-.228	-.516	-.561	.926
Lower surface	.035	-.342	-.217	-.087	.039	.148	.361	.553	.704	.837	.035
	.082										.082
	.150	-.230	-.147	-.069	.005	.079	.230	.376	.495	.606	.150
	.250										.250
	.350	-.245	-.172	-.105	-.040	.021	.146	.263	.366	.463	.350
	.450	-.278	-.211	-.143	-.081	-.023	.096	.203	.299	.394	.450
	.550	-.323	-.235	-.157	-.096	-.040	.070	.165	.250	.341	.550
	.650	-.323	-.213	-.145	-.093	-.045	.051	.130	.205	.289	.650
	.750	-.193	-.135	-.088	-.049	-.010	.067	.126	.186	.255	.750
	.850	-.113	-.083	-.051	-.022	.008	.067	.103	.144	.204	.850
	.900	-.084	-.062	-.038	-.017	.008	.055	.074	.100	.150	.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 743 lb/sq ft										
Upper surface	.000	.361	.503	.563	.578	.562	.428	.206	-.008	-.218	.000
	.032	.330	.226	.124	.018	-.087	-.342	-.704	-1.221	-1.421	.032
	.078	.218	.131	.051	-.025	-.096	-.227	-.424	-.645	-1.356	.078
	.150	.105	.027	-.044	-.115	-.174	-.330	-.447	-.560	-.648	.150
	.250	.014	-.063	-.132	-.202	-.250	-.371	-.484	-.580	-.637	.250
	.350	-.010	-.078	-.138	-.197	-.261	-.380	-.488	-.585	-.579	.350
	.450	-.069	-.136	-.194	-.248	-.299	-.409	-.510	-.604	-.678	.450
	.550	-.135	-.218	-.276	-.326	-.370	-.473	-.574	-.621	-.670	.550
	.650	-.104	-.174	-.266	-.325	-.364	-.477	-.573	-.445	-.667	.650
	.760	-.094	-.155	-.237	-.324	-.374	-.482	-.572	-.538	-.639	.760
	.840	-.096	-.138	-.175	-.342	-.405	-.533	-.568	-.563	-.665	.840
	.926	-.063	-.079	-.087	-.170	-.323	-.509	-.440	-.548	-.626	.926
Lower surface	.035	-.304	-.197	-.075	.045	.157	.366	.558	.716	.853	.035
	.082										.082
	.150	-.204	-.136	-.065	.005	.083	.234	.382	.512	.625	.150
	.250										.250
	.350	-.247	-.174	-.111	-.050	.018	.144	.268	.381	.485	.350
	.450	-.267	-.205	-.157	-.100	-.032	.089	.206	.314	.416	.450
	.550	-.316	-.267	-.202	-.130	-.059	.058	.166	.266	.364	.550
	.650	-.347	-.295	-.203	-.126	-.065	.034	.130	.222	.312	.650
	.750	-.327	-.254	-.116	-.071	-.028	.047	.125	.201	.281	.750
	.850	-.293	-.126	-.062	-.038	-.010	.040	.101	.163	.233	.850
	.900	-.212	-.069	-.044	-.029	-.012	.019	.067	.121	.183	.900
	M = 0.980; q = 771 lb/sq ft										
Upper surface	.000	.417	.538	.599	.605	.592	.479	.269			.000
	.032	.348	.253	.159	.056	-.046	-.291	-.624			.032
	.078	.238	.159	.087	.016	-.051	-.190	-.379			.078
	.150	.126	.052	-.010	-.075	-.139	-.272	-.388			.150
	.250	.028	-.043	-.106	-.160	-.215	-.323	-.421			.250
	.350	.004	-.056	-.113	-.175	-.226	-.332	-.431			.350
	.450	-.060	-.115	-.162	-.215	-.268	-.365	-.454			.450
	.550	-.150	-.201	-.240	-.290	-.338	-.427	-.517			.550
	.650	-.133	-.194	-.245	-.293	-.343	-.433	-.522			.650
	.760	-.130	-.202	-.256	-.307	-.355	-.444	-.535			.760
	.840	-.177	-.247	-.298	-.350	-.402	-.496	-.580			.840
	.926	-.179	-.239	-.295	-.342	-.385	-.487	-.574			.926
Lower surface	.035	-.254	-.153	-.037	.074	.177	.383	.577			.035
	.082										.082
	.150	-.158	-.103	-.035	.030	.101	.251	.402			.150
	.250										.250
	.350	-.224	-.155	-.091	-.032	.027	.157	.287			.350
	.450	-.243	-.181	-.124	-.083	-.030	.097	.224			.450
	.550	-.295	-.239	-.183	-.137	-.079	.060	.182			.550
	.650	-.327	-.277	-.223	-.173	-.105	.031	.144			.650
	.750	-.316	-.256	-.207	-.148	-.073	.041	.138			.750
	.850	-.306	-.244	-.190	-.136	-.060	.031	.114			.850
	.900	-.316	-.258	-.203	-.147	-.069	.005	.081			.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 803 lb/sq ft										
Upper surface	.000	.481	.585	.637	.645	.637	.543	.353			.000
	.032	.401	.298	.202	.106	.012	-.211	-.517			.032
	.078	.293	.209	.135	.070	.009	-.121	-.300			.078
	.150	.182	.105	.040	-.018	-.082	-.189	-.298			.150
	.250	.087	.010	-.058	-.104	-.148	-.244	-.336			.250
	.350	.065	.002	-.063	-.120	-.164	-.253	-.348			.350
	.450	.003	-.052	-.112	-.161	-.202	-.285	-.374			.450
	.550	-.083	-.134	-.181	-.228	-.267	-.346	-.435			.550
	.650	-.076	-.132	-.187	-.235	-.274	-.356	-.441			.650
	.760	-.079	-.139	-.199	-.247	-.287	-.368	-.452			.760
	.840	-.128	-.189	-.245	-.293	-.335	-.420	-.498			.840
	.926	-.129	-.182	-.238	-.281	-.317	-.411	-.495			.926
Lower surface	.035	-.177	-.096	.012	.116	.219	.430	.619			.035
	.082										.082
	.150	-.088	-.043	.017	.076	.147	.301	.448			.150
	.250										.250
	.350	-.153	-.104	-.044	.016	.080	.205	.335			.350
	.450	-.170	-.127	-.075	-.024	.027	.146	.272			.450
	.550	-.221	-.180	-.131	-.081	-.024	.108	.231			.550
	.650	-.251	-.216	-.165	-.117	-.055	.077	.193			.650
	.750	-.241	-.199	-.153	-.097	-.030	.090	.188			.750
	.850	-.235	-.185	-.140	-.081	-.026	.083	.167			.850
	.900	-.246	-.195	-.151	-.093	-.041	.062	.138			.900
	M = 1.125; q = 856 lb/sq ft										
Upper surface	.000	.536	.600	.617	.625	.616	.577	.486			.000
	.032	.337	.234	.156	.052	-.033	-.271	-.460			.032
	.078	.263	.180	.123	.054	-.008	-.156	-.293			.078
	.150	.169	.101	.046	-.019	-.074	-.170	-.261			.150
	.250	.075	.022	-.019	-.073	-.124	-.211	-.275			.250
	.350	.059	-.001	-.044	-.094	-.141	-.226	-.286			.350
	.450	.016	-.040	-.082	-.131	-.170	-.241	-.306			.450
	.550	-.050	-.102	-.139	-.183	-.219	-.287	-.359			.550
	.650	-.046	-.100	-.140	-.186	-.229	-.302	-.365			.650
	.760	-.060	-.113	-.154	-.199	-.240	-.312	-.365			.760
	.840	-.117	-.167	-.200	-.247	-.285	-.347	-.402			.840
	.926	-.102	-.156	-.193	-.236	-.269	-.341	-.401			.926
Lower surface	.035	-.195	-.088	-.008	.084	.168	.361	.583			.035
	.082										.082
	.150	-.078	-.017	.031	.088	.148	.278	.448			.150
	.250										.250
	.350	-.121	-.063	-.029	.020	.074	.209	.350			.350
	.450	-.140	-.088	-.046	.008	.061	.162	.294			.450
	.550	-.189	-.135	-.090	-.042	.007	.114	.259			.550
	.650	-.211	-.163	-.123	-.074	-.027	.091	.230			.650
	.750	-.200	-.150	-.107	-.061	-.012	.116	.237			.750
	.850	-.181	-.132	-.091	-.045	.001	.122	.228			.850
	.900	-.185	-.139	-.098	-.053	-.004	.105	.208			.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 888 lb/sq ft										
Upper surface	.000	.562	.612	.633	.609	.632	.600	.502			.000
	.032	.332	.235	.143	.067	-.043	-.275	-.473			.032
	.078	.263	.193	.123	.064	.004	-.150	-.280			.078
	.150	.154	.090	.040	-.008	-.063	-.147	-.247			.150
	.250	.106	.048	-.006	-.062	-.108	-.186	-.253			.250
	.350	.091	.034	-.016	-.058	-.110	-.196	-.259			.350
	.450	.049	.001	-.045	-.090	-.134	-.208	-.272			.450
	.550	-.006	-.055	-.099	-.142	-.181	-.250	-.313			.550
	.650	-.008	-.064	-.112	-.153	-.192	-.260	-.321			.650
	.760	-.034	-.076	-.114	-.151	-.190	-.258	-.330			.760
	.840	-.092	-.135	-.173	-.208	-.243	-.310	-.363			.840
	.926	-.079	-.124	-.165	-.199	-.238	-.309	-.367			.926
Lower surface	.035	-.182	-.092	-.001	.080	.161	.348	.539			.035
	.082										.082
	.150	-.065	-.015	.039	.085	.139	.260	.430			.150
	.250										.250
	.350	-.094	-.050	-.004	.046	.100	.221	.358			.350
	.450	-.113	-.065	-.018	.029	.079	.187	.305			.450
	.550	-.150	-.111	-.069	-.023	.027	.142	.257			.550
	.650	-.179	-.135	-.092	-.044	.012	.107	.221			.650
	.750	-.169	-.124	-.080	-.035	.010	.105	.228			.750
	.850	-.138	-.115	-.081	-.039	.004	.105	.227			.850
	.900	-.160	-.117	-.092	-.058	-.016	.091	.220			.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 627 lb/sq ft										
Upper surface	.000	-.381	-.042	.325	.413	.354	-.195	-.708	-1.040	-1.163	.000
	.025	.306	.211	.070	-.116	-.355	-1.204	-1.379	-1.331	-1.117	.025
	.076	.160	.067	-.050	-.177	-.331	-.645	-1.211	-1.353	-1.098	.076
	.150										.150
	.250	.002	-.073	-.151	-.237	-.328	-.537	-.881	-1.233	-1.029	.250
	.350	-.050	-.118	-.188	-.263	-.344	-.527	-.649	-1.076	-.969	.350
	.450	-.074	-.134	-.196	-.259	-.330	-.474	-.501	-.900	-.892	.450
	.550	-.070	-.120	-.171	-.224	-.279	-.379	-.447	-.730	-.820	.550
	.650	-.066	-.108	-.149	-.190	-.234	-.301	-.396	-.612	-.764	.650
	.750	-.045	-.076	-.107	-.137	-.167	-.210	-.298	-.525	-.720	.750
	.850	-.020	-.038	-.058	-.077	-.094	-.115	-.192	-.442	-.685	.850
	.926	.013	.004	-.006	-.014	-.022	-.031	-.094	-.338	-.634	.926
Lower surface	.023	-.630	-.441	-.195	.014	.171	.386	.493	.550	.591	.023
	.072	-.535	-.261	-.112	.001	.100	.281	.409	.499	.577	.072
	.150	-.427	-.204	-.091	-.004	.076	.226	.346	.435	.521	.150
	.250	-.301	-.185	-.101	-.032	.034	.166	.277	.360	.445	.250
	.350	-.251	-.172	-.099	-.040	.018	.135	.234	.309	.389	.350
	.450	-.227	-.166	-.104	-.053	-.001	.103	.191	.255	.327	.450
	.550	-.188	-.139	-.088	-.043	.000	.089	.165	.218	.280	.550
	.650	-.141	-.103	-.062	-.026	.010	.085	.144	.183	.231	.650
	.750	-.083	-.056	-.024	.004	.031	.091	.133	.149	.183	.750
	.860	-.030	-.014	.006	.026	.043	.085	.104	.093	.100	.860
	.900	-.009	.005	.019	.034	.047	.076	.083	.046	.031	.900
	M = 0.900; q = 713 lb/sq ft										
Upper surface	.000	-.276	.056	.331	.419	.390	.038	-.403	-.739	-.886	.000
	.025	.300	.203	.077	-.095	-.302	-1.008	-1.301	-1.372	-.985	.025
	.076	.156	.059	-.048	-.174	-.307	-.582	-.993	-1.375	-1.002	.076
	.150										.150
	.250	-.011	-.089	-.170	-.252	-.331	-.530	-.728	-1.068	-.998	.250
	.350	-.072	-.147	-.225	-.313	-.371	-.533	-.690	-.949	-.948	.350
	.450	-.101	-.171	-.251	-.350	-.432	-.573	-.687	-.851	-.886	.450
	.550	-.096	-.156	-.218	-.291	-.424	-.566	-.652	-.778	-.828	.550
	.650	-.092	-.142	-.193	-.255	-.392	-.572	-.623	-.695	-.776	.650
	.750	-.066	-.103	-.137	-.175	-.191	-.509	-.503	-.628	-.735	.750
	.850	-.034	-.055	-.074	-.092	-.092	-.160	-.271	-.571	-.709	.850
	.926	.008	-.003	-.010	-.017	-.011	-.042	-.166	-.479	-.664	.926
Lower surface	.023	-.676	-.406	-.207	-.002	.153	.376	.503	.573	.629	.023
	.072	-.514	-.260	-.128	-.013	.089	.272	.410	.510	.602	.072
	.150	-.413	-.221	-.113	-.021	.063	.219	.345	.451	.545	.150
	.250	-.341	-.218	-.129	-.052	.021	.158	.274	.373	.469	.250
	.350	-.310	-.216	-.134	-.065	.001	.125	.230	.322	.414	.350
	.450	-.327	-.223	-.144	-.080	-.021	.091	.185	.267	.354	.450
	.550	-.297	-.193	-.126	-.070	-.019	.078	.157	.229	.307	.550
	.650	-.197	-.140	-.090	-.046	-.005	.076	.135	.194	.264	.650
	.750	-.109	-.077	-.042	-.009	.023	.084	.121	.162	.222	.750
	.860	-.039	-.023	-.002	.020	.041	.078	.086	.099	.144	.860
	.900	-.013	.000	.015	.031	.046	.068	.057	.047	.080	.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 743 lb/sq ft										
Upper surface	.000	-.200	.091	.326	.419	.407	.114	-.291	-.631	-.885	.000
	.025	.302	.204	.085	-.074	-.270	-.884	-1.370	-1.292	-1.123	.025
	.076	.155	.061	-.044	-.161	-.287	-.531	-1.177	-1.280	-1.118	.076
	.150										.150
	.250	-.016	-.095	-.167	-.242	-.331	-.499	-.632	-1.233	-1.055	.250
	.350	-.083	-.165	-.236	-.298	-.366	-.507	-.632	-.973	-1.008	.350
	.450	-.119	-.212	-.293	-.361	-.415	-.548	-.647	-.769	-.947	.450
	.550	-.118	-.191	-.281	-.360	-.418	-.544	-.640	-.701	-.885	.550
	.650	-.114	-.183	-.280	-.376	-.431	-.560	-.635	-.670	-.823	.650
	.750	-.084	-.129	-.170	-.348	-.403	-.539	-.598	-.632	-.773	.750
	.850	-.045	-.065	-.077	-.155	-.349	-.508	-.546	-.585	-.739	.850
	.926	.005	-.002	-.004	-.005	-.076	-.223	-.299	-.474	-.664	.926
Lower surface	.023	-.671	-.419	-.208	-.012	.147	.374	.508	.589	.649	.023
	.072	-.476	-.260	-.135	-.023	.084	.269	.414	.524	.620	.072
	.150	-.389	-.232	-.122	-.033	.057	.214	.348	.461	.564	.150
	.250	-.333	-.217	-.145	-.068	.011	.151	.277	.386	.491	.250
	.350	-.312	-.228	-.155	-.087	-.013	.116	.232	.336	.435	.350
	.450	-.330	-.263	-.193	-.114	-.041	.076	.184	.281	.377	.450
	.550	-.333	-.278	-.179	-.102	-.040	.060	.156	.243	.333	.550
	.650	-.338	-.270	-.121	-.071	-.025	.052	.133	.210	.290	.650
	.750	-.302	-.116	-.053	-.024	.005	.056	.117	.179	.250	.750
	.860	-.093	-.016	-.003	.011	.022	.038	.075	.119	.177	.860
	.900	-.014	.010	.016	.025	.025	.015	.037	.066	.115	.900
	M = 0.980; q = 771 lb/sq ft										
Upper surface	.000	-.158	.114	.346	.438	.432	.192	-.202			.000
	.025	.312	.221	.112	-.030	-.220	-.749	-1.254			.025
	.076	.172	.079	-.015	-.122	-.244	-.469	-1.144			.076
	.150										.150
	.250	-.007	-.070	-.137	-.222	-.296	-.449	-.555			.250
	.350	-.082	-.147	-.201	-.270	-.336	-.460	-.566			.350
	.450	-.141	-.210	-.266	-.328	-.388	-.502	-.601			.450
	.550	-.144	-.213	-.273	-.333	-.393	-.497	-.602			.550
	.650	-.165	-.241	-.298	-.357	-.414	-.517	-.617			.650
	.750	-.156	-.229	-.282	-.337	-.390	-.496	-.600			.750
	.850	-.160	-.229	-.282	-.334	-.389	-.490	-.596			.850
	.926	-.139	-.188	-.232	-.267	-.319	-.396	-.521			.926
Lower surface	.023	-.622	-.428	-.185	.004	.153	.384	.525			.023
	.072	-.425	-.210	-.109	-.006	.093	.280	.429			.072
	.150	-.334	-.201	-.099	-.017	.065	.225	.365			.150
	.250	-.302	-.200	-.123	-.057	.013	.159	.292			.250
	.350	-.292	-.211	-.137	-.078	-.019	.119	.246			.350
	.450	-.311	-.242	-.178	-.128	-.064	.075	.196			.450
	.550	-.320	-.258	-.199	-.149	-.085	.054	.166			.550
	.650	-.327	-.270	-.212	-.157	-.078	.043	.140			.650
	.750	-.303	-.249	-.193	-.136	-.046	.043	.124			.750
	.860	-.293	-.239	-.190	-.121	-.037	.020	.079			.860
	.900	-.278	-.217	-.169	-.100	-.048	-.010	.038			.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 803 lb/sq ft										
Upper surface	.000	-.070	.178	.387	.474	.477	.275	-.092			.000
	.025	.363	.267	.160	.028	-.144	-.622	-1.114			.025
	.076	.223	.129	.037	-.067	-.176	-.374	-1.020			.076
	.150										.150
	.250	.056	-.022	-.093	-.165	-.233	-.367	-.456			.250
	.350	-.017	-.081	-.149	-.210	-.266	-.377	-.471			.350
	.450	-.080	-.147	-.211	-.268	-.315	-.419	-.508			.450
	.550	-.081	-.149	-.218	-.274	-.318	-.417	-.509			.550
	.650	-.116	-.177	-.242	-.295	-.342	-.441	-.531			.650
	.750	-.107	-.166	-.226	-.276	-.321	-.421	-.514			.750
	.850	-.113	-.167	-.226	-.277	-.317	-.417	-.511			.850
	.926	-.118	-.161	-.211	-.248	-.279	-.352	-.462			.926
Lower surface	.023	-.526	-.373	-.148	.047	.193	.427	.566			.023
	.072	-.342	-.155	-.048	.036	.138	.326	.473			.072
	.150	-.253	-.140	-.055	.022	.112	.273	.410			.150
	.250	-.228	-.145	-.073	-.011	.065	.207	.340			.250
	.350	-.217	-.159	-.089	-.024	.037	.168	.295			.350
	.450	-.234	-.185	-.126	-.069	-.010	.123	.246			.450
	.550	-.244	-.199	-.145	-.092	-.032	.101	.216			.550
	.650	-.251	-.209	-.159	-.102	-.040	.090	.193			.650
	.750	-.233	-.189	-.140	-.082	-.026	.094	.180			.750
	.860	-.224	-.179	-.136	-.075	-.014	.077	.140			.860
	.900	-.227	-.172	-.132	-.069	-.017	.051	.102			.900
	M = 1.125; q = 856 lb/sq ft										
Upper surface	.000	.071	.297	.419	.495	.499	.359	.080			.000
	.025	.356	.261	.178	.037	-.096	-.586	-.863			.025
	.076	.224	.135	.059	-.041	-.137	-.443	-.803			.076
	.150										.150
	.250	.064	-.008	-.065	-.133	-.193	-.307	-.368			.250
	.350	.002	-.061	-.108	-.164	-.216	-.319	-.381			.350
	.450	-.060	-.120	-.164	-.216	-.264	-.351	-.408			.450
	.550	-.064	-.129	-.171	-.220	-.265	-.345	-.412			.550
	.650	-.086	-.149	-.193	-.243	-.286	-.367	-.442			.650
	.750	-.067	-.123	-.165	-.213	-.259	-.348	-.422			.750
	.850	-.077	-.131	-.171	-.217	-.260	-.342	-.413			.850
	.926	-.085	-.133	-.170	-.212	-.249	-.316	-.388			.926
Lower surface	.023	-.471	-.377	-.126	.058	.198	.405	.572			.023
	.072	-.250	-.115	-.036	.062	.145	.314	.481			.072
	.150	-.178	-.091	-.028	.045	.115	.273	.426			.150
	.250	-.177	-.085	-.021	.043	.099	.219	.364			.250
	.350	-.164	-.094	-.048	.012	.066	.189	.326			.350
	.450	-.180	-.123	-.078	-.028	.024	.148	.282			.450
	.550	-.189	-.137	-.093	-.043	.010	.126	.259			.550
	.650	-.196	-.141	-.098	-.048	.003	.113	.243			.650
	.750	-.185	-.130	-.086	-.036	.009	.121	.240			.750
	.860	-.177	-.128	-.086	-.040	.004	.124	.218			.860
	.900	-.183	-.132	-.093	-.048	.001	.111	.190			.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 888 lb/sq f										
Upper surface	.000	.168	.332	.457	.527	.520	.422	.171			.000
	.025	.366	.284	.182	.069	-.062	-.475	-.723			.025
	.076	.235	.153	.065	-.016	-.113	-.402	-.670			.076
	.150										.150
	.250	.083	.016	-.047	-.102	-.161	-.272	-.322			.250
	.350	.027	-.030	-.080	-.131	-.190	-.295	-.352			.350
	.450	-.028	-.083	-.136	-.187	-.236	-.319	-.380			.450
	.550	-.038	-.099	-.153	-.204	-.249	-.319	-.377			.550
	.650	-.066	-.117	-.165	-.207	-.250	-.334	-.406			.650
	.750	-.055	-.104	-.147	-.192	-.237	-.315	-.387			.750
	.850	-.057	-.109	-.154	-.193	-.236	-.316	-.382			.850
	.926	-.064	-.109	-.152	-.192	-.234	-.302	-.362			.926
Lower surface	.023	-.575	-.367	-.105	.069	.191	.398	.565			.023
	.072	-.180	-.086	-.027	.060	.154	.311	.472			.072
	.150	-.145	-.078	-.005	.062	.130	.269	.421			.150
	.250	-.148	-.087	-.028	.033	.098	.220	.361			.250
	.350	-.142	-.082	-.033	.015	.071	.196	.327			.350
	.450	-.164	-.115	-.064	-.012	.038	.157	.280			.450
	.550	-.182	-.135	-.087	-.036	.022	.136	.249			.550
	.650	-.183	-.137	-.088	-.037	.010	.116	.229			.650
	.750	-.154	-.130	-.086	-.036	.014	.118	.234			.750
	.860	-.139	-.099	-.055	-.018	.015	.103	.240			.860
	.900	-.145	-.104	-.063	-.018	.028	.115	.226			.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 627 lb/sq ft										
Upper surface	.000	.194	.243	.316	.239	-.457	-1.217	-1.294	-.979	-.881	.000
	.020	.354	.270	.127	-.115	-.451	-1.196	-1.272	-.992	-.873	.020
	.073	.177	.079	-.044	-.213	-.412	-1.121	-1.201	-1.008	-.865	.073
	.150	.099	.011	-.099	-.222	-.370	-1.042	-1.129	-1.003	-.837	.150
	.250	.028	-.048	-.139	-.237	-.351	-.769	-1.137	-.950	-.815	.250
	.350	-.022	-.088	-.163	-.248	-.340	-.528	-1.033	-.901	-.792	.350
	.440	-.043	-.104	-.169	-.240	-.315	-.428	-.960	-.858	-.777	.440
	.550	-.046	-.098	-.151	-.209	-.267	-.343	-.824	-.811	-.756	.550
	.650	-.049	-.088	-.132	-.176	-.221	-.273	-.714	-.767	-.735	.650
	.750	-.038	-.070	-.100	-.131	-.162	-.195	-.579	-.723	-.715	.750
	.850	-.014	-.035	-.055	-.075	-.093	-.119	-.455	-.684	-.698	.850
	.923	.024	.010	-.000	-.011	-.021	-.035	-.304	-.632	-.675	.923
Lower surface	.024	-.787	-.568	-.329	-.020	.185	.384	.472	.505	.524	.024
	.073	-.764	-.431	-.143	-.014	.109	.282	.398	.466	.526	.073
	.150	-.801	-.359	-.132	-.025	.059	.208	.319	.392	.463	.150
	.250	-.532	-.230	-.116	-.038	.034	.160	.259	.327	.398	.250
	.350	-.211	-.170	-.097	-.032	.027	.137	.221	.282	.346	.350
	.450	-.147	-.140	-.084	-.029	.020	.115	.185	.234	.289	.450
	.550	-.109	-.097	-.053	-.009	.032	.110	.164	.200	.245	.550
	.650	-.071	-.056	-.022	.013	.043	.109	.145	.163	.195	.650
	.750	-.023	-.011	.014	.041	.065	.114	.131	.127	.143	.750
	.850	.015	.023	.039	.058	.075	.108	.100	.054	.052	.850
	.900	.030	.036	.048	.064	.075	.102	.077	.001	-.015	.900
	M = 0.900; q = 713 lb/sq ft										
Upper surface	.000	.214	.261	.314	.256	-.229	-1.425	-1.214	-.808	-.904	.000
	.020	.335	.251	.121	-.101	-.420	-1.441	-1.229	-.819	-.944	.020
	.073	.157	.053	-.064	-.230	-.414	-1.158	-1.190	-.845	-.872	.073
	.150	.076	-.017	-.129	-.259	-.387	-.664	-1.163	-.870	-.850	.150
	.250	.005	-.077	-.175	-.290	-.414	-.619	-1.120	-.876	-.831	.250
	.350	-.045	-.121	-.206	-.328	-.433	-.623	-1.055	-.865	-.811	.350
	.440	-.068	-.138	-.213	-.301	-.478	-.636	-.932	-.831	-.794	.440
	.550	-.071	-.130	-.192	-.272	-.432	-.644	-.827	-.795	-.773	.550
	.650	-.069	-.117	-.165	-.216	-.223	-.639	-.737	-.764	-.754	.650
	.750	-.053	-.087	-.119	-.152	-.159	-.253	-.655	-.735	-.732	.750
	.850	-.022	-.045	-.064	-.082	-.089	-.075	-.583	-.707	-.717	.850
	.923	.019	.008	-.001	-.009	-.011	-.001	-.485	-.672	-.699	.923
Lower surface	.024	-.815	-.602	-.392	-.055	.150	.368	.467	.524	.553	.024
	.073	-.730	-.431	-.176	-.042	.084	.265	.384	.474	.546	.073
	.150	-.792	-.376	-.173	-.054	.033	.191	.306	.399	.484	.150
	.250	-.657	-.309	-.159	-.069	.008	.143	.246	.334	.417	.250
	.350	-.377	-.234	-.133	-.058	.006	.121	.210	.288	.369	.350
	.450	-.262	-.192	-.115	-.053	.001	.101	.174	.240	.318	.450
	.550	-.163	-.130	-.076	-.026	.018	.098	.154	.208	.274	.550
	.650	-.099	-.074	-.037	.002	.037	.101	.133	.173	.230	.650
	.750	-.038	-.020	.006	.036	.063	.109	.119	.137	.182	.750
	.850	.010	.021	.038	.059	.078	.107	.080	.070	.103	.850
	.900	.028	.036	.050	.066	.081	.101	.051	.020	.044	.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 743 lb/sq ft										
Upper surface	.000	.224	.264	.307	.280	-.109	-1.291	-1.186	-1.071	-.870	.000
	.020	.326	.241	.121	-.074	-.372	-1.285	-1.199	-1.039	-.884	.020
	.073	.143	.040	-.070	-.215	-.404	-1.089	-1.170	-.990	-.903	.073
	.150	.060	-.037	-.143	-.246	-.371	-.585	-1.151	-.924	-.927	.150
	.250	-.013	-.107	-.197	-.302	-.388	-.592	-1.101	-.857	-.906	.250
	.350	-.065	-.150	-.266	-.335	-.416	-.599	-1.052	-.796	-.880	.350
	.440	-.092	-.176	-.282	-.384	-.454	-.611	-.979	-.796	-.863	.440
	.550	-.091	-.167	-.254	-.395	-.467	-.614	-.838	-.808	-.845	.550
	.650	-.087	-.139	-.184	-.402	-.473	-.631	-.762	-.789	-.819	.650
	.750	-.063	-.096	-.120	-.232	-.465	-.622	-.680	-.762	-.794	.750
	.850	-.024	-.046	-.062	-.048	-.162	-.396	-.613	-.745	-.777	.850
	.923	.025	.010	.002	.016	-.012	-.142	-.512	-.718	-.753	.923
Lower surface	.024	-.790	-.601	-.453	-.088	.126	.356	.469	.536	.572	.024
	.073	-.669	-.438	-.199	-.069	.063	.251	.382	.484	.562	.073
	.150	-.629	-.362	-.206	-.088	.016	.176	.303	.410	.500	.150
	.250	-.631	-.345	-.220	-.108	-.015	.125	.242	.345	.437	.250
	.350	-.452	-.307	-.180	-.093	-.016	.101	.207	.299	.390	.350
	.450	-.388	-.317	-.157	-.083	-.020	.076	.169	.254	.339	.450
	.550	-.354	-.247	-.092	-.048	-.002	.071	.148	.222	.298	.550
	.650	-.299	-.079	-.044	-.011	.021	.069	.127	.187	.255	.650
	.750	-.099	-.006	.005	.030	.051	.074	.112	.152	.212	.750
	.850	.013	.032	.040	.059	.070	.063	.076	.092	.136	.850
	.900	.040	.045	.053	.069	.075	.053	.049	.045	.083	.900
	M = 0.980; q = 771 lb/sq ft										
Upper surface	.000	.247	.285	.331	.315	.007	-1.122	-1.145			.000
	.020	.327	.252	.142	-.031	-.306	-1.127	-1.198			.020
	.073	.141	.050	-.039	-.186	-.367	-.955	-1.073			.073
	.150	.053	-.030	-.114	-.217	-.342	-.537	-.998			.150
	.250	-.032	-.104	-.185	-.268	-.365	-.550	-.948			.250
	.350	-.088	-.176	-.234	-.312	-.394	-.548	-.918			.350
	.440	-.125	-.209	-.283	-.356	-.421	-.566	-.879			.440
	.550	-.151	-.231	-.304	-.370	-.444	-.568	-.834			.550
	.650	-.164	-.248	-.320	-.387	-.455	-.587	-.779			.650
	.750	-.179	-.262	-.323	-.389	-.455	-.580	-.720			.750
	.850	-.149	-.231	-.308	-.375	-.437	-.562	-.677			.850
	.923	-.024	-.052	-.111	-.169	-.266	-.398	-.498			.923
Lower surface	.024	-.779	-.585	-.468	-.088	.114	.357	.483			.024
	.073	-.614	-.408	-.160	-.063	.054	.252	.394			.073
	.150	-.523	-.327	-.180	-.092	-.002	.174	.314			.150
	.250	-.491	-.309	-.212	-.137	-.049	.120	.252			.250
	.350	-.431	-.282	-.204	-.130	-.058	.094	.216			.350
	.450	-.398	-.300	-.231	-.159	-.070	.065	.177			.450
	.550	-.368	-.289	-.220	-.145	-.055	.056	.156			.550
	.650	-.342	-.267	-.201	-.131	-.036	.047	.134			.650
	.750	-.299	-.228	-.164	-.085	-.010	.044	.116			.750
	.850	-.227	-.157	-.084	-.006	-.001	.016	.073			.850
	.900	-.138	-.069	-.018	.018	-.000	-.007	.044			.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 803 lb/sq ft										
Upper surface	.000	.307	.333	.373	.374	.105	-.968	-1.156			.000
	.020	.378	.300	.186	.037	-.220	-.989	-1.200			.020
	.073	.198	.105	.008	-.124	-.284	-.823	-1.109			.073
	.150	.113	.029	-.061	-.157	-.253	-.457	-.975			.150
	.250	.028	-.036	-.129	-.208	-.293	-.470	-.793			.250
	.350	-.043	-.109	-.176	-.250	-.322	-.462	-.704			.350
	.440	-.073	-.146	-.224	-.287	-.348	-.480	-.647			.440
	.550	-.105	-.167	-.241	-.308	-.368	-.488	-.624			.550
	.650	-.117	-.187	-.258	-.322	-.384	-.508	-.621			.650
	.750	-.140	-.198	-.262	-.327	-.381	-.505	-.614			.750
	.850	-.127	-.190	-.256	-.315	-.367	-.487	-.602			.850
	.923	-.095	-.137	-.185	-.236	-.283	-.374	-.471			.923
Lower surface	.024	-.689	-.530	-.440	-.043	.157	.396	.523			.024
	.073	-.515	-.345	-.114	-.012	.105	.294	.436			.073
	.150	-.406	-.263	-.124	-.037	.051	.218	.357			.150
	.250	-.360	-.243	-.148	-.079	-.004	.165	.295			.250
	.350	-.337	-.216	-.146	-.075	-.006	.140	.260			.350
	.450	-.324	-.238	-.175	-.105	-.031	.112	.223			.450
	.550	-.302	-.229	-.167	-.093	-.021	.105	.201			.550
	.650	-.279	-.209	-.150	-.079	-.017	.099	.180			.650
	.750	-.241	-.172	-.115	-.048	.012	.099	.162			.750
	.850	-.205	-.137	-.081	-.022	.022	.075	.119			.850
	.900	-.166	-.103	-.055	-.008	.019	.051	.086			.900
	M = 1.125; q = 856 lb/sq ft										
Upper surface	.000	.347	.375	.420	.406	.221	-.697	-.970			.000
	.020	.372	.292	.207	.079	-.138	-.758	-.951			.020
	.073	.193	.099	.025	-.084	-.221	-.643	-.904			.073
	.150	.125	.037	-.034	-.120	-.199	-.376	-.859			.150
	.250	.049	-.020	-.089	-.164	-.247	-.401	-.818			.250
	.350	-.010	-.076	-.135	-.208	-.277	-.413	-.597			.350
	.440	-.055	-.120	-.172	-.235	-.296	-.419	-.544			.440
	.550	-.078	-.143	-.193	-.247	-.309	-.411	-.505			.550
	.650	-.098	-.160	-.213	-.275	-.328	-.426	-.499			.650
	.750	-.106	-.172	-.217	-.271	-.328	-.432	-.500			.750
	.850	-.112	-.168	-.212	-.268	-.321	-.414	-.488			.850
	.923	-.085	-.149	-.192	-.242	-.286	-.356	-.447			.923
Lower surface	.024	-.730	-.539	-.407	-.046	.156	.396	.539			.024
	.073	-.435	-.248	-.097	.014	.123	.300	.453			.073
	.150	-.329	-.193	-.093	-.006	.083	.223	.379			.150
	.250	-.276	-.185	-.097	-.022	.034	.163	.325			.250
	.350	-.265	-.160	-.100	-.035	.020	.156	.298			.350
	.450	-.258	-.175	-.125	-.070	-.013	.135	.269			.450
	.550	-.244	-.162	-.114	-.057	.001	.137	.256			.550
	.650	-.229	-.154	-.107	-.047	.009	.135	.244			.650
	.750	-.198	-.129	-.080	-.020	.037	.150	.237			.750
	.850	-.151	-.088	-.044	.008	.056	.143	.205			.850
	.900	-.134	-.069	-.030	.019	.062	.130	.178			.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 888 lb/sq ft										
Upper surface	.000	.404	.421	.463	.451	.267	-.521	-.830			.000
	.020	.407	.338	.254	.108	-.080	-.597	-.817			.020
	.073	.229	.151	.068	-.031	-.185	-.525	-.767			.073
	.150	.152	.079	-.002	-.079	-.161	-.448	-.733			.150
	.250	.076	.005	-.058	-.129	-.206	-.369	-.708			.250
	.350	.018	-.045	-.102	-.165	-.233	-.359	-.666			.350
	.440	-.019	-.081	-.136	-.195	-.261	-.380	-.522			.440
	.550	-.052	-.106	-.158	-.216	-.268	-.376	-.462			.550
	.650	-.076	-.136	-.191	-.241	-.291	-.381	-.448			.650
	.750	-.095	-.145	-.195	-.247	-.300	-.378	-.452			.750
	.850	-.095	-.144	-.193	-.238	-.285	-.370	-.452			.850
	.923	-.083	-.131	-.172	-.214	-.254	-.335	-.416			.923
Lower surface	.024	-.670	-.570	-.352	-.023	.164	.417	.551			.024
	.073	-.433	-.242	-.040	.039	.149	.326	.458			.073
	.150	-.277	-.154	-.061	.025	.100	.247	.377			.150
	.250	-.220	-.154	-.071	.006	.070	.183	.315			.250
	.350	-.208	-.134	-.074	-.012	.046	.167	.288			.350
	.450	-.224	-.152	-.096	-.045	.008	.123	.258			.450
	.550	-.229	-.165	-.109	-.051	.006	.119	.248			.550
	.650	-.202	-.160	-.110	-.053	.002	.109	.259			.650
	.750	-.165	-.117	-.073	-.034	.012	.122	.264			.750
	.850	-.136	-.080	-.030	.014	.057	.167	.244			.850
	.900	-.126	-.072	-.023	.025	.069	.169	.224			.900

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 627 lb/sq ft										
Upper surface	.000	.130	.349	.459	.363	-.359	-1.029	-.949	-.852	-.804	.000
	.025	.386	.303	.166	-.134	-.540	-.998	-.818	-.707	-.694	.025
	.075	.249	.153	.027	-.170	-.409	-.965	-.796	-.689	-.683	.075
	.150	.156	.067	-.041	-.192	-.376	-.934	-.786	-.679	-.679	.150
	.250	.091	.012	-.075	-.185	-.321	-.911	-.761	-.665	-.672	.250
	.350	.042	-.024	-.103	-.197	-.301	-.864	-.723	-.649	-.664	.350
	.450										.450
	.550	-.009	-.060	-.117	-.180	-.244	-.678	-.649	-.618	-.648	.550
	.650	-.026	-.066	-.114	-.165	-.215	-.518	-.613	-.600	-.637	.650
	.750	-.023	-.053	-.087	-.125	-.157	-.345	-.583	-.582	-.624	.750
	.850	-.003	-.027	-.050	-.076	-.093	-.195	-.546	-.564	-.609	.850
	.900	.012	-.009	-.027	-.042	-.056	-.135	-.535	-.554	-.602	.900
	.925										.925
Lower surface	.040	-.641	-.512	-.225	.004	.188	.359	.434	.456	.478	.040
	.091	-.614	-.466	-.156	-.001	.126	.282	.367	.405	.449	.091
	.150	-.602	-.455	-.120	.012	.111	.242	.319	.358	.402	.150
	.250	-.609	-.462	-.089	.003	.078	.192	.259	.292	.338	.250
	.360	-.593	-.273	-.069	.001	.062	.157	.211	.236	.279	.360
	.450	-.525	-.130	-.047	.011	.060	.140	.179	.193	.230	.450
	.550	-.390	-.060	-.022	.024	.063	.126	.144	.145	.175	.550
	.650	-.218	-.017	.008	.045	.073	.122	.117	.104	.126	.650
	.800	.001	.032	.046	.071	.088	.112	.052	.019	.029	.800
	.874	.055	.044	.053	.071	.082	.095	-.013	-.054	-.052	.874
	M = 0.900; q = 713 lb/sq ft										
Upper surface	.000	.149	.357	.443	.356	-.212	-1.316	-1.010	-.881	-.805	.000
	.025	.374	.289	.151	-.161	-.899	-1.559	-.990	-.746	-.724	.025
	.075	.234	.133	.005	-.212	-.445	-1.396	-.763	-.715	-.708	.075
	.150	.141	.045	-.066	-.243	-.475	-1.322	-.755	-.705	-.705	.150
	.250	.076	-.007	-.103	-.239	-.455	-.954	-.741	-.695	-.696	.250
	.350	.028	-.046	-.130	-.246	-.437	-.768	-.722	-.682	-.689	.350
	.450										.450
	.550	-.023	-.077	-.139	-.213	-.256	-.558	-.674	-.659	-.675	.550
	.650	-.036	-.083	-.135	-.194	-.241	-.243	-.647	-.644	-.668	.650
	.750	-.032	-.067	-.102	-.142	-.168	-.153	-.616	-.625	-.658	.750
	.850	-.008	-.035	-.060	-.081	-.093	-.104	-.580	-.606	-.647	.850
	.900	.007	-.014	-.030	-.044	-.049	-.072	-.564	-.593	-.640	.900
	.925										.925
Lower surface	.040	-.737	-.551	-.270	-.021	.158	.333	.415	.457	.492	.040
	.091	-.701	-.496	-.195	-.022	.103	.256	.348	.404	.459	.091
	.150	-.678	-.479	-.154	-.008	.092	.218	.301	.357	.418	.150
	.250	-.673	-.502	-.117	-.014	.062	.173	.242	.295	.355	.250
	.360	-.662	-.339	-.090	-.012	.051	.143	.195	.241	.300	.360
	.450	-.595	-.180	-.062	.002	.053	.130	.161	.199	.253	.450
	.550	-.453	-.093	-.033	.018	.061	.119	.126	.153	.199	.550
	.650	-.251	-.037	.001	.042	.076	.120	.098	.115	.154	.650
	.800	.022	.025	.047	.072	.093	.116	.038	.039	.065	.800
	.874	.080	.040	.055	.075	.089	.104	-.019	-.026	-.006	.874

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 743 lb/sq ft										
Upper surface	.000	.169	.342	.411	.360	-.050	-1.083	-1.077	-.969	-.890	.000
	.025	.351	.261	.129	-.139	-.725	-1.386	-1.038	-.865	-.823	.025
	.075	.211	.104	-.027	-.220	-.411	-1.246	-.856	-.751	-.768	.075
	.150	.120	.019	-.103	-.278	-.449	-1.193	-.863	-.742	-.758	.150
	.250	.055	-.031	-.137	-.307	-.437	-1.079	-.857	-.732	-.750	.250
	.350	.012	-.064	-.161	-.350	-.474	-.712	-.819	-.721	-.740	.350
	.450										.450
	.550	-.030	-.085	-.149	-.354	-.505	-.656	-.752	-.700	-.722	.550
	.650	-.041	-.088	-.144	-.155	-.515	-.675	-.705	-.686	-.711	.650
	.750	-.033	-.069	-.107	-.112	-.222	-.525	-.655	-.672	-.701	.750
	.850	-.009	-.036	-.062	-.069	-.048	-.175	-.613	-.655	-.691	.850
	.900	.006	-.013	-.032	-.037	-.008	-.113	-.587	-.643	-.684	.900
	.925										.925
Lower surface	.040	-.761	-.599	-.334	-.077	.121	.304	.405	.461	.506	.040
	.091	-.729	-.548	-.246	-.065	.072	.227	.335	.409	.475	.091
	.150	-.717	-.518	-.195	-.043	.065	.191	.289	.365	.433	.150
	.250	-.720	-.487	-.142	-.042	.038	.143	.230	.304	.373	.250
	.360	-.708	-.380	-.101	-.032	.030	.112	.185	.251	.319	.360
	.450	-.628	-.236	-.069	-.013	.036	.098	.154	.212	.275	.450
	.550	-.474	-.135	-.036	.010	.047	.087	.121	.167	.222	.550
	.650	-.308	-.063	.000	.039	.068	.088	.096	.128	.178	.650
	.800	-.093	.012	.048	.075	.094	.086	.041	.060	.095	.800
	.874	-.021	.032	.057	.079	.094	.076	-.012	.001	.029	.874
	M = 0.980; q = 771 lb/sq ft										
Upper surface	.000	.207	.344	.413	.371	.075	-.818	-1.290			.000
	.025	.319	.236	.128	-.084	-.498	-1.214	-1.382			.025
	.075	.171	.076	-.026	-.182	-.362	-1.094	-1.328			.075
	.150	.070	-.019	-.111	-.247	-.407	-1.048	-1.280			.150
	.250	-.004	-.083	-.171	-.286	-.399	-.969	-1.231			.250
	.350	-.052	-.154	-.240	-.337	-.429	-.635	-1.209			.350
	.450										.450
	.550	-.128	-.210	-.294	-.386	-.476	-.616	-1.086			.550
	.650	-.143	-.247	-.332	-.409	-.493	-.643	-.947			.650
	.750	-.056	-.222	-.352	-.447	-.520	-.653	-.882			.750
	.850	-.036	-.003	-.139	-.350	-.504	-.644	-.740			.850
	.900	-.021	.020	-.014	-.118	-.289	-.638	-.653			.900
	.925										.925
Lower surface	.040	-.708	-.543	-.365	-.145	.054	.282	.400			.040
	.091	-.674	-.533	-.286	-.154	.009	.205	.324			.091
	.150	-.667	-.508	-.262	-.116	.007	.165	.276			.150
	.250	-.685	-.480	-.250	-.146	-.022	.114	.216			.250
	.360	-.718	-.477	-.267	-.171	-.030	.080	.170			.360
	.450	-.712	-.419	-.246	-.144	-.030	.058	.137			.450
	.550	-.634	-.355	-.221	-.097	-.020	.038	.102			.550
	.650	-.473	-.270	-.151	-.004	-.001	.025	.075			.650
	.800	-.207	-.063	.036	.056	.028	.001	.028			.800
	.874	-.128	.008	.071	.066	.028	-.023	-.012			.874

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 803 lb/sq ft										
Upper surface	.000	.266	.379	.431	.414	.139	-.717	-1.159			.000
	.025	.361	.288	.184	-.017	-.381	-1.091	-1.249			.025
	.075	.218	.135	.028	-.122	-.295	-.951	-1.177			.075
	.150	.120	.043	-.058	-.178	-.320	-.911	-1.132			.150
	.250	.042	-.032	-.122	-.217	-.328	-.850	-1.087			.250
	.350	-.023	-.084	-.170	-.270	-.353	-.548	-1.066			.350
	.450										.450
	.550	-.082	-.158	-.226	-.313	-.408	-.531	-1.042			.550
	.650	-.113	-.189	-.264	-.335	-.407	-.561	-.841			.650
	.750	-.138	-.214	-.289	-.374	-.442	-.571	-.792			.750
	.850	-.128	-.211	-.294	-.374	-.447	-.565	-.718			.850
	.900	-.085	-.164	-.262	-.351	-.428	-.563	-.638			.900
	.925										.925
Lower surface	.040	-.610	-.472	-.363	-.085	.096	.319	.442			.040
	.091	-.575	-.465	-.225	-.094	.045	.244	.369			.091
	.150	-.566	-.437	-.202	-.058	.042	.208	.323			.150
	.250	-.576	-.401	-.187	-.084	-.002	.160	.264			.250
	.360	-.611	-.405	-.204	-.119	-.016	.127	.220			.360
	.450	-.622	-.357	-.188	-.097	-.009	.107	.187			.450
	.550	-.571	-.301	-.171	-.081	-.003	.088	.154			.550
	.650	-.461	-.247	-.133	-.056	.010	.076	.127			.650
	.800	-.263	-.146	-.061	.000	.028	.050	.077			.800
	.874	-.187	-.091	-.022	.012	.023	.026	.043			.874
	M = 1.125; q = 856 lb/sq ft										
Upper surface	.000	.327	.426	.461	.462	.234	-.419	-.880			.000
	.025	.383	.309	.226	.032	-.249	-.851	-.948			.025
	.075	.242	.161	.075	-.059	-.225	-.736	-.920			.075
	.150	.144	.071	-.002	-.113	-.264	-.714	-.889			.150
	.250	.071	-.007	-.062	-.151	-.251	-.681	-.861			.250
	.350	.015	-.062	-.120	-.195	-.275	-.594	-.850			.350
	.450										.450
	.550	-.057	-.128	-.184	-.266	-.336	-.434	-.827			.550
	.650	-.088	-.153	-.210	-.283	-.357	-.448	-.835			.650
	.750	-.110	-.186	-.245	-.306	-.372	-.474	-.684			.750
	.850	-.118	-.193	-.252	-.325	-.381	-.477	-.608			.850
	.900	-.111	-.189	-.252	-.321	-.391	-.483	-.557			.900
	.925										.925
Lower surface	.040	-.503	-.402	-.294	-.017	.132	.333	.480			.040
	.091	-.457	-.355	-.151	-.029	.079	.259	.408			.091
	.150	-.444	-.304	-.132	-.009	.087	.225	.368			.150
	.250	-.432	-.262	-.122	-.030	.040	.176	.315			.250
	.360	-.420	-.260	-.148	-.075	-.007	.153	.279			.360
	.450	-.422	-.241	-.141	-.066	.001	.144	.254			.450
	.550	-.406	-.221	-.137	-.068	.001	.136	.227			.550
	.650	-.364	-.186	-.110	-.038	.021	.135	.206			.650
	.800	-.272	-.127	-.061	-.000	.050	.124	.164			.800
	.874	-.225	-.094	-.034	.021	.063	.108	.136			.874

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Concluded

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.200; q = 888 lb/sq ft										
Upper surface	.000	.330	.410	.454	.489	.282	-.216	-.702			.000
	.025	.390	.321	.227	.074	-.167	-.659	-.832			.025
	.075	.252	.174	.087	-.029	-.184	-.597	-.783			.075
	.150	.156	.088	.009	-.093	-.258	-.588	-.764			.150
	.250	.077	.018	-.052	-.132	-.223	-.562	-.744			.250
	.350	.024	-.039	-.103	-.170	-.254	-.544	-.737			.350
	.450										.450
	.550	-.030	-.097	-.159	-.228	-.283	-.391	-.721			.550
	.650	-.051	-.111	-.179	-.244	-.318	-.414	-.730			.650
	.750	-.084	-.152	-.208	-.274	-.339	-.432	-.732			.750
	.850	-.095	-.160	-.227	-.291	-.354	-.431	-.628			.850
	.900	-.090	-.161	-.227	-.294	-.360	-.438	-.492			.900
	.925										.925
Lower surface	.040	-.601	-.652	-.291	-.041	.132	.323	.466			.040
	.091	-.491	-.515	-.179	-.034	.076	.250	.395			.091
	.150	-.430	-.309	-.127	-.016	.085	.225	.360			.150
	.250	-.370	-.209	-.121	-.027	.044	.169	.320			.250
	.360	-.325	-.201	-.139	-.072	-.005	.132	.299			.360
	.450	-.301	-.186	-.121	-.058	.003	.155	.281			.450
	.550	-.286	-.174	-.107	-.043	.019	.157	.265			.550
	.650	-.267	-.145	-.077	-.013	.052	.162	.255			.650
	.800	-.209	-.098	-.029	.035	.093	.170	.218			.800
	.874	-.176	-.066	.005	.061	.109	.170	.195			.874

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 627 lb/sq ft										
Upper surface	.022	.403	.351	.252	.011	-.671	-.578	-.493	-.505	-.561	.022
	.076	.298	.220	.109	-.080	-.348	-.567	-.487	-.499	-.558	.076
	.150	.217	.147	.050	-.117	-.306	-.548	-.480	-.493	-.553	.150
	.250	.138	.074	-.012	-.134	-.290	-.517	-.467	-.485	-.547	.250
	.370	.077	.022	-.055	-.158	-.281	-.485	-.451	-.478	-.543	.370
	.450	.032	-.014	-.081	-.170	-.268	-.448	-.435	-.468	-.537	.450
	.550	-.001	-.036	-.095	-.164	-.241	-.415	-.423	-.461	-.535	.550
	.600										.600
	.650	-.034	-.053	-.099	-.157	-.217	-.384	-.412	-.455	-.529	.650
	.750	-.055	-.047	-.084	-.127	-.166	-.349	-.399	-.444	-.521	.750
	.850	-.094	-.037	-.064	-.092	-.115	-.324	-.391	-.438	-.512	.850
	.900										.900
Lower surface	.078	-.460	-.471	-.521	-.049	.128	.277	.323	.352	.375	.078
	.150	-.451	-.468	-.411	-.030	.090	.217	.259	.291	.325	.150
	.250	-.441	-.480	-.133	-.012	.076	.172	.201	.231	.266	.250
	.360	-.439	-.498	-.043	-.008	.061	.135	.150	.176	.207	.360
	.450										.450
	.560	-.453	-.419	-.009	.024	.063	.085	.072	.081	.103	.560
	.610	-.454	-.365	.002	.030	.063	.073	.053	.060	.078	.610
	.700	-.452	-.242	.035	.058	.081	.064	.034	.033	.044	.700
	.800	-.429	-.085	.060	.077	.089	.035	-.006	-.013	-.012	.800
	M = 0.900; q = 713 lb/sq ft										
Upper surface	.022	.403	.344	.242	-.020	-.755	-1.419	-.531	-.559	-.609	.022
	.076	.299	.213	.100	-.109	-.391	-1.182	-.525	-.551	-.603	.076
	.150	.215	.141	.039	-.144	-.361	-1.097	-.515	-.542	-.598	.150
	.250	.141	.064	-.027	-.164	-.344	-.959	-.506	-.535	-.594	.250
	.370	.079	.008	-.074	-.194	-.342	-.839	-.498	-.530	-.590	.370
	.450	.032	-.027	-.103	-.206	-.328	-.737	-.487	-.524	-.585	.450
	.550	.000	-.050	-.114	-.197	-.286	-.646	-.480	-.523	-.583	.550
	.600										.600
	.650	-.031	-.067	-.121	-.189	-.245	-.567	-.475	-.521	-.579	.650
	.750	-.047	-.061	-.101	-.149	-.178	-.483	-.467	-.514	-.573	.750
	.850	-.081	-.051	-.079	-.105	-.115	-.402	-.462	-.507	-.565	.850
	.900										.900
Lower surface	.078	-.507	-.489	-.480	-.051	.132	.272	.306	.344	.377	.078
	.150	-.496	-.484	-.419	-.034	.093	.215	.244	.284	.331	.150
	.250	-.487	-.492	-.237	-.017	.079	.175	.190	.228	.273	.250
	.360	-.488	-.506	-.077	-.011	.063	.145	.140	.173	.218	.360
	.450										.450
	.560	-.521	-.449	-.015	.023	.068	.112	.063	.085	.119	.560
	.610	-.523	-.406	-.002	.030	.069	.105	.045	.064	.094	.610
	.700	-.520	-.302	.035	.061	.089	.106	.027	.040	.066	.700
	.800	-.492	-.150	.063	.081	.099	.091	-.010	-.006	.013	.800

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 743 lb/sq ft										
Upper surface	.022	.388	.346	.234	-.080	-.677	-1.342	-.601	-.625	-.663	.022
	.076	.292	.214	.094	-.174	-.452	-1.193	-.586	-.611	-.654	.076
	.150	.209	.124	.029	-.222	-.472	-1.136	-.574	-.602	-.648	.150
	.250	.140	.065	-.034	-.198	-.450	-1.082	-.564	-.595	-.642	.250
	.370	.078	.008	-.081	-.195	-.487	-1.044	-.555	-.589	-.636	.370
	.450	.036	-.030	-.111	-.222	-.532	-.967	-.545	-.583	-.630	.450
	.550	.005	-.053	-.122	-.216	-.552	-.615	-.540	-.583	-.626	.550
	.600										.600
	.650	-.023	-.072	-.129	-.206	-.234	-.513	-.533	-.579	-.622	.650
	.750	-.036	-.066	-.111	-.162	-.092	-.484	-.528	-.576	-.615	.750
	.850	-.065	-.051	-.085	-.113	-.056	-.470	-.523	-.571	-.608	.850
	.900										.900
Lower surface	.078	-.537	-.514	-.462	-.057	.098	.230	.294	.343	.387	.078
	.150	-.514	-.510	-.408	-.037	.067	.176	.233	.288	.345	.150
	.250	-.495	-.512	-.278	-.017	.062	.141	.180	.232	.289	.250
	.360	-.486	-.525	-.113	-.011	.053	.112	.132	.181	.235	.360
	.450										.450
	.560	-.523	-.489	-.021	.025	.067	.083	.062	.097	.142	.560
	.610	-.545	-.462	-.006	.032	.070	.077	.045	.076	.122	.610
	.700	-.551	-.390	.034	.063	.092	.081	.031	.056	.093	.700
	.800	-.509	-.236	.064	.085	.106	.069	-.002	.016	.044	.800
	M = 0.980; q = 771 lb/sq ft										
Upper surface	.022	.341	.260	.171	.013	-.324	-1.182	-.964			.022
	.076	.229	.123	.023	-.124	-.379	-1.065	-.838			.076
	.150	.143	.032	-.057	-.205	-.396	-1.013	-.835			.150
	.250	.084	-.033	-.124	-.250	-.373	-.970	-.822			.250
	.370	.032	-.108	-.200	-.299	-.418	-.946	-.800			.370
	.450	-.005	-.133	-.261	-.352	-.472	-.947	-.772			.450
	.550	-.029	-.046	-.310	-.381	-.516	-.955	-.751			.550
	.600										.600
	.650	-.050	-.058	-.311	-.429	-.540	-.940	-.724			.650
	.750	-.055	-.053	-.043	-.450	-.568	-.756	-.696			.750
	.850	-.075	-.039	.003	-.131	-.563	-.707	-.672			.850
	.900										.900
Lower surface	.078	-.682	-.810	-.578	-.237	-.006	.159	.288			.078
	.150	-.639	-.789	-.501	-.209	-.035	.096	.228			.150
	.250	-.602	-.770	-.396	-.075	-.017	.066	.179			.250
	.360	-.574	-.711	-.279	-.050	-.020	.035	.136			.360
	.450										.450
	.560	-.552	-.503	-.039	.006	.006	.003	.074			.560
	.610	-.570	-.394	.002	.020	.010	-.004	.059			.610
	.700	-.619	-.093	.058	.059	.036	.004	.050			.700
	.800	-.603	.154	.095	.089	.051	-.001	.024			.800

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 803 lb/sq ft										
Upper surface	.022	.358	.302	.224	.086	-.202	-1.028	-1.238			.022
	.076	.244	.168	.076	-.061	-.281	-.916	-1.165			.076
	.150	.153	.077	-.007	-.142	-.317	-.875	-1.110			.150
	.250	.081	.014	-.069	-.179	-.299	-.836	-1.067			.250
	.370	.004	-.061	-.139	-.238	-.339	-.819	-1.050			.370
	.450	-.052	-.124	-.192	-.291	-.387	-.824	-1.042			.450
	.550	-.093	-.158	-.237	-.330	-.430	-.831	-1.041			.550
	.600										.600
	.650	-.127	-.205	-.278	-.359	-.466	-.831	-1.012			.650
	.750	-.148	-.239	-.311	-.396	-.478	-.675	-.833			.750
	.850	-.123	-.252	-.336	-.425	-.511	-.622	-.710			.850
	.900										.900
Lower surface	.078	-.725	-.725	-.632	-.196	-.014	.199	.300			.078
	.150	-.718	-.713	-.556	-.204	-.038	.142	.243			.150
	.250	-.715	-.705	-.477	-.157	-.027	.110	.194			.250
	.360	-.711	-.670	-.354	-.144	-.026	.081	.155			.360
	.450										.450
	.560	-.664	-.535	-.198	-.071	-.010	.050	.099			.560
	.610	-.645	-.475	-.155	-.055	-.007	.042	.086			.610
	.700	-.606	-.345	-.076	-.014	.017	.050	.082			.700
	.800	-.554	-.167	-.023	.011	.029	.044	.061			.800
	M = 1.125; q = 856 lb/sq ft										
Upper surface	.022	.379	.316	.260	.128	-.088	-.772	-.980			.022
	.076	.264	.185	.116	-.012	-.198	-.678	-.913			.076
	.150	.164	.083	.013	-.087	-.260	-.663	-.874			.150
	.250	.118	.038	-.024	-.111	-.223	-.635	-.848			.250
	.370	.050	-.024	-.080	-.162	-.258	-.628	-.834			.370
	.450	-.011	-.074	-.136	-.212	-.302	-.642	-.826			.450
	.550	-.049	-.117	-.175	-.254	-.332	-.653	-.840			.550
	.600										.600
	.650	-.091	-.160	-.214	-.284	-.367	-.668	-.838			.650
	.750	-.118	-.182	-.246	-.312	-.388	-.676	-.840			.750
	.850	-.134	-.209	-.269	-.341	-.411	-.665	-.802			.850
	.900										.900
Lower surface	.078	-.653	-.703	-.581	-.203	-.006	.234	.356			.078
	.150	-.646	-.652	-.448	-.160	-.007	.186	.304			.150
	.250	-.643	-.506	-.338	-.130	-.009	.163	.261			.250
	.360	-.627	-.411	-.265	-.125	-.008	.147	.229			.360
	.450										.450
	.560	-.570	-.318	-.189	-.059	.025	.123	.183			.560
	.610	-.551	-.277	-.159	-.044	.036	.119	.173			.610
	.700	-.508	-.208	-.086	.012	.079	.131	.172			.700
	.800	-.456	-.121	-.010	.063	.102	.132	.157			.800

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 888 lb/sq ft										
Upper surface	.022	.383	.337	.281	.176	-.011	-.573	-.037			.022
	.076	.273	.213	.145	.037	-.127	-.504	-.760			.076
	.150	.187	.123	.058	-.047	-.236	-.511	-.733			.150
	.250	.137	.075	.013	-.071	-.181	-.500	-.710			.250
	.370	.076	.018	-.044	-.120	-.212	-.504	-.702			.370
	.450	.020	-.036	-.096	-.164	-.254	-.524	-.701			.450
	.550	-.021	-.073	-.133	-.202	-.284	-.546	-.713			.550
	.600										.600
	.650	-.060	-.115	-.169	-.238	-.314	-.562	-.714			.650
	.750	-.087	-.150	-.200	-.260	-.338	-.572	-.720			.750
	.850	-.114	-.172	-.229	-.285	-.356	-.581	-.722			.850
	.900										.900
Lower surface	.078	-.758	-.725	-.619	-.295	.002	.225	.365			.078
	.150	-.704	-.689	-.404	-.122	-.003	.182	.315			.150
	.250	-.667	-.615	-.203	-.106	-.008	.155	.279			.250
	.360	-.637	-.458	-.192	-.117	-.016	.137	.258			.360
	.450										.450
	.560	-.477	-.242	-.175	-.087	.003	.136	.222			.560
	.610	-.401	-.219	-.156	-.072	.012	.143	.215			.610
	.700	-.347	-.194	-.116	-.032	.053	.171	.219			.700
	.800	-.299	-.114	-.037	.048	.116	.182	.210			.800

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 627 lb/sq ft										
Upper surface	.073	.315	.284	.210	.054	-.229	-.437	-.335	-.371	-.433	.073
	.150	.205	.184	.110	-.039	-.239	-.420	-.332	-.370	-.432	.150
	.250	.111	.084	.018	-.092	-.246	-.401	-.334	-.371	-.432	.250
	.350	.043	.024	-.033	-.119	-.226	-.369	-.331	-.371	-.432	.350
	.460	-.026	-.029	-.072	-.138	-.213	-.332	-.327	-.373	-.436	.460
	.550	-.078	-.065	-.094	-.144	-.199	-.302	-.324	-.372	-.439	.550
	.660	-.109	-.083	-.087	-.123	-.164	-.272	-.318	-.373	-.443	.660
	.750	-.122	-.086	-.062	-.083	-.118	-.250	-.318	-.374	-.445	.750
	.800										.800
Lower surface	.100	-.226	-.317	-.433	-.133	.057	.168	.214	.245	.277	.100
	.200	-.214	-.294	-.344	-.090	.033	.104	.138	.169	.200	.200
	.250	-.208	-.284	-.321	-.066	.027	.080	.107	.132	.164	.250
	.350	-.197	-.268	-.293	-.039	.019	.037	.052	.069	.094	.350
	.450	-.187	-.251	-.242	-.015	.016	.000	-.000	.003	.024	.450
	.550	-.175	-.239	-.160	.012	.023	-.018	-.036	-.039	-.034	.550
	.600	-.170	-.236	-.108	.020	.026	-.027	-.052	-.059	-.060	.600
	.690	-.160	-.238	-.027	.038	.036	-.039	-.078	-.097	-.111	.690
	M = 0.900; q = 713 lb/sq ft										
Upper surface	.073	.326	.289	.212	.033	-.300	-.518	-.397	-.429	-.485	.073
	.150	.208	.183	.116	-.064	-.286	-.507	-.396	-.429	-.483	.150
	.250	.120	.084	.010	-.122	-.292	-.491	-.400	-.430	-.480	.250
	.350	.044	.018	-.046	-.156	-.290	-.468	-.401	-.429	-.479	.350
	.460	-.034	-.044	-.100	-.179	-.242	-.437	-.403	-.432	-.482	.460
	.550	-.100	-.087	-.126	-.174	-.216	-.408	-.405	-.434	-.484	.550
	.660	-.139	-.102	-.111	-.140	-.171	-.379	-.405	-.437	-.486	.660
	.750	-.152	-.098	-.073	-.090	-.117	-.355	-.409	-.440	-.488	.750
	.800										.800
Lower surface	.100	-.247	-.332	-.443	-.139	.073	.203	.202	.239	.276	.100
	.200	-.234	-.309	-.343	-.094	.042	.132	.127	.168	.207	.200
	.250	-.229	-.298	-.324	-.069	.035	.107	.096	.134	.174	.250
	.350	-.218	-.282	-.298	-.041	.022	.056	.036	.069	.108	.350
	.450	-.208	-.266	-.255	-.017	.011	.003	-.031	-.003	.029	.450
	.550	-.197	-.256	-.180	.012	.020	-.029	-.077	-.060	-.035	.550
	.600	-.192	-.253	-.131	.022	.025	-.041	-.095	-.088	-.064	.600
	.690	-.182	-.255	-.039	.041	.037	-.061	-.131	-.135	-.125	.690

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 743 lb/sq ft										
Upper surface	.073	.340	.295	.211	.013	-.429	-.555	-.465	-.480	-.536	.073
	.150	.235	.179	.119	-.076	-.458	-.535	-.464	-.476	-.533	.150
	.250	.137	.088	.009	-.138	-.464	-.507	-.466	-.474	-.529	.250
	.350	.059	.021	-.054	-.178	-.291	-.480	-.466	-.473	-.527	.350
	.460	-.024	-.048	-.120	-.240	-.156	-.454	-.470	-.473	-.526	.460
	.550	-.096	-.095	-.147	-.189	-.178	-.435	-.471	-.475	-.527	.550
	.660	-.144	-.107	-.122	-.140	-.156	-.418	-.474	-.478	-.527	.660
	.750	-.164	-.103	-.077	-.088	-.106	-.407	-.475	-.479	-.525	.750
	.800										.800
Lower surface	.100	-.297	-.337	-.462	-.137	.080	.162	.196	.247	.290	.100
	.200	-.283	-.315	-.350	-.091	.048	.097	.125	.179	.227	.200
	.250	-.278	-.306	-.329	-.066	.044	.076	.097	.149	.197	.250
	.350	-.267	-.291	-.306	-.046	.026	.025	.040	.087	.135	.350
	.450	-.254	-.278	-.266	-.020	.009	-.040	-.032	.013	.060	.450
	.550	-.243	-.272	-.193	.014	.023	-.079	-.086	-.048	-.007	.550
	.600	-.237	-.270	-.141	.024	.030	-.093	-.107	-.075	-.037	.600
	.690	-.225	-.273	-.043	.044	.042	-.122	-.153	-.134	-.104	.690
	M = 0.980; q = 771 lb/sq ft										
Upper surface	.073	.323	.298	.184	-.018	-.305	-1.048	-1.332			.073
	.150	.229	.196	.111	-.116	-.387	-.951	-1.242			.150
	.250	.127	.099	.019	-.182	-.386	-.927	-1.192			.250
	.350	.058	.035	-.018	-.249	-.427	-.909	-1.155			.350
	.460	-.021	-.039	-.075	-.310	-.463	-.906	-1.079			.460
	.550	-.090	-.090	-.107	-.392	-.518	-.921	-.937			.550
	.660	-.139	-.108	-.084	-.218	-.554	-.906	-.813			.660
	.750	-.151	-.096	-.036	.031	-.523	-.822	-.761			.750
	.800										.800
Lower surface	.100	-.372	-.313	-.333	-.182	-.067	.069	.172			.100
	.200	-.351	-.284	-.292	-.173	-.052	.024	.109			.200
	.250	-.342	-.273	-.285	-.087	-.040	.014	.089			.250
	.350	-.326	-.267	-.289	-.053	-.044	-.014	.043			.350
	.450	-.321	-.264	-.277	-.031	-.086	-.071	-.025			.450
	.550	-.321	-.281	-.242	.021	-.072	-.125	-.086			.550
	.600	-.319	-.287	-.208	.038	-.060	-.142	-.112			.600
	.690	-.326	-.283	-.130	.070	-.042	-.171	-.167			.690

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 803 lb/sq ft										
Upper surface	.073	.299	.245	.171	.039	-.171	-.891	-1.171			.073
	.150	.199	.155	.089	-.030	-.256	-.803	-1.075			.150
	.250	.101	.046	-.013	-.121	-.287	-.784	-1.036			.250
	.350	.031	-.021	-.074	-.188	-.334	-.774	-1.015			.350
	.460	-.054	-.110	-.152	-.236	-.368	-.772	-1.005			.460
	.550	-.142	-.209	-.250	-.330	-.433	-.789	-1.012			.550
	.660	-.218	-.303	-.359	-.422	-.485	-.779	-.994			.660
	.750	-.202	-.303	-.391	-.446	-.489	-.780	-.979			.750
	.800										.800
Lower surface	.100	-.518	-.543	-.867	-.279	-.079	.092	.200			.100
	.200	-.491	-.497	-.697	-.308	-.136	.044	.142			.200
	.250	-.482	-.479	-.579	-.267	-.118	.041	.126			.250
	.350	-.466	-.454	-.413	-.205	-.089	.025	.089			.350
	.450	-.452	-.429	-.315	-.156	-.086	-.022	.030			.450
	.550	-.437	-.414	-.220	-.092	-.090	-.076	-.026			.550
	.600	-.428	-.406	-.185	-.062	-.081	-.096	-.052			.600
	.690	-.408	-.405	-.155	-.054	-.081	-.125	-.102			.690
	M = 1.125; q = 856 lb/sq ft										
Upper surface	.073	.327	.264	.199	.102	-.049	-.652	-.911			.073
	.150	.249	.176	.125	.031	-.143	-.572	-.837			.150
	.250	.154	.092	.036	-.057	-.177	-.572	-.808			.250
	.350	.091	.030	-.030	-.114	-.220	-.566	-.792			.350
	.460	.021	-.023	-.072	-.155	-.256	-.576	-.791			.460
	.550	-.068	-.110	-.152	-.229	-.314	-.593	-.794			.550
	.660	-.161	-.212	-.250	-.317	-.380	-.595	-.783			.660
	.750	-.176	-.246	-.289	-.352	-.394	-.597	-.768			.750
	.800										.800
Lower surface	.100	-.624	-.815	-.756	-.574	-.070	.160	.269			.100
	.200	-.588	-.789	-.728	-.288	-.120	.102	.218			.200
	.250	-.567	-.789	-.715	-.258	-.115	.104	.205			.250
	.350	-.533	-.751	-.536	-.245	-.092	.104	.183			.350
	.450	-.499	-.721	-.372	-.233	-.045	.081	.133			.450
	.550	-.484	-.659	-.303	-.129	-.010	.039	.088			.550
	.600	-.475	-.531	-.232	-.056	.011	.018	.062			.600
	.690	-.468	-.315	-.050	.014	.027	-.011	.015			.690

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Concluded

(f) 95-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 888 lb/sq ft										
Upper surface	.073	.334	.281	.224	.140	-.007	-.482	-.776			.073
	.150	.261	.206	.140	.056	-.132	-.439	-.707			.150
	.250	.166	.114	.062	-.019	-.123	-.442	-.686			.250
	.350	.105	.054	.003	-.072	-.170	-.449	-.671			.350
	.460	.057	.010	-.043	-.112	-.207	-.465	-.670			.460
	.550	-.018	-.060	-.110	-.171	-.256	-.486	-.675			.550
	.660	-.113	-.153	-.199	-.253	-.323	-.495	-.668			.660
	.750	-.147	-.192	-.242	-.291	-.345	-.498	-.658			.750
	.800										.800
Lower surface	.100	-.752	-.745	-.650	-.505	-.070	.180	.300			.100
	.200	-.727	-.717	-.634	-.341	-.110	.116	.238			.200
	.250	-.729	-.706	-.623	-.175	-.102	.121	.236			.250
	.350	-.693	-.680	-.569	-.204	-.091	.124	.223			.350
	.450	-.637	-.653	-.399	-.220	-.084	.110	.186			.450
	.550	-.660	-.640	-.248	-.173	-.022	.100	.156			.550
	.600	-.652	-.623	-.228	-.139	.012	.080	.131			.600
	.690	-.635	-.493	-.125	-.010	.061	.054	.087			.690

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE

OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING

(a) Station A

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.055										.055
.166	.041	.018	.007	-.006	-.023	-.035	-.028	-.029	-.028	.166
.277	.017	.004	-.007	-.028	-.036	-.036	-.036	-.033	-.053	.277
.367	.107	.069	.045	.007	-.002	-.045	-.092	-.138	-.153	.367
.387	.097	.052	.022	-.004	-.037	-.108	-.186	-.263	-.284	.387
.415	.074	.022	-.019	-.060	-.102	-.201	-.312	-.395	-.429	.415
.443	.045	-.007	-.056	-.095	-.151	-.260	-.389	-.426	-.489	.443
.498	.019	-.041	-.088	-.136	-.191	-.301	-.396	-.408	-.433	.498
.553	-.010	-.061	-.104	-.145	-.191	-.285	-.309	-.404	-.416	.553
.581										.581
.609	-.012	-.044	-.065	-.125	-.160	-.207	-.238	-.340	-.385	.609
.636	-.022	-.053	-.080	-.097	-.128	-.155	-.210	-.301	-.386	.636
.664	-.022	-.050	-.065	-.076	-.097	-.114	-.172	-.270	-.408	.664
.692	-.011	-.028	-.036	-.048	-.055	-.065	-.114	-.199	-.369	.692
.719	.004	-.007	-.015	-.016	-.024	-.023	-.060	-.136	-.305	.719
.774	.010	.000	.003	.007	.003	.003	-.016	-.072	-.199	.774
.830	-.004	-.008	-.003	.001	.003	.007	-.004	-.036	-.109	.830
.871	.007	.000	.006	.010	.011	.018	.013	-.007	-.046	.871
.954	.015	.020	.027	.037	.041	.037	.034	.024	.028	.954
M = 0.900; q = 358 lb/sq ft										
.055										.055
.166	.039	.015	.003	-.014	-.024	-.030	-.019	-.014	-.009	.166
.277	.015	.000	-.014	-.034	-.041	-.031	-.013	-.001	-.008	.277
.367	.120	.081	.064	.011	.028	.006	-.022	-.052	-.072	.367
.387	.116	.071	.042	.015	-.004	-.059	-.114	-.179	-.220	.387
.415	.083	.031	-.006	-.047	-.094	-.157	-.234	-.334	-.388	.415
.443	.052	-.007	-.052	-.099	-.151	-.236	-.320	-.405	-.412	.443
.498	.008	-.056	-.105	-.155	-.204	-.306	-.414	-.488	-.445	.498
.553	-.034	-.091	-.156	-.231	-.287	-.377	-.483	-.321	-.486	.553
.581										.581
.609	-.030	-.098	-.139	-.184	-.231	-.438	-.483	-.415	-.450	.609
.636	-.045	-.081	-.111	-.143	-.173	-.451	-.418	-.479	-.441	.636
.664	-.036	-.063	-.084	-.100	-.100	-.243	-.202	-.506	-.442	.664
.692	-.018	-.036	-.044	-.052	-.047	-.035	-.100	-.460	-.452	.692
.719	.000	-.011	-.016	-.020	-.017	.006	-.080	-.169	-.349	.719
.774	.008	-.001	.002	.004	.008	.027	-.050	-.131	-.306	.774
.830	-.012	-.013	-.006	-.003	.004	.013	-.022	-.077	-.202	.830
.871	-.001	-.001	.005	.008	.013	.023	.007	-.035	-.118	.871
.954	.013	.020	.031	.033	.041	.040	.030	.015	-.016	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Continued

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.940; q = 368 lb/sq ft										
.055										.055
.166	.040	.014	.001	-.011	-.020	-.035	-.021	-.004	.001	.166
.277	.012	-.004	-.016	-.036	-.039	-.029	-.005	.018	.014	.277
.367	.130	.095	.071	.023	.043	.026	.009	-.010	-.043	.367
.387	.126	.085	.057	.039	.013	-.035	-.086	-.137	-.196	.387
.415	.092	.040	-.001	-.024	-.071	-.136	-.201	-.289	-.371	.415
.443	.057	-.002	-.050	-.088	-.134	-.214	-.287	-.367	-.461	.443
.498	.005	-.055	-.107	-.145	-.199	-.292	-.389	-.454	-.325	.498
.553	-.048	-.118	-.187	-.227	-.272	-.362	-.447	-.490	-.446	.553
.581										.581
.609	-.043	-.133	-.152	-.281	-.331	-.421	-.492	-.410	-.515	.609
.636	-.055	-.102	-.142	-.284	-.350	-.447	-.509	-.405	-.544	.636
.664	-.048	-.070	-.085	-.179	-.347	-.474	-.483	-.451	-.517	.664
.692	-.019	-.032	-.034	-.027	-.082	-.387	-.404	-.482	-.493	.692
.719	.005	-.005	-.009	.002	.005	-.030	-.055	-.385	-.372	.719
.774	.009	.005	.008	.013	.026	.034	-.039	-.242	-.318	.774
.830	-.011	-.011	-.008	.002	.014	.029	-.032	-.096	-.253	.830
.871	.000	.002	.005	.011	.021	.031	-.006	-.050	-.152	.871
.954	.019	.024	.033	.039	.049	.051	.033	.015	-.038	.954
M = 0.980; q = 385 lb/sq ft										
.055										.055
.166	.044	.023	.004	-.008	-.020	-.024	.000	.017	.031	.166
.277	.020	-.003	-.015	-.030	-.033	-.017	.020	.050	.055	.277
.367	.153	.124	.104	.039	.082	.068	.057	.044	.013	.367
.387	.154	.113	.087	.100	.048	.010	-.031	-.079	-.134	.387
.415	.115	.067	.030	.003	-.037	-.090	-.154	-.229	-.308	.415
.443	.077	.021	-.021	-.063	-.103	-.169	-.234	-.312	-.396	.443
.498	.022	-.036	-.080	-.126	-.174	-.251	-.337	-.405	-.481	.498
.553	-.068	-.121	-.164	-.206	-.245	-.322	-.400	-.453	-.529	.553
.581										.581
.609	-.064	-.123	-.228	-.263	-.311	-.388	-.457	-.510	-.535	.609
.636	-.111	-.181	-.241	-.288	-.334	-.416	-.489	-.544	-.535	.636
.664	-.143	-.205	-.257	-.311	-.362	-.440	-.514	-.578	-.338	.664
.692	-.133	-.201	-.257	-.308	-.354	-.450	-.537	-.589	-.389	.692
.719	-.028	-.053	-.088	-.133	-.206	-.343	-.466	-.351	-.499	.719
.774	-.003	.024	.040	.034	.010	-.073	-.085	-.108	-.202	.774
.830	-.011	.009	.028	.037	.040	.014	-.066	-.161	-.266	.830
.871	.003	.019	.032	.039	.047	.048	-.030	-.165	-.254	.871
.954	.030	.041	.054	.064	.076	.091	.054	-.039	-.096	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Continued

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.055										.055
.166	.083	.069	.053	.041	.026	.012	.026	.048	.073	.166
.277	.024	.025	.010	-.011	-.016	-.021	.061	.102	.112	.277
.367	.197	.159	.129	.014	.108	.135	.122	.113	.090	.367
.387	.204	.167	.139	.173	.115	.079	.038	-.006	-.048	.387
.415	.171	.123	.081	.065	.021	-.013	-.082	-.153	-.218	.415
.443	.132	.075	.028	-.007	-.047	-.092	-.155	-.236	-.309	.443
.498	.076	.024	-.034	-.073	-.118	-.179	-.263	-.323	-.393	.498
.553	-.003	-.056	-.103	-.143	-.182	-.244	-.321	-.376	-.439	.553
.581										.581
.609	-.034	-.074	-.115	-.216	-.243	-.311	-.385	-.428	-.485	.609
.636	-.066	-.123	-.185	-.225	-.270	-.338	-.411	-.467	-.524	.636
.664	-.091	-.142	-.206	-.252	-.296	-.364	-.440	-.507	-.549	.664
.692	-.104	-.149	-.209	-.250	-.294	-.381	-.466	-.532	-.587	.692
.719	-.084	-.108	-.154	-.185	-.207	-.309	-.440	-.528	-.589	.719
.774	-.048	-.039	-.026	-.029	-.052	-.094	-.117	-.017	.007	.774
.830	-.070	-.055	-.048	-.045	-.059	-.069	-.066	-.067	-.186	.830
.871	-.057	-.042	-.042	-.041	-.049	-.048	-.032	-.067	-.140	.871
.954	-.143	-.124	-.085	-.065	-.053	-.033	-.056	-.154	-.205	.954
M = 1.125; q = 421 lb/sq ft										
.055										.055
.166	.063	.042	.031	.013	-.002	-.014	-.026	-.027	-.037	.166
.277	.017	.009	.000	-.027	-.032	-.031	-.034	-.039	-.088	.277
.367	-.007	-.015	-.006	-.012	-.033	-.012	.010	.124	.188	.367
.387	.123	.087	.055	.003	.044	.036	.031	.057	.076	.387
.415	.165	.124	.103	.072	.038	-.022	-.058	-.075	-.088	.415
.443	.135	.089	.055	.032	-.015	-.074	-.118	-.148	-.183	.443
.498	.088	.035	-.007	-.046	-.089	-.149	-.208	-.241	-.276	.498
.553	.029	-.021	-.061	-.102	-.140	-.205	-.250	-.278	-.318	.553
.581										.581
.609	-.018	-.052	-.082	-.159	-.198	-.260	-.309	-.342	-.353	.609
.636	-.038	-.095	-.136	-.180	-.215	-.281	-.327	-.378	-.376	.636
.664	-.056	-.106	-.149	-.196	-.236	-.300	-.361	-.398	-.391	.664
.692	-.067	-.115	-.158	-.201	-.245	-.315	-.377	-.414	-.422	.692
.719	-.066	-.108	-.140	-.179	-.216	-.285	-.369	-.417	-.424	.719
.774	-.039	-.047	-.050	-.054	-.066	-.068	-.083	.056	.112	.774
.830	-.050	-.050	-.042	-.042	-.043	-.034	-.029	.056	-.042	.830
.871	-.039	-.038	-.027	-.024	-.025	-.011	.008	.058	-.004	.871
.954	-.074	-.074	-.052	-.037	-.011	.012	.071	-.016	-.084	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.200; q = 436 lb/sq ft										
.055										.055
.166	.075	.057	.042	.027	.014	-.003	-.003	-.008	-.025	.166
.277	.033	.024	.001	-.002	-.003	-.013	-.012	-.044	-.088	.277
.367	.018	.006	.010	.008	-.011	-.010	-.008	-.053	-.051	.367
.387	.036	.030	.023	.019	.025	.034	.028	.067	.056	.387
.415	.171	.137	.059	.019	.048	.003	-.042	-.071	-.063	.415
.443	.160	.117	.102	.074	.025	-.049	-.099	-.133	-.161	.443
.498	.118	.067	.028	-.014	-.055	-.119	-.173	-.218	-.246	.498
.553	.057	.014	-.026	-.061	-.099	-.164	-.217	-.260	-.284	.553
.581										.581
.609	.006	-.027	-.059	-.129	-.150	-.224	-.271	-.307	-.340	.609
.636	-.023	-.066	-.107	-.139	-.173	-.241	-.293	-.339	-.376	.636
.664	-.035	-.075	-.125	-.168	-.201	-.264	-.311	-.355	-.393	.664
.692	-.054	-.096	-.139	-.172	-.203	-.274	-.336	-.383	-.406	.692
.719	-.062	-.099	-.140	-.175	-.209	-.274	-.342	-.397	-.412	.719
.774	-.048	-.057	-.066	-.069	-.068	-.070	-.076	-.074	-.070	.774
.830	-.039	-.041	-.044	-.038	-.029	-.025	-.014	-.026	.010	.830
.871	-.041	-.034	-.027	-.016	-.008	-.000	.010	.000	.026	.871
.954	-.100	-.082	-.075	-.056	-.036	-.031	-.033	-.050	-.084	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(b) Station B

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.166	.030	.013	.004	-.006	-.020	-.052	-.074	-.110	-.120	.166
.277	.003	-.010	-.013	-.027	-.034	-.059	-.080	-.083	-.073	.277
.367	.102	.074	.050	.026	-.011	-.085	-.167	-.234	-.255	.367
.387	.161	.109	.069	.021	-.031	-.147	-.273	-.395	-.426	.387
.443	.068	.002	-.051	-.111	-.173	-.309	-.466	-.550	-.660	.443
.498	.009	-.049	-.099	-.155	-.214	-.349	-.478	-.560	-.646	.498
.553	-.045	-.095	-.140	-.187	-.240	-.359	-.406	-.558	-.644	.553
.609	-.037	-.075	-.106	-.139	-.173	-.244	-.332	-.462	-.611	.609
.664	-.024	-.049	-.066	-.074	-.092	-.122	-.198	-.356	-.612	.664
.719	-.001	-.015	-.013	-.013	-.019	-.020	-.076	-.189	-.391	.719
.774	.011	.002	.005	.007	.008	.006	-.019	-.071	-.171	.774
.830	-.007	-.007	-.005	.000	.003	.006	-.003	-.029	-.080	.830
.871	-.012	-.015	-.007	-.004	-.001	.006	.002	-.010	-.041	.871
M = 0.900; q = 358 lb/sq ft										
.166	.029	.012	.002	-.013	-.022	-.051	-.067	-.096	-.105	.166
.277	-.001	-.013	-.019	-.034	-.036	-.047	-.055	-.051	-.036	.277
.367	.122	.091	.068	.044	.025	-.026	-.082	-.122	-.168	.367
.387	.176	.129	.086	.044	.004	-.090	-.184	-.285	-.365	.387
.443	.069	.005	-.055	-.108	-.167	-.271	-.386	-.502	-.542	.443
.498	-.002	-.063	-.120	-.176	-.228	-.347	-.470	-.539	-.582	.498
.553	-.070	-.128	-.198	-.282	-.342	-.446	-.561	-.467	-.648	.553
.609	-.060	-.107	-.151	-.201	-.326	-.481	-.528	-.566	-.626	.609
.664	-.042	-.063	-.080	-.096	-.095	-.215	-.248	-.568	-.648	.664
.719	-.004	-.013	-.012	-.013	-.010	-.001	-.100	-.266	-.452	.719
.774	.008	.002	.005	.005	.012	.021	-.059	-.139	-.267	.774
.830	-.013	-.013	-.004	-.004	.004	.013	-.030	-.072	-.180	.830
.871	-.020	-.014	-.010	-.010	-.003	.010	-.006	-.033	-.120	.871
M = 0.940; q = 368 lb/sq ft										
.166	.029	.009	.000	-.008	-.018	-.051	-.061	-.092	-.098	.166
.277	-.004	-.017	-.024	-.033	-.037	-.049	-.047	-.035	-.015	.277
.367	.131	.102	.079	.060	.043	-.005	-.050	-.078	-.140	.367
.387	.188	.142	.102	.066	.022	-.060	-.147	-.235	-.327	.387
.443	.073	.010	-.045	-.098	-.148	-.248	-.349	-.461	-.569	.443
.498	-.003	-.063	-.117	-.164	-.220	-.331	-.437	-.537	-.512	.498
.553	-.086	-.162	-.236	-.280	-.324	-.426	-.530	-.591	-.632	.553
.609	-.076	-.133	-.222	-.303	-.353	-.470	-.562	-.440	-.672	.609
.664	-.050	-.069	-.081	-.155	-.334	-.494	-.500	-.530	-.680	.664
.719	.002	-.006	-.006	.005	.006	-.062	-.114	-.344	-.399	.719
.774	.011	.007	.009	.017	.028	.022	-.066	-.238	-.282	.774
.830	-.014	-.011	-.008	.000	.013	.026	-.043	-.105	-.221	.830
.871	-.018	-.014	-.009	-.002	.006	.008	-.019	-.053	-.165	.871
M = 0.980; q = 385 lb/sq ft										
.166	.038	.018	.007	-.005	-.016	-.038	-.047	-.066	-.064	.166
.277	.002	-.013	-.021	-.031	-.034	-.037	-.022	-.006	.021	.277
.367	.158	.133	.110	.087	.075	.038	.003	-.023	-.076	.367
.387	.216	.170	.133	.102	.060	-.015	-.094	-.171	-.268	.387
.443	.092	.032	-.015	-.062	-.110	-.202	-.292	-.391	-.503	.443
.498	.017	-.044	-.088	-.145	-.192	-.283	-.379	-.475	-.585	.498
.553	-.104	-.159	-.206	-.248	-.296	-.380	-.474	-.558	-.663	.553
.609	-.102	-.187	-.241	-.286	-.337	-.426	-.520	-.607	-.615	.609
.664	-.150	-.217	-.277	-.323	-.372	-.465	-.567	-.653	-.577	.664
.719	-.043	-.051	-.071	-.103	-.166	-.258	-.304	-.267	-.316	.719
.774	.001	.023	.035	.028	.001	-.085	-.130	-.141	-.243	.774
.830	-.016	.010	.026	.033	.032	.004	-.095	-.165	-.244	.830
.871	-.010	.007	.017	.025	.033	.012	-.090	-.201	-.258	.871

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(b) Station B - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.166	.074	.067	.054	.045	.030	.001	-.023	-.031	-.021	.166
.277	.011	.011	.005	-.005	-.015	-.044	.016	.043	.071	.277
.367	.204	.172	.146	.107	.111	.104	.066	.053	.008	.367
.387	.265	.218	.179	.151	.111	.058	-.026	-.093	-.175	.387
.443	.143	.085	.035	-.007	-.056	-.121	-.213	-.308	-.407	.443
.498	.069	.012	-.041	-.090	-.133	-.209	-.306	-.393	-.487	.498
.553	-.048	-.096	-.146	-.184	-.231	-.300	-.392	-.469	-.565	.553
.609	-.063	-.120	-.181	-.227	-.273	-.352	-.440	-.522	-.625	.609
.664	-.104	-.154	-.217	-.259	-.307	-.393	-.486	-.579	-.684	.664
.719	-.091	-.098	-.123	-.138	-.157	-.217	-.285	-.360	-.291	.719
.774	-.054	-.037	-.028	-.035	-.056	-.110	-.168	-.090	-.089	.774
.830	-.075	-.058	-.050	-.048	-.061	-.084	-.096	-.083	-.181	.830
.871	-.064	-.054	-.060	-.057	-.066	-.102	-.105	-.154	-.194	.871
M = 1.125; q = 421 lb/sq ft										
.166	.047	.037	.033	.019	.002	-.029	-.067	-.119	-.147	.166
.277	.004	.000	-.002	-.017	-.024	-.046	-.081	-.102	-.115	.277
.367	-.035	-.028	-.031	-.022	-.032	-.021	-.009	.031	.116	.367
.387	.220	.178	.150	.110	.074	.009	-.037	-.047	-.051	.387
.443	.141	.091	.057	.012	-.036	-.113	-.183	-.232	-.290	.443
.498	.079	.021	-.019	-.061	-.102	-.181	-.253	-.307	-.359	.498
.553	-.015	-.063	-.100	-.144	-.187	-.255	-.326	-.381	-.437	.553
.609	-.046	-.097	-.135	-.182	-.229	-.296	-.364	-.437	-.491	.609
.664	-.066	-.124	-.164	-.211	-.255	-.336	-.411	-.493	-.534	.664
.719	-.068	-.089	-.106	-.126	-.146	-.168	-.203	-.225	-.146	.719
.774	-.039	-.039	-.041	-.049	-.060	-.080	-.109	.025	.092	.774
.830	-.061	-.056	-.047	-.044	-.048	-.051	-.051	.042	-.030	.830
.871	-.042	-.042	-.037	-.036	-.038	-.066	-.054	.005	-.036	.871
M = 1.200; q = 436 lb/sq ft										
.166	.064	.051	.043	.031	.015	-.016	-.046	-.096	-.133	.166
.277	.020	.014	.001	-.003	-.007	-.026	-.051	-.089	-.121	.277
.367	-.043	-.040	.019	.011	-.049	-.049	-.036	-.055	-.062	.367
.387	.200	.169	.050	.029	.094	.024	-.032	-.088	-.098	.387
.443	.163	.116	.084	.035	-.003	-.086	-.163	-.225	-.281	.443
.498	.101	.059	.026	-.015	-.064	-.144	-.216	-.287	-.353	.498
.553	.023	-.024	-.056	-.099	-.137	-.213	-.284	-.345	-.431	.553
.609	-.018	-.056	-.094	-.128	-.171	-.247	-.323	-.395	-.485	.609
.664	-.057	-.103	-.148	-.188	-.220	-.293	-.370	-.448	-.546	.664
.719	-.071	-.088	-.111	-.122	-.135	-.160	-.185	-.238	-.251	.719
.774	-.042	-.046	-.057	-.063	-.071	-.096	-.147	-.148	-.188	.774
.830	-.062	-.057	-.056	-.050	-.048	-.049	-.055	-.066	-.046	.830
.871	-.037	-.032	-.030	-.028	-.029	-.048	-.049	-.054	-.024	.871

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(c) Station C

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.055										.055
.166	-.007	-.004	-.001	.000	-.006	-.047	-.110	-.191	-.296	.166
.277	-.026	-.020	-.013	-.016	-.025	-.065	-.127	-.206	-.305	.277
.353										.353
.367	.225	.297	.322	.330	.332	.280	.140	-.040	-.179	.367
.692	.032	.031	.036	.042	.047	.043	-.015	-.052	-.226	.692
.719	.009	.004	.009	.013	.017	.022	-.010	-.071	-.248	.719
.774	-.006	-.008	-.004	.001	.000	.007	-.016	-.065	-.192	.774
.830	.008	.003	.004	.012	.018	.025	.014	-.018	-.103	.830
.871	.020	.017	.022	.025	.031	.040	.032	.021	-.028	.871
.954	.036	.033	.036	.039	.041	.035	.036	.046	.030	.954
M = 0.900; q = 358 lb/sq ft										
.055										.055
.166	-.008	-.005	-.002	-.006	-.008	-.045	-.100	-.181	-.274	.166
.277	-.027	-.024	-.020	-.027	-.027	-.058	-.107	-.172	-.258	.277
.353										.353
.367	.261	.305	.333	.334	.342	.318	.223	.078	-.090	.367
.692	.032	.031	.037	.039	.046	.025	-.059	-.153	-.216	.692
.719	.012	.009	.010	.012	.025	.040	-.042	-.171	-.298	.719
.774	-.006	-.006	-.006	-.001	.007	.025	-.063	-.172	-.318	.774
.830	.001	.002	.003	.006	.016	.032	-.018	-.097	-.240	.830
.871	.014	.017	.019	.022	.029	.045	.019	-.022	-.116	.871
.954	.035	.036	.041	.038	.041	.038	.039	.040	.003	.954
M = 0.940; q = 368 lb/sq ft										
.055										.055
.166	-.005	-.005	-.001	-.002	-.004	-.049	-.095	-.168	-.263	.166
.277	-.025	-.026	-.022	-.026	-.030	-.059	-.099	-.151	-.242	.277
.353										.353
.367	.276	.317	.341	.346	.355	.332	.252	.122	-.035	.367
.692	.031	.035	.040	.043	.015	-.079	-.145	-.168	-.201	.692
.719	.022	.015	.016	.030	.032	-.022	-.077	-.196	-.270	.719
.774	.000	-.002	.004	.011	.021	.021	-.078	-.265	-.330	.774
.830	.000	.002	.003	.012	.025	.034	-.045	-.174	-.306	.830
.871	.014	.016	.020	.027	.038	.049	.001	-.066	-.190	.871
.954	.041	.041	.042	.043	.047	.039	.037	.037	-.010	.954
M = 0.980; q = 385 lb/sq ft										
.055										.055
.166	.001	.005	.008	.004	-.001	-.036	-.083	-.146	-.233	.166
.277	-.022	-.021	-.018	-.021	-.030	-.048	-.078	-.126	-.202	.277
.353										.353
.367	.303	.338	.361	.361	.369	.356	.290	.170	.011	.367
.692	-.094	-.095	-.093	-.087	-.109	-.152	-.214	-.207	-.195	.692
.719	-.057	-.045	-.039	-.051	-.101	-.169	-.178	-.088	-.152	.719
.774	-.009	.016	.027	.016	-.011	-.094	-.119	-.153	-.269	.774
.830	.007	.025	.032	.033	.024	-.014	-.138	-.215	-.331	.830
.871	.023	.035	.042	.051	.053	.034	-.071	-.173	-.263	.871
.954	.057	.059	.059	.062	.063	.070	.032	-.063	-.132	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(c) Station C - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.055										.055
.166	.045	.049	.050	.050	.042	.006	-.056	-.109	-.192	.166
.277	-.013	.003	.009	.005	-.006	-.045	-.045	-.077	-.147	.277
.353										.353
.367	.322	.348	.363	.361	.372	.397	.339	.224	.084	.367
.692	-.073	-.063	-.074	-.067	-.072	-.089	-.163	-.182	-.138	.692
.719	-.090	-.072	-.080	-.076	-.084	-.120	-.158	-.158	-.072	.719
.774	-.078	-.049	-.035	-.046	-.066	-.108	-.143	-.073	-.089	.774
.830	-.057	-.047	-.041	-.040	-.058	-.094	-.129	-.147	-.249	.830
.871	-.033	-.029	-.028	-.025	-.038	-.053	-.060	-.138	-.199	.871
.954	-.093	-.085	-.074	-.071	-.084	-.088	-.130	-.209	-.254	.954
M = 1.125; q = 421 lb/sq ft										
.055										.055
.166	.009	.020	.027	.026	.016	-.024	-.094	-.187	-.291	.166
.277	-.013	-.011	.004	-.004	-.015	-.051	-.127	-.226	-.340	.277
.353										.353
.367	.226	.273	.286	.283	.299	.246	.114	-.035	.089	.367
.692	-.052	-.051	-.045	-.047	-.051	-.060	-.109	-.081	.012	.692
.719	-.065	-.063	-.061	-.065	-.067	-.074	-.089	-.074	.030	.719
.774	-.062	-.058	-.050	-.054	-.060	-.075	-.095	.008	.040	.774
.830	-.044	-.042	-.034	-.035	-.044	-.062	-.089	-.053	-.097	.830
.871	-.013	-.017	-.008	-.008	-.006	-.015	-.017	-.004	-.055	.871
.954	-.030	-.038	-.039	-.049	-.051	-.058	.005	-.065	-.109	.954
M = 1.200; q = 436 lb/sq ft										
.055										.055
.166	.027	.037	.038	.037	.028	-.013	-.073	-.172	-.276	.166
.277	.003	.005	.003	.003	-.004	-.033	-.104	-.205	-.347	.277
.353										.353
.367	.222	.277	.027	.039	.316	.246	.132	-.064	-.307	.367
.692	-.047	-.041	-.042	-.039	-.035	-.047	-.085	-.084	-.092	.692
.719	-.065	-.059	-.059	-.056	-.053	-.059	-.063	-.083	-.078	.719
.774	-.062	-.057	-.055	-.053	-.057	-.072	-.092	-.108	-.137	.774
.830	-.045	-.040	-.040	-.042	-.049	-.066	-.092	-.151	-.172	.830
.871	-.004	.001	.002	.006	.003	-.013	-.035	-.068	-.083	.871
.954	-.076	-.071	-.070	-.072	-.073	-.091	-.114	-.125	-.137	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(d) Station D

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.166	.003	.005	.004	.020	.031	.058	.067	.068	.048	.166
.277										.277
.367	-.010	.023	.054	.081	.103	.145	.178	.220	.240	.367
.387	-.108	-.055	-.003	.045	.086	.163	.233	.296	.346	.387
.443	-.214	-.145	-.080	-.026	.032	.136	.243	.335	.417	.443
.498	-.192	-.134	-.077	-.026	.026	.127	.218	.308	.382	.498
.553	-.148	-.093	-.045	.002	.044	.127	.200	.265	.331	.553
.609	-.130	-.099	-.062	-.026	.006	.070	.123	.163	.211	.609
.664	-.069	-.052	-.032	-.009	.010	.046	.064	.074	.074	.664
.719	-.020	-.018	-.011	.004	.013	.026	.014	-.028	-.124	.719
.774	.001	.004	.008	.020	.022	.030	.015	-.031	-.145	.774
.830	.015	.011	.011	.014	.019	.021	.009	-.020	-.113	.830
.871	-.014	-.016	-.019	-.012	-.013	-.018	-.035	-.053	-.131	.871
M = 0.900; q = 358 lb/sq ft										
.166	.002	.004	.002	.014	.026	.063	.074	.079	.071	.166
.277										.277
.367	.025	.049	.074	.093	.113	.158	.206	.249	.285	.367
.387	-.084	-.037	.011	.051	.095	.180	.261	.329	.388	.387
.443	-.226	-.157	-.095	-.039	.024	.145	.254	.359	.455	.443
.498	-.218	-.156	-.101	-.044	.013	.125	.226	.324	.419	.498
.553	-.244	-.155	-.084	-.034	.021	.116	.201	.283	.366	.553
.609	-.195	-.133	-.087	-.050	-.011	.067	.120	.183	.252	.609
.664	-.076	-.061	-.042	-.022	.004	.039	.052	.078	.123	.664
.719	-.021	-.016	-.004	.000	.018	.030	-.024	-.098	-.126	.719
.774	.001	.008	.013	.015	.025	.040	-.034	-.138	-.242	.774
.830	.006	.008	.011	.012	.018	.026	-.024	-.102	-.250	.830
.871	-.018	-.021	-.020	-.022	-.017	-.014	-.055	-.116	-.245	.871
M = 0.940; q = 368 lb/sq ft										
.166	.006	.005	.004	.015	.028	.061	.077	.090	.082	.166
.277										.277
.367	.048	.064	.088	.110	.129	.167	.216	.268	.302	.367
.387	-.055	-.020	.023	.066	.107	.190	.272	.350	.412	.387
.443	-.209	-.158	-.096	-.040	.027	.146	.261	.374	.475	.443
.498	-.217	-.159	-.101	-.046	.011	.125	.229	.338	.441	.498
.553	-.233	-.169	-.131	-.055	.009	.106	.203	.300	.390	.553
.609	-.311	-.241	-.110	-.062	-.021	.048	.123	.201	.279	.609
.664	-.136	-.053	-.040	-.025	-.007	.010	.046	.093	.155	.664
.719	-.002	-.008	-.005	.013	.014	-.017	-.054	-.093	-.095	.719
.774	.011	.011	.014	.026	.037	.030	-.060	-.206	-.245	.774
.830	.005	.007	.010	.016	.024	.024	-.054	-.191	-.308	.830
.871	-.018	-.021	-.023	-.017	-.014	-.020	-.078	-.179	-.427	.871
M = 0.980; q = 389 lb/sq ft										
.166	.013	.011	.010	.016	.033	.072	.088	.106	.110	.166
.277										.277
.367	.083	.099	.113	.137	.147	.194	.246	.299	.343	.367
.387	-.017	.019	.057	.095	.129	.214	.305	.381	.448	.387
.443	-.175	-.127	-.075	-.021	.039	.165	.289	.404	.510	.443
.498	-.195	-.136	-.079	-.031	.019	.140	.260	.367	.475	.498
.553	-.212	-.161	-.113	-.073	-.015	.111	.224	.331	.428	.553
.609	-.300	-.252	-.199	-.144	-.065	.051	.147	.237	.323	.609
.664	-.323	-.262	-.207	-.150	-.070	.001	.070	.139	.201	.664
.719	-.100	-.071	-.041	-.036	-.087	-.118	-.096	-.029	-.029	.719
.774	.009	.031	.038	.024	-.002	-.070	-.076	-.087	-.158	.774
.830	.021	.033	.038	.036	.026	-.009	-.109	-.162	-.271	.830
.871	.001	.004	.004	-.001	-.006	-.031	-.204	-.316	-.453	.871

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.8 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(d) Station D - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.166	.052	.058	.054	.069	.077	.110	.14	.140	.155	.166
.277										.277
.367	.135	.135	.141	.129	.179	.246	.298	.354	.393	.367
.387	.048	.078	.104	.145	.174	.270	.353	.437	.499	.387
.443	-.104	-.068	-.028	.025	.086	.218	.338	.457	.559	.443
.498	-.125	-.080	-.034	.022	.073	.190	.306	.421	.524	.498
.553	-.143	-.105	-.059	-.015	.034	.164	.276	.382	.477	.553
.609	-.228	-.188	-.142	-.092	-.027	.103	.196	.291	.379	.609
.664	-.254	-.203	-.158	-.101	-.041	.061	.126	.198	.267	.664
.719	-.137	-.111	-.100	-.073	-.069	-.062	-.058	-.027	.033	.719
.774	-.062	-.033	-.024	-.032	-.044	-.072	-.070	-.003	.013	.774
.830	-.048	-.035	-.031	-.028	-.041	-.061	-.082	-.062	-.137	.830
.871	-.076	-.077	-.087	-.094	-.114	-.151	-.193	-.271	-.360	.871
M = 1.105; q = 421 lb/sq ft										
.166	.022	.024	.035	.042	.049	.076	.079	.081	.073	.166
.277										.277
.367	.004	-.007	-.007	-.025	-.041	-.066	-.086	.162	.406	.367
.387	.024	.050	.079	.105	.137	.187	.306	.440	.547	.387
.443	-.079	-.047	-.008	.039	.081	.202	.337	.483	.622	.443
.498	-.104	-.059	-.008	.033	.088	.197	.316	.458	.595	.498
.553	-.084	-.054	-.012	.022	.066	.171	.297	.434	.559	.553
.609	-.190	-.151	-.095	-.056	-.011	.121	.239	.364	.470	.609
.664	-.204	-.158	-.109	-.062	-.012	.100	.193	.287	.371	.664
.719	-.123	-.105	-.083	-.065	-.048	-.006	.018	.069	.151	.719
.774	-.052	-.044	-.038	-.035	-.036	-.034	-.034	.076	.128	.774
.830	-.029	-.026	-.017	-.019	-.025	-.029	-.038	.019	-.005	.830
.871	-.056	-.066	-.070	-.085	-.097	-.127	-.150	-.158	-.206	.871
M = 1.200; q = 436 lb/sq ft										
.166	.021	.035	.044	.057	.065	.087	.094	.091	.100	.166
.277										.277
.367	-.025	-.008	.008	.009	-.035	-.054	-.065	-.086	-.098	.367
.387	.037	.046	.012	.030	.140	.162	.208	.314	.467	.387
.443	-.059	-.019	.019	.058	.107	.209	.333	.455	.592	.443
.498	-.074	-.024	.019	.063	.111	.214	.326	.429	.557	.498
.553	-.053	-.013	.017	.055	.097	.195	.291	.410	.531	.553
.609	-.140	-.112	-.072	-.032	.016	.108	.226	.347	.458	.609
.664	-.179	-.138	-.100	-.053	-.001	.104	.216	.300	.377	.664
.719	-.119	-.098	-.078	-.056	-.033	.015	.061	.095	.138	.719
.774	-.062	-.050	-.045	-.039	-.028	-.016	-.012	-.008	.002	.774
.830	-.019	-.013	-.016	-.017	-.015	-.022	-.042	-.054	-.077	.830
.871	-.056	-.058	-.068	-.077	-.094	-.127	-.155	-.207	-.222	.871

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.8 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(e) Station E

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.055	.028	.043	.063	.092	.118	.178	.245	.330	.406	.055
.166										.166
.277	-.029	-.032	-.020	-.000	.025	.070	.131	.209	.285	.277
.367	.003	.025	.020	.055	.120	.193	.287	.379	.467	.367
.387	-.084	-.053	-.011	.023	.054	.138	.229	.324	.409	.387
.443	-.185	-.129	-.081	-.032	.021	.118	.217	.306	.394	.443
.498	-.193	-.137	-.087	-.040	.011	.106	.194	.275	.357	.498
.553	-.169	-.122	-.074	-.033	.011	.094	.168	.238	.299	.553
.609	-.127	-.094	-.059	-.024	.008	.071	.128	.175	.217	.609
.664	-.058	-.038	-.016	.007	.031	.070	.094	.111	.127	.664
.719	-.036	-.028	-.020	-.007	.006	.029	.026	-.002	-.044	.719
.774	-.001	-.003	.005	.010	.017	.029	.018	-.020	-.086	.774
.830	.003	-.004	-.001	.004	.006	.013	.010	-.052	-.130	.830
.871	-.020	-.025	-.027	-.027	-.031	-.036	-.049	-.062	-.104	.871
.954	.060	.055	.052	.052	.048	.038	.030	.030	.012	.954
M = 0.900; q = 358 lb/sq ft										
.055	.032	.047	.072	.089	.121	.185	.259	.344	.423	.055
.166										.166
.277	-.028	-.030	-.024	-.014	.020	.069	.134	.215	.303	.277
.367	.035	.048	.030	.090	.127	.208	.303	.403	.501	.367
.387	-.062	-.036	.001	.029	.065	.148	.247	.345	.444	.387
.443	-.196	-.140	-.093	-.044	.015	.122	.228	.328	.434	.443
.498	-.230	-.172	-.111	-.060	-.001	.103	.203	.297	.393	.498
.553	-.270	-.181	-.111	-.061	-.011	.086	.171	.257	.341	.553
.609	-.192	-.131	-.085	-.048	-.009	.066	.123	.186	.261	.609
.664	-.064	-.048	-.023	-.004	.023	.068	.087	.115	.167	.664
.719	-.030	-.025	-.015	-.008	.004	.028	-.014	-.052	-.043	.719
.774	-.001	-.002	.003	.005	.020	.035	-.029	-.114	-.172	.774
.830	-.004	-.002	-.002	-.001	.007	.017	-.027	-.130	-.200	.830
.871	-.023	-.027	-.030	-.037	-.035	-.034	-.074	-.122	-.237	.871
.954	.059	.053	.055	.049	.051	.045	.030	.013	-.033	.954
M = 0.940; q = 368 lb/sq ft										
.055	.038	.053	.075	.097	.126	.185	.259	.354	.436	.055
.166										.166
.277	-.026	-.031	-.025	-.011	.018	.063	.134	.223	.316	.277
.367	.057	.065	.037	.101	.143	.213	.308	.414	.518	.367
.387	-.040	-.018	.009	.042	.076	.158	.256	.356	.462	.387
.443	-.186	-.139	-.091	-.040	.015	.126	.237	.344	.451	.443
.498	-.227	-.170	-.121	-.066	-.008	.100	.207	.313	.411	.498
.553	-.257	-.215	-.155	-.084	-.023	.078	.174	.272	.361	.553
.609	-.305	-.247	-.113	-.062	-.020	.052	.126	.207	.285	.609
.664	-.132	-.042	-.023	-.005	.014	.039	.082	.135	.195	.664
.719	-.014	-.018	-.013	.000	.005	-.010	-.032	-.049	-.007	.719
.774	.005	.002	.005	.016	.028	.018	-.057	-.165	-.162	.774
.830	-.003	-.004	-.002	.004	.014	.010	-.058	-.169	-.195	.830
.871	-.026	-.031	-.030	-.032	-.028	-.042	-.098	-.221	-.367	.871
.954	.059	.058	.056	.057	.052	.044	.032	.006	-.040	.954
M = 0.980; q = 385 lb/sq ft										
.055	.050	.064	.087	.114	.138	.205	.277	.363	.463	.055
.166										.166
.277	-.017	-.025	-.020	-.007	.019	.073	.152	.242	.346	.277
.367	.097	.100	.067	.131	.161	.237	.339	.445	.554	.367
.387	-.003	.024	.048	.068	.097	.181	.284	.387	.494	.387
.443	-.157	-.108	-.073	-.025	.030	.144	.267	.370	.485	.443
.498	-.208	-.150	-.099	-.055	-.001	.117	.236	.340	.448	.498
.553	-.238	-.190	-.139	-.102	-.046	.086	.200	.301	.400	.553
.609	-.293	-.248	-.196	-.148	-.066	.053	.150	.240	.328	.609
.664	-.298	-.240	-.187	-.128	-.043	.027	.104	.176	.244	.664
.719	-.119	-.081	-.044	-.036	-.082	-.095	-.055	.011	.047	.719
.774	.012	.030	.032	.016	-.009	-.077	-.069	-.071	-.082	.774
.830	.015	.025	.024	.020	.014	-.017	-.102	-.115	-.132	.830
.871	-.004	-.005	-.010	-.018	-.023	-.055	-.233	-.290	-.333	.871
.954	.081	.074	.074	.066	.066	.058	.007	-.209	-.311	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Concluded

(e) Station E - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.055	.096	.113	.131	.151	.176	.243	.318	.406	.498	.055
.166										.166
.277	-.009	-.002	.005	.016	.036	.072	.180	.284	.385	.277
.367	.156	.108	.044	.130	.191	.289	.388	.499	.600	.367
.387	.066	.088	.116	.119	.140	.232	.331	.440	.543	.387
.443	-.083	-.052	-.021	.022	.074	.197	.315	.429	.535	.443
.498	-.138	-.096	-.055	-.001	.049	.169	.283	.396	.501	.498
.553	-.163	-.127	-.085	-.043	.009	.133	.248	.354	.452	.553
.609	-.216	-.182	-.138	-.093	-.028	.103	.201	.299	.385	.609
.664	-.225	-.177	-.134	-.079	-.018	.084	.159	.238	.305	.664
.719	-.173	-.129	-.118	-.079	-.068	-.041	-.009	.049	.115	.719
.774	-.062	-.039	-.026	-.040	-.049	-.073	-.056	.021	.057	.774
.830	-.059	-.047	-.040	-.037	-.049	-.062	-.068	-.023	-.040	.830
.871	-.088	-.090	-.101	-.114	-.139	-.170	-.191	-.217	-.248	.871
.954	-.042	-.049	-.053	-.073	-.091	-.107	-.141	-.183	-.213	.954
M = 1.125; q = 421 lb/sq ft										
.055	.060	.077	.103	.123	.148	.209	.279	.364	.442	.055
.166										.166
.277	-.012	-.014	-.002	.006	.021	.062	.114	.169	.239	.277
.367	-.015	-.014	-.019	-.013	-.005	.045	.088	.219	.596	.367
.387	.021	.014	.025	.021	.020	.050	.215	.409	.579	.387
.443	-.061	-.030	.002	.038	.070	.179	.310	.451	.597	.443
.498	-.099	-.068	-.024	.015	.060	.176	.294	.434	.570	.498
.553	-.131	-.091	-.043	-.008	.036	.147	.273	.409	.535	.553
.609	-.173	-.136	-.091	-.049	-.006	.120	.240	.364	.472	.609
.664	-.182	-.138	-.083	-.042	.006	.122	.229	.321	.405	.664
.719	-.173	-.141	-.104	-.072	-.043	.026	.070	.143	.228	.719
.774	-.054	-.056	-.049	-.048	-.044	-.030	-.014	.081	.169	.774
.830	-.033	-.034	-.027	-.029	-.032	-.021	-.01	.073	.068	.830
.871	-.064	-.081	-.092	-.107	-.122	-.131	-.13	-.105	-.107	.871
.954	.030	.002	-.019	-.043	-.056	-.071	-.03	-.058	-.071	.954
M = 1.200; q = 436 lb/sq ft										
.055	.077	.098	.116	.139	.170	.231	.293	.373	.464	.055
.166										.166
.277	-.005	.005	.013	.023	.040	.078	.134	.183	.257	.277
.367	-.019	-.005	.001	.001	.012	.041	.109	.158	.235	.367
.387	.010	.012	.019	.021	.018	.046	.078	.124	.378	.387
.443	-.029	.001	.031	.066	.104	.209	.305	.417	.552	.443
.498	-.082	-.034	.003	.036	.084	.185	.303	.412	.537	.498
.553	-.102	-.056	-.018	.017	.067	.165	.271	.392	.511	.553
.609	-.139	-.101	-.071	-.028	.020	.122	.224	.359	.464	.609
.664	-.153	-.111	-.061	-.014	.033	.128	.241	.333	.414	.664
.719	-.172	-.137	-.101	-.065	-.025	.049	.122	.169	.229	.719
.774	-.060	-.059	-.056	-.047	-.039	-.011	.018	.034	.082	.774
.830	-.017	-.016	-.015	-.011	-.004	.006	.008	.056	.076	.830
.871	-.067	-.078	-.090	-.095	-.105	-.115	-.110	-.111	-.097	.871
.954	-.031	-.040	-.053	-.058	-.065	-.082	-.087	-.099	-.097	.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE

OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING

(a) Station A

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 627 lb/sq ft										
.055										.055
.166	.046	.026	.010	-.003	-.017	-.029	-.031	-.030	-.044	.166
.277	.029	.009	-.005	-.018	-.031	-.036	-.023	-.010	-.013	.277
.367	.091	.062	.036	.010	-.015	-.060	-.102	-.137	-.160	.367
.387	.103	.064	.027	-.006	-.045	-.107	-.170	-.255	-.281	.387
.415	.075	.028	-.018	-.062	-.109	-.202	-.304	-.394	-.439	.415
.443	.049	-.002	-.052	-.100	-.154	-.262	-.386	-.437	-.472	.443
.498	.038	-.015	-.068	-.120	-.174	-.266	-.346	-.376	-.378	.498
.553	.012	-.034	-.081	-.128	-.176	-.244	-.269	-.331	-.304	.553
.581										.581
.609	-.009	-.034	-.059	-.110	-.144	-.189	-.221	-.300	-.305	.609
.636	-.013	-.042	-.068	-.092	-.117	-.139	-.192	-.285	-.328	.636
.664	-.018	-.038	-.057	-.073	-.090	-.103	-.154	-.262	-.361	.664
.692	-.003	-.018	-.030	-.040	-.052	-.050	-.094	-.191	-.327	.692
.719	.019	.008	-.002	-.010	-.016	-.002	-.039	-.120	-.268	.719
.774	.019	.014	.011	.008	.005	.017	-.006	-.062	-.187	.774
.830	.001	-.000	.002	.003	.002	.009	.000	-.034	-.106	.830
.871	.011	.011	.014	.015	.015	.023	.017	-.004	-.037	.871
.954	.020	.026	.032	.038	.039	.038	.036	.025	.043	.954
M = 0.900; q = 713 lb/sq ft										
.055										.055
.166	.043	.022	.006	-.008	-.017	-.025	-.027	-.019	-.025	.166
.277	.021	.001	-.014	-.024	-.030	-.025	-.004	.025	.031	.277
.367	.104	.074	.051	.030	.015	-.011	-.034	-.045	-.082	.367
.387	.117	.077	.043	.012	-.014	-.057	-.105	-.174	-.225	.387
.415	.085	.035	-.009	-.053	-.091	-.161	-.234	-.337	-.417	.415
.443	.050	-.004	-.054	-.104	-.150	-.240	-.326	-.419	-.411	.443
.498	.026	-.032	-.088	-.145	-.195	-.280	-.391	-.472	-.417	.498
.553	-.011	-.069	-.131	-.207	-.265	-.347	-.454	-.314	-.424	.553
.581										.581
.609	-.027	-.059	-.093	-.163	-.214	-.429	-.468	-.400	-.360	.609
.636	-.034	-.067	-.099	-.132	-.151	-.444	-.432	-.454	-.385	.636
.664	-.032	-.055	-.074	-.091	-.094	-.142	-.121	-.486	-.393	.664
.692	-.010	-.024	-.035	-.044	-.044	-.019	-.087	-.431	-.418	.692
.719	.014	.004	-.004	-.010	-.006	.029	-.055	-.128	-.279	.719
.774	.015	.012	.009	.008	.013	.041	-.030	-.114	-.272	.774
.830	-.006	-.004	-.002	-.001	.006	.021	-.017	-.069	-.183	.830
.871	.006	.007	.009	.011	.017	.030	.008	-.023	-.103	.871
.954	.018	.025	.032	.038	.042	.041	.033	.023	-.000	.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Continued

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.940; q = 743 lb/sq ft										
.055										.055
.166	.044	.020	.003	-.010	-.017	-.027	-.022	-.009	-.013	.166
.277	.021	-.003	-.018	-.029	-.030	-.024	.007	.041	.054	.277
.367	.117	.085	.063	.045	.036	.014	.003	-.004	-.035	.367
.387	.131	.089	.056	.028	.008	-.029	-.068	-.128	-.177	.387
.415	.096	.043	-.000	-.039	-.070	-.131	-.194	-.292	-.372	.415
.443	.058	-.001	-.050	-.096	-.133	-.216	-.284	-.379	-.461	.443
.498	.025	-.037	-.090	-.141	-.189	-.267	-.364	-.437	-.342	.498
.553	-.028	-.105	-.166	-.215	-.254	-.333	-.419	-.456	-.414	.553
.581										.581
.609	-.041	-.087	-.154	-.269	-.315	-.411	-.499	-.276	-.489	.609
.636	-.051	-.094	-.132	-.287	-.343	-.442	-.511	-.465	-.512	.636
.664	-.042	-.065	-.078	-.167	-.341	-.468	-.509	-.438	-.479	.664
.692	-.010	-.023	-.030	-.029	-.087	-.426	-.467	-.443	-.455	.692
.719	.019	.008	.001	.006	.010	-.013	-.039	-.319	-.365	.719
.774	.019	.013	.011	.016	.032	.055	-.008	-.214	-.273	.774
.830	-.005	-.006	-.004	.002	.018	.041	-.017	-.096	-.254	.830
.871	.006	.006	.007	.011	.025	.044	.005	-.049	-.150	.871
.954	.023	.027	.032	.038	.048	.054	.039	.017	-.033	.954
M = 0.980; q = 771 lb/sq ft										
.055										.055
.166	.048	.028	.014	-.001	-.011	-.018	-.010			.166
.277	.023	.001	-.012	-.023	-.026	-.012	.027			.277
.367	.139	.112	.094	.078	.067	.052	.046			.367
.387	.154	.118	.090	.063	.043	.012	-.023			.387
.415	.117	.071	.032	-.004	-.036	-.084	-.151			.415
.443	.076	.024	-.021	-.062	-.102	-.168	-.233			.443
.498	.037	-.014	-.063	-.119	-.163	-.233	-.320			.498
.553	-.044	-.095	-.138	-.188	-.230	-.297	-.373			.553
.581										.581
.609	-.069	-.125	-.211	-.258	-.299	-.385	-.461			.609
.636	-.112	-.177	-.235	-.286	-.331	-.409	-.491			.636
.664	-.145	-.203	-.253	-.308	-.356	-.436	-.519			.664
.692	-.138	-.201	-.254	-.301	-.345	-.444	-.541			.692
.719	-.012	-.017	-.062	-.108	-.181	-.336	-.500			.719
.774	.013	.037	.047	.041	.025	-.054	-.091			.774
.830	-.003	.017	.033	.040	.043	.027	-.012			.830
.871	.012	.026	.036	.043	.051	.061	.020			.871
.954	.032	.043	.057	.065	.075	.091	.075			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Continued

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 803 lb/sq ft										
.055										.055
.166	.089	.068	.056	.041	.032	.014	.016			.166
.277	.033	.013	.007	-.008	-.014	-.026	.067			.277
.367	.189	.152	.128	.113	.107	.115	.114			.367
.387	.208	.166	.136	.112	.095	.078	.047			.387
.415	.174	.122	.081	.048	.022	-.012	-.079			.415
.443	.134	.074	.030	-.008	-.042	-.094	-.157			.443
.498	.098	.038	-.019	-.068	-.105	-.162	-.250			.498
.553	.019	-.035	-.086	-.131	-.167	-.226	-.298			.553
.581										.581
.609	-.033	-.077	-.153	-.201	-.232	-.309	-.392			.609
.636	-.061	-.121	-.184	-.227	-.264	-.338	-.414			.636
.664	-.091	-.145	-.204	-.249	-.287	-.365	-.440			.664
.692	-.106	-.153	-.204	-.244	-.281	-.377	-.462			.692
.719	-.078	-.106	-.138	-.160	-.190	-.305	-.444			.719
.774	-.043	-.038	-.025	-.039	-.053	-.087	-.116			.774
.830	-.069	-.063	-.049	-.053	-.058	-.070	-.072			.830
.871	-.054	-.049	-.041	-.044	-.046	-.046	-.030			.871
.954	-.141	-.124	-.089	-.066	-.050	-.038	-.049			.954
M = 1.125; q = 856 lb/sq ft										
.055										.055
.166	.058	.044	.034	.018	.002	-.009	-.015			.166
.277	.028	.017	.006	-.007	-.015	-.025	-.023			.277
.367	-.013	-.019	-.024	-.026	-.026	-.016	.007			.367
.387	.113	.086	.070	.056	.047	.032	.045			.387
.415	.164	.124	.096	.063	.032	-.021	-.052			.415
.443	.132	.087	.053	.017	-.017	-.076	-.111			.443
.498	.099	.047	.007	-.038	-.078	-.130	-.189			.498
.553	.039	-.013	-.052	-.093	-.127	-.184	-.231			.553
.581										.581
.609	-.015	-.052	-.120	-.166	-.191	-.247	-.297			.609
.636	-.047	-.099	-.139	-.181	-.214	-.273	-.322			.636
.664	-.061	-.117	-.156	-.196	-.232	-.297	-.351			.664
.692	-.066	-.112	-.153	-.199	-.238	-.307	-.370			.692
.719	-.063	-.103	-.134	-.168	-.198	-.277	-.360			.719
.774	-.027	-.033	-.038	-.040	-.046	-.054	-.077			.774
.830	-.051	-.045	-.043	-.038	-.037	-.034	-.030			.830
.871	-.036	-.031	-.027	-.021	-.018	-.009	.000			.871
.954	-.080	-.076	-.059	-.037	-.019	-.000	.019			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Concluded

x/l	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/l
M = 1.200; q = 888 lb/sq ft										
.055										.055
.166	.075	.057	.044	.032	.016	-.001	-.004			.166
.277	.040	.023	.011	.007	.005	-.012	-.008			.277
.367	.004	-.006	-.016	-.020	-.016	-.017	-.022			.367
.387	.038	.029	.028	.034	.032	.029	.043			.387
.415	.169	.131	.101	.069	.039	-.004	-.038			.415
.443	.154	.111	.070	.039	.008	-.053	-.098			.443
.498	.126	.076	.031	-.013	-.054	-.108	-.163			.498
.553	.070	.023	-.022	-.059	-.097	-.157	-.206			.553
.581										.581
.609	.006	-.028	-.062	-.092	-.159	-.219	-.278			.609
.636	-.024	-.068	-.107	-.139	-.179	-.240	-.294			.636
.664	-.037	-.083	-.130	-.171	-.206	-.260	-.311			.664
.692	-.054	-.096	-.134	-.172	-.207	-.270	-.335			.692
.719	-.054	-.097	-.134	-.168	-.205	-.269	-.340			.719
.774	-.048	-.057	-.059	-.064	-.069	-.069	-.083			.774
.830	-.046	-.050	-.050	-.044	-.040	-.028	-.017			.830
.871	-.043	-.037	-.028	-.018	-.013	.000	.004			.871
.954	-.098	-.086	-.072	-.056	-.042	-.035	-.042			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(b) Station B

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 627 lb/sq ft										
.166	.030	.022	.012	-.001	-.016	-.045	-.077	-.101	-.122	.166
.277	.005	-.002	-.009	-.020	-.035	-.059	-.076	-.086	-.100	.277
.367	.108	.084	.057	.029	-.008	-.083	-.163	-.235	-.265	.367
.387	.158	.114	.068	.020	-.035	-.147	-.270	-.405	-.444	.387
.443	.070	.011	-.046	-.105	-.169	-.306	-.466	-.547	-.629	.443
.498	.012	-.041	-.093	-.146	-.206	-.341	-.465	-.560	-.587	.498
.553	-.040	-.086	-.131	-.177	-.230	-.351	-.397	-.514	-.614	.553
.609	-.029	-.062	-.094	-.127	-.163	-.235	-.314	-.433	-.607	.609
.664	-.021	-.039	-.055	-.071	-.088	-.113	-.191	-.349	-.649	.664
.719	.005	-.001	-.005	-.010	-.016	-.015	-.070	-.180	-.378	.719
.774	.016	.013	.013	.011	.008	.010	-.007	-.051	-.150	.774
.830	-.001	-.000	.002	.003	.003	.007	-.000	-.021	-.073	.830
.871	-.010	-.009	-.005	-.004	-.003	.007	-.001	-.007	-.040	.871
M = 0.900; q = 713 lb/sq ft										
.166	.027	.018	.007	-.005	-.016	-.041	-.072	-.087	-.103	.166
.277	-.000	-.010	-.018	-.027	-.034	-.048	-.057	-.053	-.059	.277
.367	.120	.095	.071	.048	.024	-.029	-.085	-.125	-.173	.367
.387	.177	.130	.086	.041	-.001	-.089	-.191	-.291	-.375	.387
.443	.071	.010	-.048	-.107	-.161	-.270	-.394	-.508	-.510	.443
.498	.002	-.057	-.113	-.169	-.220	-.341	-.473	-.557	-.580	.498
.553	-.065	-.124	-.186	-.269	-.331	-.445	-.563	-.470	-.628	.553
.609	-.053	-.094	-.137	-.185	-.292	-.477	-.520	-.501	-.605	.609
.664	-.036	-.056	-.073	-.089	-.092	-.170	-.207	-.560	-.640	.664
.719	.003	-.003	-.006	-.008	-.005	.010	-.082	-.232	-.421	.719
.774	.012	.011	.011	.011	.014	.027	-.038	-.124	-.222	.774
.830	-.008	-.006	-.003	.000	.005	.016	-.017	-.059	-.155	.830
.871	-.016	-.012	-.010	-.007	-.001	.013	-.006	-.024	-.105	.871
M = 0.940; q = 743 lb/sq ft										
.166	.029	.016	.005	-.006	-.015	-.042	-.066	-.080	-.091	.166
.277	-.001	-.014	-.023	-.031	-.035	-.047	-.045	-.036	-.037	.277
.367	.135	.106	.083	.061	.043	-.003	-.045	-.078	-.121	.367
.387	.191	.143	.101	.060	.023	-.059	-.148	-.238	-.324	.387
.443	.078	.013	-.043	-.097	-.140	-.243	-.345	-.459	-.568	.443
.498	.000	-.061	-.112	-.163	-.213	-.322	-.431	-.533	-.516	.498
.553	-.085	-.165	-.228	-.275	-.316	-.424	-.520	-.563	-.591	.553
.609	-.072	-.129	-.213	-.298	-.346	-.464	-.563	-.466	-.674	.609
.664	-.046	-.065	-.076	-.153	-.343	-.494	-.519	-.534	-.669	.664
.719	.009	.003	-.001	.008	.009	-.059	-.116	-.302	-.369	.719
.774	.018	.013	.013	.019	.032	.033	-.032	-.211	-.236	.774
.830	-.008	-.007	-.004	.001	.016	.034	-.024	-.101	-.207	.830
.871	-.015	-.013	-.011	-.006	.008	.018	-.007	-.049	-.154	.871
M = 0.980; q = 771 lb/sq ft										
.166	.035	.025	.016	.003	-.008	-.033	-.054			.166
.277	.002	-.009	-.016	-.024	-.031	-.037	-.026			.277
.367	.158	.134	.115	.094	.075	.036	.000			.367
.387	.216	.172	.135	.095	.059	-.015	-.098			.387
.443	.094	.036	-.013	-.061	-.108	-.197	-.294			.443
.498	.012	-.038	-.083	-.138	-.186	-.282	-.378			.498
.553	-.107	-.157	-.195	-.244	-.290	-.381	-.468			.553
.609	-.107	-.181	-.231	-.281	-.330	-.427	-.517			.609
.664	-.157	-.216	-.272	-.319	-.363	-.466	-.574			.664
.719	-.036	-.028	-.052	-.084	-.148	-.262	-.331			.719
.774	.012	.036	.045	.037	.018	-.076	-.141			.774
.830	-.006	.016	.031	.036	.039	.017	-.043			.830
.871	-.007	.008	.020	.027	.036	.021	-.034			.871

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(b) Station B - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 803 lb/sq ft										
.166	.074	.065	.059	.045	.035	.001	-.027			.166
.277	.014	.004	.003	-.009	-.018	-.048	.015			.277
.367	.207	.171	.148	.126	.115	.099	.069			.367
.387	.268	.218	.177	.142	.111	.052	-.025			.387
.443	.151	.085	.035	-.010	-.047	-.121	-.217			.443
.498	.073	.016	-.039	-.087	-.125	-.207	-.297			.498
.553	-.043	-.094	-.142	-.184	-.223	-.303	-.387			.553
.609	-.061	-.121	-.178	-.222	-.262	-.351	-.437			.609
.664	-.106	-.160	-.217	-.257	-.297	-.395	-.493			.664
.719	-.088	-.099	-.114	-.127	-.149	-.221	-.291			.719
.774	-.050	-.041	-.026	-.041	-.057	-.109	-.159			.774
.830	-.074	-.067	-.051	-.056	-.062	-.083	-.089			.830
.871	-.066	-.064	-.057	-.061	-.065	-.096	-.060			.871
M = 1.125; q = 856 lb/sq ft										
.166	.047	.041	.036	.024	.006	-.021	-.058			.166
.277	.012	.010	.006	-.004	-.017	-.043	-.074			.277
.367	-.032	-.019	-.012	-.014	-.020	-.016	-.003			.367
.387	.218	.174	.142	.106	.076	.009	-.033			.387
.443	.143	.096	.060	.015	-.027	-.106	-.180			.443
.498	.075	.019	-.020	-.064	-.102	-.178	-.242			.498
.553	-.017	-.065	-.102	-.140	-.176	-.252	-.323			.553
.609	-.043	-.093	-.132	-.175	-.217	-.297	-.360			.609
.664	-.079	-.135	-.173	-.214	-.251	-.330	-.420			.664
.719	-.067	-.087	-.106	-.124	-.142	-.167	-.191			.719
.774	-.029	-.029	-.034	-.037	-.047	-.075	-.095			.774
.830	-.058	-.048	-.044	-.039	-.042	-.050	-.043			.830
.871	-.044	-.038	-.036	-.032	-.036	-.062	-.053			.871
M = 1.200; q = 888 lb/sq ft										
.166	.062	.054	.045	.036	.019	-.014	-.045			.166
.277	.020	.013	.009	.005	-.002	-.030	-.053			.277
.367	-.042	-.036	-.036	-.031	-.033	-.037	-.033			.367
.387	.202	.167	.134	.104	.078	.017	-.039			.387
.443	.158	.112	.067	.027	-.012	-.088	-.168			.443
.498	.099	.055	.012	-.026	-.070	-.146	-.216			.498
.553	.023	-.024	-.065	-.102	-.142	-.216	-.282			.553
.609	-.018	-.055	-.092	-.128	-.169	-.244	-.324			.609
.664	-.066	-.112	-.149	-.189	-.227	-.296	-.378			.664
.719	-.068	-.088	-.107	-.121	-.139	-.158	-.181			.719
.774	-.043	-.048	-.052	-.057	-.070	-.096	-.125			.774
.830	-.064	-.059	-.057	-.053	-.052	-.050	-.053			.830
.871	-.044	-.041	-.036	-.030	-.035	-.051	-.042			.871

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(c) Station C

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 627 lb/sq ft										
.055										.055
.166	-.004	.005	.009	.004	-.008	-.046	-.109	-.193	-.304	.166
.277	-.023	-.013	-.009	-.013	-.027	-.063	-.122	-.199	-.289	.277
.353										.353
.367	.223	.278	.313	.330	.335	.287	.143	-.068	-.300	.367
.692	.039	.040	.044	.048	.051	.050	.009	-.027	-.168	.692
.719	.014	.014	.015	.018	.020	.026	.007	-.064	-.230	.719
.774	-.001	.000	.002	.002	.005	.010	-.008	-.064	-.178	.774
.830	.019	.021	.018	.022	.028	.038	.033	.008	-.073	.830
.871	.025	.026	.028	.031	.035	.045	.040	.034	-.004	.871
.954	.039	.040	.042	.041	.039	.037	.041	.049	.041	.954
M = 0.900; q = 713 lb/sq ft										
.055										.055
.166	-.007	.001	.004	.000	-.008	-.041	-.104	-.181	-.281	.166
.277	-.026	-.019	-.017	-.020	-.028	-.056	-.107	-.169	-.246	.277
.353										.353
.367	.257	.298	.320	.337	.349	.326	.227	.057	-.137	.367
.692	.039	.039	.044	.047	.055	.043	-.048	-.113	-.159	.692
.719	.014	.014	.015	.020	.030	.048	-.018	-.150	-.265	.719
.774	-.004	-.001	.001	.004	.011	.030	-.047	-.159	-.310	.774
.830	.015	.017	.014	.019	.029	.049	.008	-.066	-.201	.830
.871	.021	.023	.025	.028	.035	.051	.027	-.007	-.080	.871
.954	.039	.041	.042	.043	.043	.043	.044	.047	.021	.954
M = 0.940; q = 743 lb/sq ft										
.055										.055
.166	-.007	.000	.001	-.003	-.008	-.043	-.096	-.173	-.276	.166
.277	-.025	-.023	-.022	-.025	-.029	-.056	-.096	-.153	-.226	.277
.353										.353
.367	.276	.310	.327	.344	.358	.341	.260	.101	-.079	.367
.692	.041	.045	.047	.048	.019	-.073	-.159	-.154	-.152	.692
.719	.028	.021	.021	.034	.036	-.016	-.062	-.182	-.238	.719
.774	.004	.000	.003	.011	.026	.031	-.060	-.260	-.323	.774
.830	.016	.015	.013	.020	.037	.056	-.016	-.155	-.282	.830
.871	.022	.023	.023	.028	.042	.058	.014	-.049	-.167	.871
.954	.042	.043	.043	.043	.048	.048	.046	.038	-.001	.954
M = 0.980; q = 771 lb/sq ft										
.055										.055
.166	-.001	.008	.011	.007	-.001	-.033	-.084			.166
.277	-.021	-.017	-.015	-.019	-.025	-.045	-.079			.277
.353										.353
.367	.301	.332	.350	.363	.375	.365	.294			.367
.692	-.090	-.084	-.080	-.078	-.098	-.147	-.222			.692
.719	-.048	-.023	-.024	-.038	-.084	-.170	-.190			.719
.774	-.001	.026	.035	.024	.004	-.087	-.137			.774
.830	.023	.041	.043	.043	.043	.011	-.103			.830
.871	.032	.042	.050	.053	.059	.047	-.024			.871
.954	.058	.061	.063	.064	.067	.072	.053			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(c) Station C - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 803 lb/sq ft										
.055										.055
.166	.042	.049	.055	.050	.042	.002	-.016			.166
.277	-.009	-.004	.005	-.000	-.008	-.050	-.042			.277
.353										.353
.367	.323	.333	.350	.361	.377	.403	.347			.367
.692	-.069	-.062	-.067	-.062	-.061	-.091	-.160			.692
.719	-.085	-.073	-.073	-.073	-.078	-.122	-.154			.719
.774	-.078	-.060	-.039	-.051	-.064	-.111	-.150			.774
.830	-.047	-.043	-.034	-.040	-.046	-.084	-.125			.830
.871	-.031	-.032	-.025	-.029	-.032	-.052	-.056			.871
.954	-.092	-.090	-.077	-.076	-.078	-.089	-.113			.954
M = 1.125; q = 856 lb/sq ft										
.055										.055
.166	.016	.024	.030	.028	.016	-.022	-.086			.166
.277	-.004	.006	.010	.007	-.006	-.045	-.119			.277
.353										.353
.367	.207	.253	.271	.281	.281	.242	.102			.367
.692	-.045	-.043	-.040	-.039	-.040	-.050	-.091			.692
.719	-.061	-.057	-.059	-.059	-.061	-.070	-.080			.719
.774	-.056	-.046	-.044	-.046	-.051	-.072	-.095			.774
.830	-.030	-.022	-.025	-.025	-.027	-.049	-.082			.830
.871	-.008	-.006	-.005	-.002	-.003	-.015	-.022			.871
.954	-.040	-.050	-.052	-.056	-.058	-.068	-.052			.954
M = 1.200; q = 888 lb/sq ft										
.055										.055
.166	.032	.038	.041	.041	.026	-.014	-.077			.166
.277	.006	.011	.010	.011	.001	-.033	-.107			.277
.353										.353
.367	.207	.236	.257	.277	.275	.238	.116			.367
.692	-.046	-.045	-.040	-.034	-.033	-.034	-.069			.692
.719	-.062	-.061	-.061	-.058	-.058	-.059	-.062			.719
.774	-.064	-.060	-.055	-.054	-.059	-.072	-.092			.774
.830	-.037	-.033	-.036	-.037	-.043	-.063	-.097			.830
.871	-.002	-.004	-.003	.000	-.002	-.016	-.036			.871
.954	-.078	-.073	-.072	-.072	-.078	-.096	-.121			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(d) Station D

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 627 lb/sq ft										
.166	.012	.026	.037	.046	.053	.067	.078	.078	.075	.166
.277										.277
.367	-.004	.029	.061	.085	.106	.148	.189	.221	.250	.367
.387	-.104	-.049	.003	.046	.084	.161	.235	.292	.347	.387
.443	-.204	-.135	-.073	-.020	.031	.141	.244	.332	.422	.443
.498	-.186	-.123	-.069	-.020	.026	.128	.222	.302	.384	.498
.553	-.164	-.117	-.070	-.032	.005	.089	.168	.234	.301	.553
.609	-.127	-.091	-.056	-.024	.005	.072	.124	.166	.213	.609
.664	-.065	-.046	-.026	-.010	.007	.047	.067	.074	.081	.664
.719	-.015	-.007	.001	.008	.014	.033	.024	-.018	-.095	.719
.774	.009	.013	.019	.022	.025	.036	.024	-.021	-.123	.774
.830	.018	.021	.023	.026	.025	.031	.024	.000	-.083	.830
.871	-.007	-.009	-.010	-.012	-.013	-.015	-.029	-.029	-.096	.871
M = 0.900; q = 713 lb/sq ft										
.166	.011	.024	.033	.043	.054	.072	.083	.093	.098	.166
.277										.277
.367	.028	.052	.076	.099	.121	.166	.211	.253	.289	.367
.387	-.080	-.032	.013	.057	.098	.179	.259	.328	.387	.387
.443	-.217	-.149	-.087	-.029	.028	.145	.256	.356	.453	.443
.498	-.212	-.148	-.090	-.034	.018	.127	.228	.320	.415	.498
.553	-.248	-.169	-.107	-.057	-.012	.083	.164	.247	.333	.553
.609	-.175	-.124	-.081	-.043	-.007	.065	.120	.178	.248	.609
.664	-.074	-.055	-.035	-.016	.004	.044	.052	.073	.120	.664
.719	-.015	-.008	.000	.008	.019	.041	-.009	-.082	-.105	.719
.774	.007	.012	.016	.022	.029	.049	-.015	-.114	-.214	.774
.830	.012	.017	.020	.022	.026	.039	-.005	-.072	-.200	.830
.871	-.014	-.013	-.016	-.015	-.014	-.011	-.048	-.104	-.184	.871
M = 0.940; q = 743 lb/sq ft										
.166	.013	.023	.033	.041	.054	.071	.088	.100	.109	.166
.277										.277
.367	.051	.070	.087	.108	.133	.176	.227	.272	.313	.367
.387	-.055	-.015	.024	.065	.109	.190	.275	.348	.413	.387
.443	-.208	-.148	-.092	-.034	.028	.148	.266	.373	.476	.443
.498	-.212	-.147	-.098	-.046	.014	.124	.234	.336	.437	.498
.553	-.250	-.208	-.157	-.084	-.027	.071	.167	.262	.354	.553
.609	-.312	-.241	-.106	-.063	-.023	.046	.120	.195	.276	.609
.664	-.142	-.049	-.039	-.024	-.014	.005	.043	.091	.152	.664
.719	.002	-.000	.003	.013	.017	-.011	-.044	-.087	-.071	.719
.774	.014	.014	.018	.026	.040	.039	-.041	-.191	-.222	.774
.830	.012	.016	.018	.023	.035	.041	-.031	-.160	-.269	.830
.871	-.013	-.016	-.017	-.016	-.010	-.014	-.070	-.153	-.406	.871
M = 0.980; q = 771 lb/sq ft										
.166	.019	.033	.043	.052	.062	.079	.099			.166
.277										.277
.367	.084	.102	.119	.137	.156	.199	.253			.367
.387	-.018	.021	.058	.095	.133	.215	.302			.387
.443	-.175	-.122	-.068	-.014	.043	.165	.288			.443
.498	-.192	-.130	-.073	-.027	.021	.136	.254			.498
.553	-.233	-.183	-.140	-.101	-.048	.072	.182			.553
.609	-.301	-.245	-.195	-.141	-.066	.041	.136			.609
.664	-.320	-.269	-.207	-.150	-.071	-.011	.056			.664
.719	-.093	-.047	-.028	-.028	-.071	-.121	-.107			.719
.774	.020	.040	.045	.034	.015	-.070	-.090			.774
.830	.027	.042	.047	.044	.039	.006	-.070			.830
.871	.008	.010	.009	.004	.001	-.026	-.143			.871

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(d) Station D - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 803 lb/sq ft										
.166	.061	.072	.086	.093	.102	.114	.127			.166
.277										.277
.347	.138	.137	.146	.162	.186	.249	.303			.347
.387	.048	.073	.105	.139	.178	.267	.353			.387
.443	-.102	-.067	-.018	.032	.091	.215	.337			.443
.498	-.124	-.082	-.030	.024	.076	.186	.303			.498
.553	-.169	-.136	-.093	-.050	.001	.120	.230			.553
.609	-.226	-.189	-.143	-.087	-.025	.090	.187			.609
.664	-.254	-.200	-.158	-.098	-.048	.046	.115			.664
.719	-.140	-.107	-.091	-.075	-.063	-.066	-.060			.719
.774	-.060	-.045	-.021	-.033	-.041	-.075	-.084			.774
.830	-.044	-.038	-.025	-.028	-.034	-.054	-.075			.830
.871	-.073	-.084	-.085	-.099	-.116	-.156	-.196			.871
M = 1.125; q = 856 lb/sq ft										
.166	.025	.042	.055	.067	.076	.085	.098			.166
.277										.277
.367	.008	.006	.003	-.006	-.022	-.056	-.075			.367
.387	.027	.055	.077	.102	.125	.189	.311			.387
.443	-.069	-.030	.005	.049	.091	.199	.342			.443
.498	-.097	-.050	-.012	.039	.089	.196	.318			.498
.553	-.131	-.082	-.056	-.010	.031	.125	.257			.553
.609	-.182	-.139	-.099	-.052	-.004	.115	.235			.609
.664	-.193	-.144	-.106	-.059	-.008	.090	.190			.664
.719	-.116	-.095	-.079	-.061	-.042	-.003	.023			.719
.774	-.045	-.034	-.031	-.027	-.026	-.033	-.035			.774
.830	-.020	-.011	-.008	-.009	-.011	-.023	-.036			.830
.871	-.057	-.084	-.079	-.083	-.096	-.130	-.161			.871
M = 1.200; q = 888 lb/sq ft										
.166	.042	.055	.067	.077	.083	.093	.101			.166
.277										.277
.367	-.008	-.011	-.019	-.022	-.029	-.046	-.057			.367
.387	.038	.062	.081	.100	.123	.169	.215			.387
.443	-.059	-.018	.018	.057	.104	.205	.337			.443
.498	-.078	-.035	.007	.052	.102	.211	.328			.498
.553	-.109	-.068	-.030	.012	.097	.162	.262			.553
.609	-.139	-.113	-.070	-.028	.013	.104	.223			.609
.664	-.180	-.141	-.101	-.054	-.007	.094	.204			.664
.719	-.115	-.091	-.075	-.055	-.035	.013	.064			.719
.774	-.059	-.053	-.045	-.038	-.032	-.020	-.018			.774
.830	-.016	-.014	-.012	-.014	-.018	-.024	-.040			.830
.871	-.055	-.060	-.069	-.082	-.099	-.135	-.166			.871

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(e) Station E

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 627 lb/sq ft										
.055	.027	.045	.065	.088	.113	.175	.248	.321	.406	.055
.166										.166
.277	-.023	-.016	-.004	.009	.025	.074	.140	.210	.294	.277
.367	-.002	.029	.065	.102	.138	.225	.317	.399	.490	.367
.387	-.084	-.048	-.013	.022	.055	.138	.232	.319	.414	.387
.443	-.175	-.120	-.070	-.025	.020	.120	.222	.305	.397	.443
.498	-.187	-.131	-.080	-.035	.007	.104	.195	.273	.356	.498
.553	-.164	-.114	-.069	-.028	.011	.096	.174	.236	.304	.553
.609	-.118	-.084	-.049	-.019	.011	.079	.135	.178	.228	.609
.664	-.046	-.026	-.003	.016	.036	.082	.112	.122	.142	.664
.719	-.027	-.020	-.011	-.001	.006	.031	.034	.007	-.031	.719
.774	.002	.005	.010	.013	.017	.032	.026	-.008	-.070	.774
.830	.007	.007	.009	.010	.011	.020	.016	.000	-.043	.830
.871	-.017	-.018	-.022	-.027	-.032	-.036	-.046	-.062	-.100	.871
.954	.058	.057	.055	.051	.047	.040	.034	.039	.024	.954
M = 0.900; q = 713 lb/sq ft										
.055	.031	.049	.069	.094	.121	.183	.256	.335	.422	.055
.166										.166
.277	-.023	-.019	-.012	.003	.021	.072	.139	.218	.308	.277
.367	.029	.053	.080	.115	.152	.237	.332	.420	.516	.367
.387	-.062	-.034	-.004	.030	.066	.151	.247	.341	.441	.387
.443	-.191	-.134	-.084	-.033	.017	.125	.232	.327	.427	.443
.498	-.225	-.160	-.104	-.052	-.002	.103	.201	.292	.386	.498
.553	-.257	-.169	-.106	-.054	-.005	.091	.173	.250	.336	.553
.609	-.170	-.118	-.076	-.037	-.001	.073	.130	.191	.263	.609
.664	-.056	-.036	-.013	.010	.033	.080	.099	.128	.178	.664
.719	-.028	-.021	-.012	-.002	.010	.035	.001	-.045	-.035	.719
.774	.002	.004	.008	.014	.021	.040	-.013	-.093	-.153	.774
.830	.003	.003	.005	.007	.013	.027	-.012	-.068	-.150	.830
.871	-.022	-.024	-.027	-.031	-.032	-.034	-.070	-.122	-.195	.871
.954	.060	.058	.056	.053	.051	.048	.034	.024	-.012	.954
M = 0.940; q = 743 lb/sq ft										
.055	.036	.052	.073	.094	.126	.187	.262	.343	.437	.055
.166										.166
.277	-.023	-.022	-.015	-.004	.019	.070	.142	.225	.320	.277
.367	.056	.073	.095	.124	.163	.245	.342	.437	.536	.367
.387	-.039	-.018	.006	.037	.077	.160	.259	.359	.463	.387
.443	-.180	-.134	-.086	-.038	.019	.129	.242	.340	.451	.443
.498	-.225	-.163	-.118	-.066	-.008	.101	.208	.307	.409	.498
.553	-.258	-.210	-.150	-.084	-.022	.080	.178	.268	.359	.553
.609	-.300	-.237	-.105	-.060	-.017	.055	.132	.206	.288	.609
.664	-.185	-.031	-.017	.001	.018	.048	.091	.143	.207	.664
.719	-.007	-.013	-.009	.001	.008	-.005	-.027	-.039	-.002	.719
.774	.009	.006	.010	.017	.031	.029	-.040	-.151	-.151	.774
.830	.003	.002	.004	.008	.020	.026	-.039	-.156	-.189	.830
.871	-.024	-.026	-.030	-.033	-.030	-.037	-.091	-.176	-.370	.871
.954	.061	.059	.058	.055	.055	.047	.034	.012	-.023	.954
M = 0.980; q = 771 lb/sq ft										
.055	.048	.067	.089	.112	.138	.200	.277			.055
.166										.166
.277	-.016	-.016	-.008	.004	.022	.075	.155			.277
.367	.096	.110	.131	.156	.186	.264	.365			.367
.387	-.002	.017	.040	.067	.099	.181	.285			.387
.443	-.149	-.106	-.062	-.016	.033	.145	.265			.443
.498	-.205	-.151	-.096	-.050	-.002	.111	.229			.498
.553	-.235	-.184	-.138	-.096	-.042	.081	.195			.553
.609	-.287	-.237	-.189	-.138	-.063	.048	.147			.609
.664	-.285	-.228	-.174	-.117	-.034	.032	.103			.664
.719	-.110	-.058	-.032	-.028	-.071	-.103	-.070			.719
.774	.022	.037	.038	.024	.006	-.082	-.090			.774
.830	.024	.032	.033	.029	.023	-.006	-.057			.830
.871	.002	.002	-.004	-.014	-.020	-.049	-.169			.871
.954	.081	.079	.076	.071	.068	.062	.028			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Concluded

(e) Station E - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 803 lb/sq ft										
.055	.093	.106	.128	.150	.177	.239	.315			.055
.166										.166
.277	-.007	-.003	.014	.022	.039	.073	.184			.277
.367	.153	.147	.156	.175	.210	.314	.412			.367
.387	.064	.070	.088	.111	.144	.234	.334			.387
.443	-.079	-.052	-.012	.030	.083	.196	.315			.443
.498	-.136	-.099	-.053	-.001	.052	.161	.278			.498
.553	-.163	-.128	-.086	-.040	.012	.129	.244			.553
.609	-.213	-.179	-.134	-.084	-.024	.098	.199			.609
.664	-.220	-.169	-.126	-.063	-.014	.088	.162			.664
.719	-.174	-.130	-.112	-.084	-.062	-.046	-.013			.719
.774	-.060	-.052	-.027	-.041	-.050	-.079	-.068			.774
.830	-.055	-.053	-.041	-.041	-.046	-.058	-.063			.830
.871	-.086	-.100	-.106	-.124	-.144	-.179	-.194			.871
.954	-.040	-.055	-.055	-.073	-.089	-.106	-.126			.954
M = 1.125; q = 856 lb/sq ft										
.055	.062	.083	.103	.126	.149	.208	.285			.055
.166										.166
.277	-.003	.002	.012	.027	.041	.075	.126			.277
.367	.002	.007	.014	.028	.041	.078	.139			.367
.387	.019	.024	.027	.030	.032	.046	.210			.387
.443	-.049	-.027	.000	.040	.080	.181	.315			.443
.498	-.098	-.057	-.019	.028	.072	.172	.299			.498
.553	-.125	-.081	-.044	-.000	.043	.143	.276			.553
.609	-.164	-.120	-.083	-.037	.009	.118	.246			.609
.664	-.164	-.119	-.079	-.030	.019	.124	.231			.664
.719	-.168	-.130	-.105	-.070	-.039	.028	.074			.719
.774	-.050	-.045	-.043	-.038	-.036	-.034	-.014			.774
.830	-.027	-.024	-.022	-.020	-.020	-.021	-.006			.830
.871	-.069	-.082	-.094	-.110	-.121	-.141	-.138			.871
.954	.020	-.014	-.027	-.043	-.055	-.074	-.077			.954
M = 1.200; q = 888 lb/sq ft										
.055	.078	.095	.115	.137	.164	.226	.300			.055
.166										.166
.277	.009	.016	.024	.034	.048	.089	.140			.277
.367	.009	.015	.021	.036	.057	.093	.150			.367
.387	.009	.009	.010	.007	.010	.039	.085			.387
.443	-.035	-.009	.019	.050	.093	.191	.314			.443
.498	-.081	-.043	-.003	.040	.087	.190	.308			.498
.553	-.100	-.063	-.022	.020	.066	.167	.280			.553
.609	-.133	-.107	-.071	-.027	.023	.124	.233			.609
.664	-.146	-.105	-.058	-.007	.040	.136	.239			.664
.719	-.167	-.134	-.101	-.067	-.031	.044	.124			.719
.774	-.061	-.061	-.059	-.053	-.043	-.017	.013			.774
.830	-.016	-.017	-.013	-.011	-.007	.006	.013			.830
.871	-.067	-.079	-.091	-.102	-.113	-.128	-.121			.871
.954	-.034	-.044	-.054	-.062	-.070	-.082	-.088			.954

TABLE VII.- WING SECTION DATA

α, deg	$\frac{y}{b/2}$ = 0.12				$\frac{y}{b/2}$ = 0.29						$\frac{y}{b/2}$ = 0.40					
	c_n		c_m		$\Delta\alpha$, deg		c_n		c_m		$\Delta\alpha$, deg		c_n		c_m	
	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm
M = 0.800																
-4	-0.2066	-0.2073	0.0076	0.0075	0.03	0.05	-0.2268	-0.2262	-0.0076	-0.0069	0.06	0.13	-0.2825	-0.2828	-0.0241	-0.0257
-2	-.0946	-.0879	-.0096	-.0064	.01	.03	-.0989	-.0874	-.0149	-.0136	.04	.08	-.1271	-.1183	-.0241	-.0249
0	.0221	.0258	-.0186	-.0185	0	-.01	.0366	.0398	-.0214	-.0215	0	0	.0263	.0310	-.0284	-.0292
2	.1276	.1358	-.0302	-.0300	-.01	-.03	.1544	.1600	-.0285	-.0287	-.03	-.07	.1613	.1714	-.0339	-.0350
4	.2459	.2430	-.0433	-.0416	-.02	-.05	.2858	.2836	-.0364	-.0358	-.06	-.13	.3167	.3114	-.0399	-.0393
8	.4710	.4733	-.0661	-.0651	-.05	-.09	.5379	.5446	-.0481	-.0449	-.12	-.24	.6183	.6338	-.0388	-.0389
12	.6776	.6881	-.0850	-.0846	-.06	-.11	.7886	.7900	-.0661	-.0604	-.14	-.28	1.0187	1.0301	-.1371	-.1391
16	.9219	.9067	-.1198	-.1141	-.06	-.12	1.1074	1.0970	-.1423	-.1279	-.15	-.30	1.0446	1.0506	-.1839	-.1821
20	1.1304	1.1340	-.1572	-.1480	-.07	-.13	1.1800	1.1670	-.1968	-.1920	-.17	-.33	1.0240	1.0222	-.1879	-.1865
M = 0.900																
-4	-0.2283	-0.2252	0.0159	0.0153	0.03	0.06	-0.2579	-0.2493	-0.0003	-0.0016	0.09	0.17	-0.3279	-0.3134	-0.0133	-0.0210
-2	-.1039	-.0948	-.0032	-.0041	.02	.03	-.1063	-.0909	-.0136	-.0128	.05	.10	-.1388	-.1237	-.0231	-.0253
0	.0256	.0275	-.0217	-.0212	0	-.01	.0390	.0416	-.0248	-.0242	0	-.01	.0294	.0335	-.0326	-.0329
2	.1458	.1486	-.0393	-.0377	-.02	-.03	.1768	.1762	-.0380	-.0356	-.05	-.09	.1876	.1898	-.0419	-.0404
4	.2705	.2682	-.0581	-.0559	-.03	-.06	.3153	.3104	-.0504	-.0479	-.09	-.17	.3513	.3454	-.0496	-.0478
8	.5270	.5234	-.1050	-.1014	-.06	-.12	.6030	.6078	-.0850	-.0815	-.16	-.33	.6831	.6849	-.0737	-.0653
12	.7246	.7287	-.1195	-.1170	-.07	-.14	.8322	.8524	-.1056	-.0989	-.17	-.34	1.0447	1.0574	-.1657	-.1612
16	.9371	.9257	-.1534	-.1504	-.08	-.14	1.1119	1.1228	-.1706	-.1546	-.19	-.37	1.0513	1.0236	-.1993	-.1942
20	1.1366	1.1361	-.1805	-.1730	-.08	-.16	1.1869	1.1848	-.2171	-.2103	-.21	-.41	1.0802	1.0761	-.2068	-.2034
M = 0.940																
-4	-0.2479	-0.2491	0.0324	0.0341	0.04	0.07	-0.2674	-0.2689	0.0117	0.0102	0.09	0.20	-0.3333	-0.3411	0.0033	-0.0021
-2	-.1200	-.1013	.0035	-.0009	.02	.04	-.1302	-.1041	-.0074	-.0089	.07	.12	-.1691	-.1359	-.0182	-.0230
0	.0276	.0302	-.0238	-.0241	0	-.01	.0435	.0440	-.0285	-.0276	0	-.01	.0360	.0338	-.0356	-.0364
2	.1536	.1643	-.0508	-.0521	-.02	-.04	.1798	.1920	-.0472	-.0481	-.05	-.11	.2000	.2106	-.0528	-.0534
4	.2907	.2951	-.0809	-.0818	-.04	-.08	.3352	.3361	-.0695	-.0680	-.10	-.21	.3818	.3808	-.0739	-.0734
8	.5324	.5405	-.1255	-.1278	-.07	-.14	.5919	.6090	-.0988	-.1006	-.19	-.38	.6870	.7000	-.1011	-.0965
12	.7318	.7432	-.1427	-.1437	-.07	-.15	.8216	.8732	-.1221	-.1168	-.19	-.39	1.0207	1.0641	-.1685	-.1678
16	.9284	.9269	-.1561	-.1587	-.08	-.16	1.1347	1.1301	-.1701	-.1560	-.21	-.42	1.0804	1.0666	-.2073	-.2022
20	1.1483	1.1497	-.1957	-.1940	-.09	-.18	1.2858	1.2659	-.2250	-.2229	-.24	-.47	1.1590	1.1599	-.2308	-.2287
M = 0.980																
-4	-0.2365	-0.2275	0.0332	0.0312	0.04	0.08	-0.2703	-0.2632	0.0237	0.0216	0.12	0.23	-0.3335	-0.3253	0.0187	0.0135
-2	-.1066	-.0993	.0041	.0030	.02	.04	-.1142	-.1038	-.0049	-.0052	.07	.11	-.1495	-.1325	-.0136	-.0168
0	.0189	.0211	-.0215	-.0215	0	-.01	.0349	.0342	-.0291	-.0274	0	0	.0334	.0287	-.0427	-.0432
2	.1339	.1380	-.0440	-.0450	-.02	-.04	.1697	.1737	-.0521	-.0510	-.06	-.11	.1895	.1936	-.0666	-.0656
4	.2615	.2631	-.0731	-.0731	-.04	-.09	.3187	.3196	-.0770	-.0758	-.13	-.24	.3685	.3640	-.0905	-.0875
8	.5039	.5021	-.1181	-.1170	-.08	-.16	.5876	.5855	-.1105	-.1089	-.23	-.44	.6780	.6784	-.1165	-.1133
12	.7341	.7362	-.1571	-.1560	-.09	-.20	.8416	.8818	-.1456	-.1406	-.24	-.53	.9915	1.0182	-.1692	-.1693
16	.9983		-.1887		-.10		1.0829		-.1597		-.26		1.3547		-.2582	
20	1.1325		-.1975		-.11		1.2997		-.2042		-.28		1.3431		-.2598	
M = 1.030																
-4	-0.2132	-0.2116	0.0269	0.0269	0.04	0.09	-0.2376	-0.2432	0.0173	0.0173	0.12	0.24	-0.2871	-0.2930	0.0104	0.0095
-2	-.1097	-.0990	.0059	.0033	.02	.04	-.1209	-.1032	-.0017	-.0051	.07	.12	-.1503	-.1301	-.0138	-.0183
0	.0169	.0198	-.0204	-.0204	0	-.01	.0309	.0337	-.0280	-.0274	0	-.01	.0271	.0259	-.0414	-.0429
2	.1234	.1321	-.0413	-.0437	-.02	-.05	.1592	.1671	-.0499	-.0506	-.06	-.13	.1767	.1863	-.0644	-.0659
4	.2427	.2428	-.0665	-.0654	-.04	-.08	.2982	.2967	-.0723	-.0699	-.12	-.23	.3449	.3371	-.0875	-.0845
8	.4728	.4743	-.1116	-.1109	-.08	-.16	.5565	.5578	-.1076	-.1078	-.22	-.43	.6412	.6500	-.1169	-.1147
12	.7067	.7022	-.1516	-.1497	-.11	-.21	.8401	.8410	-.1451	-.1377	-.29	-.59	.9279	.9472	-.1602	-.1508
16	.9371		-.1874		-.12		1.1066		-.1807		-.31		1.3614		-.2914	
20	1.1422		-.2204		-.12		1.2849		-.2032		-.32		1.4526		-.2948	
M = 1.125																
-4	-0.2012	-0.1861	0.0241	0.0206	0.04	0.08	-0.2234	-0.2080	0.0181	0.0147	0.12	0.23	-0.2713	-0.2578	0.0117	0.0044
-2	-.0967	-.0687	.0031	-.0026	.02	.05	-.1042	-.0735	-.0035	-.0100	.07	.10	-.1349	-.0919	-.0167	-.0257
0	.0148	.0204	-.0194	-.0208	0	-.01	.0254	.0354	-.0244	-.0266	0	-.01	.0258	.0312	-.0416	-.0450
2	.1150	.1268	-.0391	-.0416	-.02	-.05	.1433	.1600	-.0440	-.0462	-.06	-.12	.1620	.1793	-.0639	-.0665
4	.2235	.2275	-.0612	-.0615	-.04	-.08	.2748	.2761	-.0669	-.0660	-.11	-.22	.3220	.3199	-.0868	-.0866
8	.4408	.4422	-.1035	-.1026	-.08	-.16	.5280	.5357	-.1049	-.1034	-.22	-.43	.6178	.6055	-.1220	-.1213
12	.6596	.6605	-.1428	-.1422	-.11	-.22	.7692	.7736	-.1424	-.1386	-.30	-.59	.8921	.9022	-.1546	-.1551
16	.8990		-.1819		-.12		1.0507		-.1894		-.31		1.2532		-.2651	
20	1.0810		-.2101		-.12		1.2052		-.1932		-.32		1.3606		-.2770	
M = 1.200																
-4	-0.1923	-0.1850	0.0236	0.0219	0.04	0.08	-0.2131	-0.2062	0.0147	0.0123	0.12	0.24	-0.2582	-0.2490	0.0089	0.0031
-2	-.0906	-.0807	.0022	.0007	.02	.04	-.0975	-.0883	-.0048	-.0069	.07	.13	-.1311	-.1136	-.0166	-.0204
0	.0081	.0150	-.0156	-.0162	0	0	.0210	.0297	-.0246	-.0254	.01	.01	.0104	.0209	-.0371	-.0400
2	.1013	.1074	-.0320	-.0339	-.02	-.04	.1338	.1445	-.0430	-.0454	-.05	-.11	.1448	.1574	-.0566	-.0593
4	.2003	.2077	-.0516	-.0534	-.04	-.08	.2535	.2629	-.0636	-.0661	-.11	-.21	.2856	.2930	-.0765	-.0783
8	.4045	.4122	-.0924	-.0924	-.07	-.15	.4941	.5063	-.1007	-.1009	-.21	-.41	.5667	.5733	-.1162	-.1153
12	.6195	.6275	-.1353	-.1342	-.11	-.21	.7209	.7265	-.1412	-.1379	-.29	-.58	.8460	.8588	-.1584	-.1567
16	.8195		-.1720		-.13		.9455		-.1843		-.35		1.0588		-.2089	
20	1.0172		-.2062		-.13		1.1739		-.2264		-.36		1.2701		-.2838	

TABLE VII.- WING SECTION DATA - Concluded

α, deg	$\frac{y}{b/2}$ = 0.60						$\frac{y}{b/2}$ = 0.80						$\frac{y}{b/2}$ = 0.90					
	Δα, deg		c_n		c_m		Δα, deg		c_n		c_m		Δα, deg		c_n		c_m	
	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm
M = 0.800																		
-4	0.17	0.35	-0.4023	-0.4086	-0.0025	0.0031	0.29	0.60	-0.4790	-0.4909	0.0739	0.0787	0.55	0.75	-0.1936	-0.1875	-0.0049	-0.0036
-2	.11	.21	-.1905	-.1855	-.0331	-.0281	.18	.34	-.5851	-.5496	.0249	.0055	.22	.48	-.2197	-.2776	.0087	.0294
0	-.01	-.02	.0168	.0188	-.0322	-.0328	-.04	-.09	-.0673	-.0643	-.0485	-.0518	-.13	-.25	-.1557	-.1508	-.0462	-.0397
2	-.09	-.19	.1660	.1759	-.0385	-.0387	-.17	-.36	.1568	.1429	-.0490	-.0498	-.26	-.55	.0497	.0586	-.0355	-.0374
4	-.17	-.34	.3442	.3329	-.0416	-.0418	-.29	-.57	.3383	.3292	-.0513	-.0498	-.40	-.78	.2218	.2097	-.0330	-.0334
8	-.30	-.60	.7952	.7876	-.0506	-.0905	-.48	-.99	.5389	.5488	-.0858	-.0866	-.62	-1.29	.3089	.3407	-.0340	-.0394
12	-.35	-.71	.8327	.8327	-.1448	-.1408	-.57	-1.14	.5507	.5433	-.0931	-.0910	-.72	-1.46	.3261	.3334	-.0423	-.0465
16	-.39	-.76	.8000	.7895	-.1391	-.1352	-.63	-1.24	.5989	.5875	-.1054	-.1012	-.81	-1.59	.3862	.3774	-.0510	-.0516
20	-.44	-.86	.8584	.8397	-.1522	-.1492	-.72	-1.42	.6803	.6730	-.1210	-.1202	-.94	-1.85	.4537	.4461	-.0626	-.0618
M = 0.900																		
-4	0.23	0.45	-0.4590	-0.4581	-0.0041	-0.0001	0.41	0.78	-0.5684	-0.5519	0.1023	.0972	0.50	0.96	-0.2004	-0.1966	-0.0045	-0.0057
-2	.15	.26	-.2105	-.1984	-.0320	-.0273	.27	.44	-.4353	-.3691	.0548	.0170	.36	.63	-.2573	-.2865	.0134	.0313
0	-.01	-.03	.0187	.0216	-.0381	-.0357	-.06	-.13	-.0668	-.0600	-.0546	.0552	-.15	-.33	-.1694	-.1472	-.0428	-.0417
2	-.13	-.25	.2014	.1987	-.0440	-.0428	-.25	-.47	.1756	.1688	-.0568	.0552	-.37	-.71	.0822	.0773	-.0406	-.0396
4	-.23	-.43	.3979	.3934	-.0442	-.0421	-.38	-.73	.3991	.3715	-.0644	.0543	-.50	-.97	.2680	.2381	-.0271	-.0320
8	-.42	-.87	.7828	.7820	-.0573	-.0437	-.70	-1.42	.8012	.8788	-.1256	.1171	-.92	-1.88	.3907	.4366	-.0513	-.0572
12	-.49	-.88	.8256	.8358	-.1507	-.1447	-.73	-1.44	.5935	.5890	-.1090	.1078	-.93	-1.86	.3699	.3742	-.0486	-.0532
16	-.50	-.96	.8647	.8369	-.1597	-.1512	-.83	-1.57	.6752	.6478	-.1245	.1199	-1.06	-2.04	.4295	.4164	-.0539	-.0563
20	-.56	-1.08	.9306	.8966	-.1766	-.1681	-.93	-1.79	.7622	.7373	-.1439	.1396	-1.21	-2.33	.5004	.4846	-.0671	-.0662
M = 0.940																		
-4	0.25	0.54	-0.4519	-0.4833	0.0111	0.0202	0.42	0.94	-0.5311	-0.5681	0.0873	0.1029	0.52	1.17	-0.2489	-0.2439	-0.0024	0.0004
-2	.20	.32	-.2576	-.2128	-.0194	-.0233	.39	.57	-.4910	-.4154	.0794	.0362	.55	.81	-.3093	-.2976	.0373	.0347
0	-.02	-.04	.0252	.0248	-.0387	-.0383	-.07	-.15	-.0597	-.0602	-.0555	-.0555	-.17	-.36	-.1602	-.1480	-.0406	-.0433
2	-.14	-.30	.2103	.2213	-.0483	-.0478	-.26	-.55	.1835	.1947	-.0542	-.0574	-.38	-.80	.0828	.0927	-.0401	-.0410
4	-.28	-.55	.4277	.4326	-.0650	-.0643	-.46	-.90	.4434	.4272	-.0593	-.0593	-.57	-1.14	.3066	.2758	-.0170	-.0242
8	-.51	-1.00	.8160	.8295	-.1178	-.0859	-.82	-1.63	.9120	.9169	-.1341	-.1298	-1.04	-2.11	.3615	.4218	-.0443	-.0533
12	-.50	-1.00	.8776	.9013	-.1576	-.1554	-.82	-1.67	.6613	.6473	-.1221	-.1248	-1.05	-2.17	.4102	.4245	-.0546	-.0608
16	-.56	-1.10	.9198	.8977	-.1761	-.1681	-.93	-1.83	.7478	.7214	-.1446	-.1413	-1.21	-2.38	.4825	.4607	-.0644	-.0623
20	-.65	-1.24	1.0020	.9712	-.1959	-.1866	-1.07	-2.08	.8458	.8057	-.1651	-.1583	-1.40	-2.74	.5748	.5464	-.0825	-.0785
M = 0.980																		
-4	0.33	0.67	-0.4971	-0.4876	0.0434	0.0419	0.58	1.22	-0.6175	-0.6253	0.1135	0.1202	0.75	1.63	-0.3163	-0.3397	0.0187	0.0360
-2	.19	.26	-.2452	-.2203	-.0086	-.0129	.31	.28	-.4576	-.3377	.0243	-.0536	.44	.58	-.3078	-.2976	.0432	.0385
0	-.01	-.04	.0143	.0118	-.0472	-.0459	-.05	-.13	-.0778	-.0490	-.0634	-.0601	-.03	-.20	-.2262	-.1969	.0265	.0002
2	-.19	-.35	.2112	.2122	-.0775	-.0726	-.35	-.63	.2330	.2059	-.1037	-.0769	-.39	-.77	.0070	.0440	.0320	.0041
4	-.39	-.79	.4403	.4145	-.1089	-.0945	-.75	-1.50	.4960	.4864	-.1452	-.1528	-1.13	-2.30	.4124	.4039	-.1112	-.1216
8	-.68	-1.29	.8606	.8314	-.1493	-.1382	-1.25	-2.35	1.0161	.9397	-.2221	-.1837	-1.73	-3.33	.7699	.7850	-.1071	-.1331
12	-.67	-1.44	1.0689	1.1779	-.1969	-.1907	-1.15	-2.56	.9065	.8735	-.1781	-.1701	-1.53	-3.49	.5892	.9278	-.0933	-.1163
16	-.70		1.0836		-.2072		-1.20		.8696		-.1851		-1.60		.6289		-.0980	
20	-.77		1.0955		-.2256		-1.33		.9699		-.2015		-1.79		.7157		-.1173	
M = 1.030																		
-4	0.34	0.71	-0.4244	-0.4534	0.0322	0.0400	0.62	1.26	-0.6247	-0.6360	0.0970	0.1002	0.82	1.69	-0.4222	-0.4083	0.0365	0.0408
-2	.20	.30	-.2363	-.2092	-.0136	-.0174	.34	.41	-.4590	-.3654	.0309	-.0274	.45	.49	-.3441	-.3375	.0194	.0173
0	-.02	-.07	.0203	.0225	-.0598	-.0600	-.11	-.30	-.0837	-.0867	-.0840	-.0877	-.25	-.75	-.2460	-.1823	-.0521	-.0908
2	-.21	-.44	.2105	.2166	-.0891	-.0881	-.46	-.93	.1820	.1935	-.1138	-.1121	-.78	-1.58	.1147	.1216	-.1090	-.1100
4	-.37	-.70	.4167	.3992	-.1134	-.1086	-.70	-1.36	.4312	.3982	-.1557	-.1340	-1.05	-2.06	.3136	.2902	-.0993	-.1045
8	-.67	-1.26	.8029	.7824	-.1479	-.1327	-1.25	-2.32	.9191	.8680	-.2040	-.1723	-1.79	-3.36	.7645	.7427	-.1469	-.1491
12	-.85	-1.68	1.1383	1.1345	-.2058	-.1936	-1.52	-3.02	1.1230	1.1015	-.2149	-.2028	-2.11	-4.35	.9718	1.0141	-.1552	-.1930
16	-.83		1.1777		-.2332		-1.42		.9816		-.2026		-1.92		.6970		-.1238	
20	-.86		1.2335		-.2472		-1.48		.9697		-.2105		-2.01		.7569		-.1345	
M = 1.125																		
-4	0.36	0.71	-0.3932	-0.3749	0.0369	0.0335	0.66	1.30	-0.5831	-0.5687	0.0802	0.0758	0.93	1.82	-0.5230	-0.5209	0.0623	0.0613
-2	.22	.26	-.2050	-.1487	-.0128	-.0238	.41	.36	-.4158	-.2837	.0284	-.0337	.59	.18	-.4825	-.4094	.0489	-.0555
0	-.02	-.07	.0060	.0159	-.0538	-.0533	-.10	-.30	-.0983	-.0901	-.0690	-.0755	-.29	-.91	-.1973	-.1595	-.0771	-.1315
2	-.18	-.39	.1726	.1930	-.0786	-.0798	-.41	-.85	.1216	.1527	-.1044	-.1045	-.76	-1.57	.0480	.0435	-.1282	-.1283
4	-.36	-.68	.3716	.3612	-.1088	-.1042	-.70	-1.34	.3825	.3504	-.1333	-.1267	-1.08	-2.11	.2785	.2513	-.1120	-.1143
8	-.66	-1.28	.7316	.7240	-.1464	-.1356	-1.23	-2.40	.8367	.8127	-.1922	-.1937	-1.76	-3.46	.6842	.6359	-.1539	-.1339
12	-.88	-1.73	1.0691	1.0366	-.2224	-.1961	-1.60	-3.19	1.0469	1.0587	-.2162	-.2343	-2.28	-4.56	.9493	.8926	-.1809	-.1725
16	-.85		1.1243		-.2259		-1.46		.9386		-.1958		-1.98		.7009		-.1237	
20	-.88		1.1881		-.2383		-1.53		.9532		-.2062		-2.11		.7629		-.1409	
M = 1.200																		
-4	0.55	0.72	-0.3484	-0.3554	0.0247	0.0183	0.69	1.34	-0.5100	-0.5359	0.0323	0.0414	0.99	2.03	-0.6956	-0.6919	0.1086	0.1052
-2	.20	.35	-.1838	-.1667	-.0250	-.0304	.35	.57	-.3601	-.3310	-.0091	-.0279	.50	.75	-.5557	-.5251	.0401	.0248
0	0	-.02	-.0004	.0148	-.0540	-.0558	-.05	-.12	-.1307	-.1109	-.0551	-.0627	-.21	-.52	-.2346	-.2349	-.0703	-.0753
2	-.17	-.35	.1688	.1792	-.0818	-.0818	-.37	-.77	.0838	.0931	-.0917	-.0909	-.73	-1.47	.0124	.0273	-.1306	-.1253
4	-.33	-.66	.3444	.3454	-.1085	-.1058	-.67	-1.31	.3145	.3045	-.1215	-.1183	-1.07	-2.11	.2482	.2337	-.1260	-.1202
8	-.62	-1.23	.6807	.6717	-.1491	-.1411	-1.17	-2.33	.7272	.7123	-.1785	-.1853	-1.69	-3.38	.6153	.5631	-.1312	-.1279
12	-.87	-1.70	1.0067	.9749	-.2250	-.1954	-1.60	-3.14	.9836	.9764	-.2215	-.2267	-2.28	-4.50	.8744	.8298	-.1718	-.1657
16	-1.00		1.1540		-.2418		-1.80		1.1027		-.2373		-2.54		.9254		-.1794	
20	-1.01		1.2121		-.2700		-1.74		1.0768		-.2284		-2.36		.7468		-.1427	

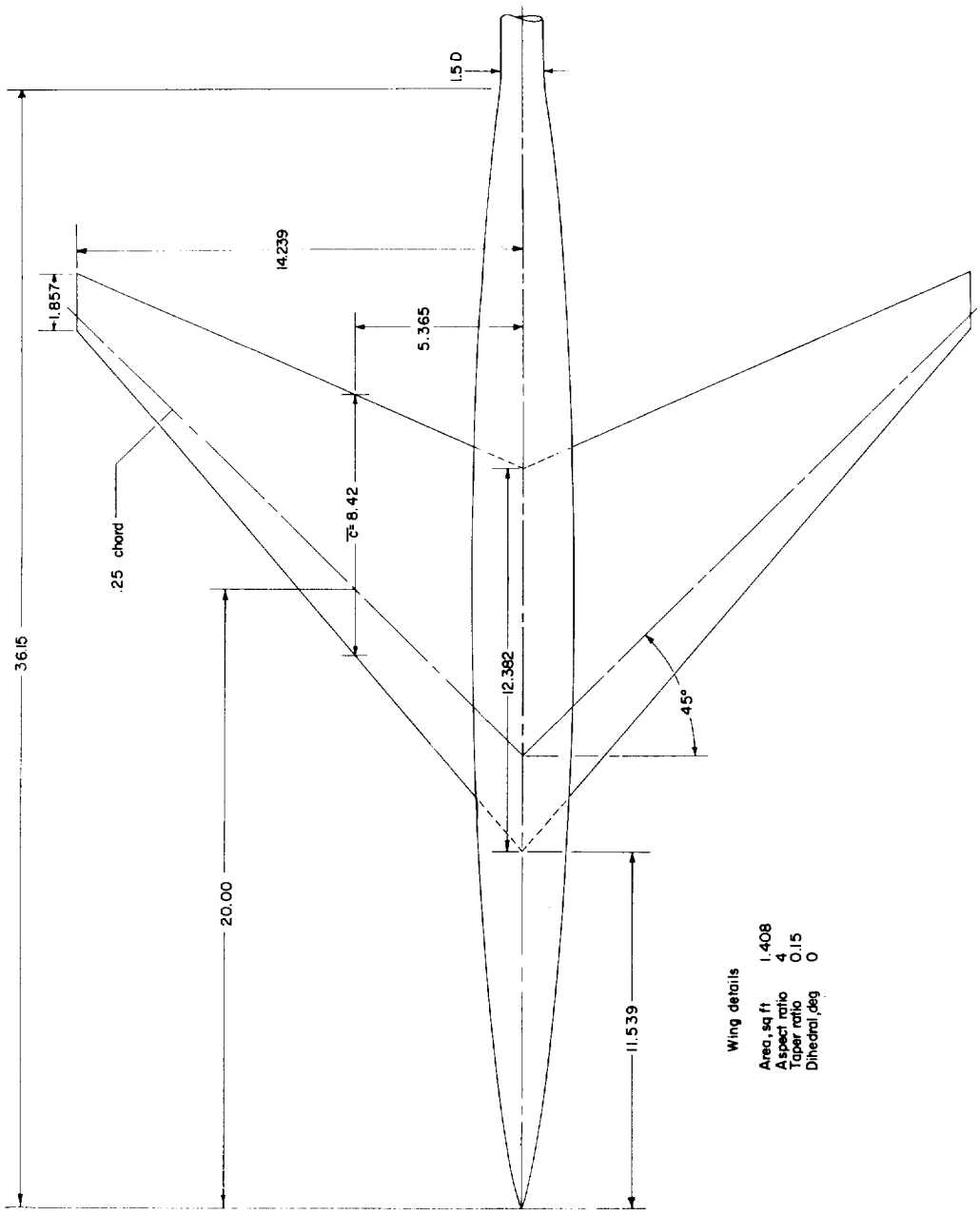


Figure 1.- Details of wing-body combination. (All dimensions in inches unless otherwise noted.)

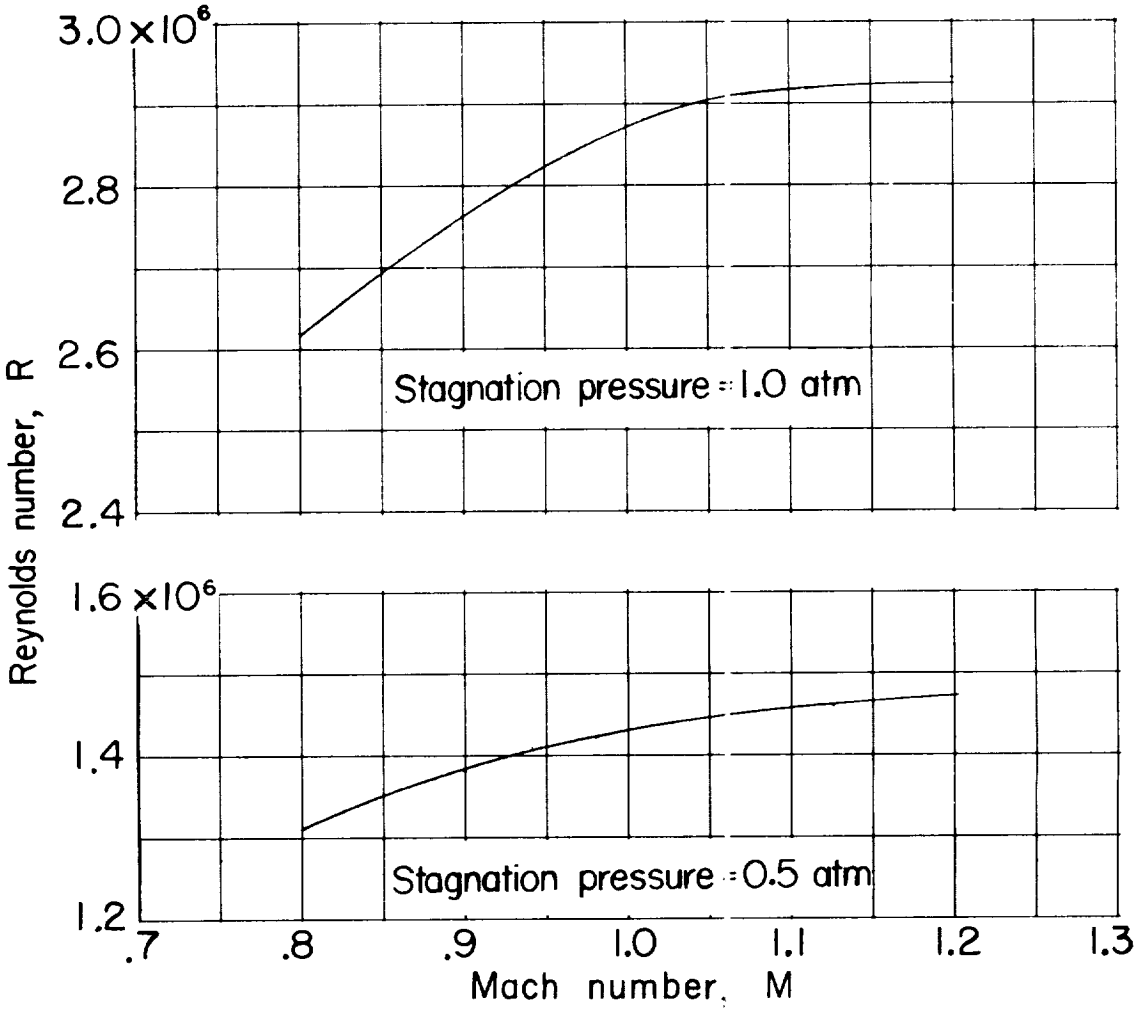


Figure 2.- Variation with Mach number of average Reynolds number based on wing mean aerodynamic chord.

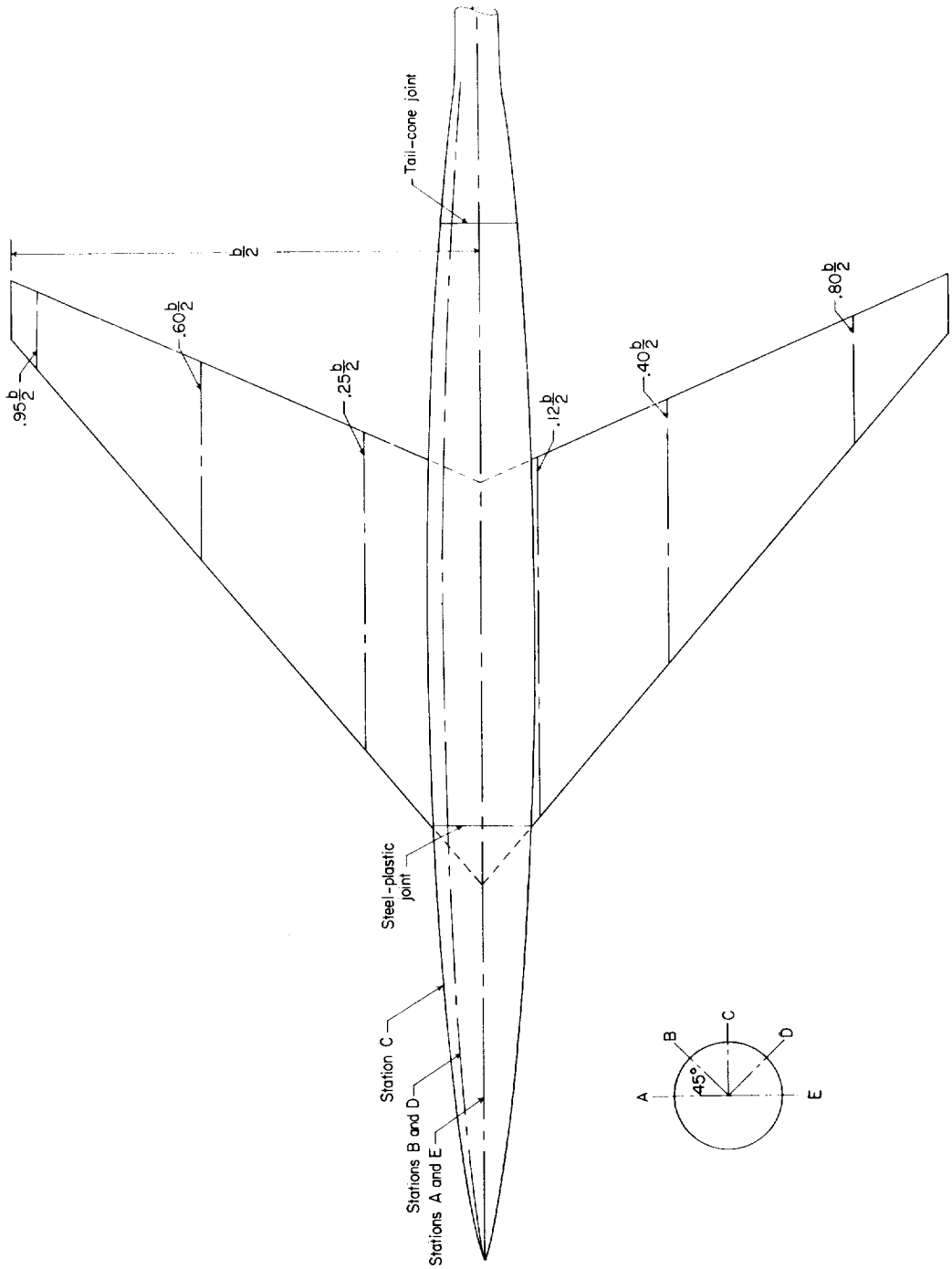
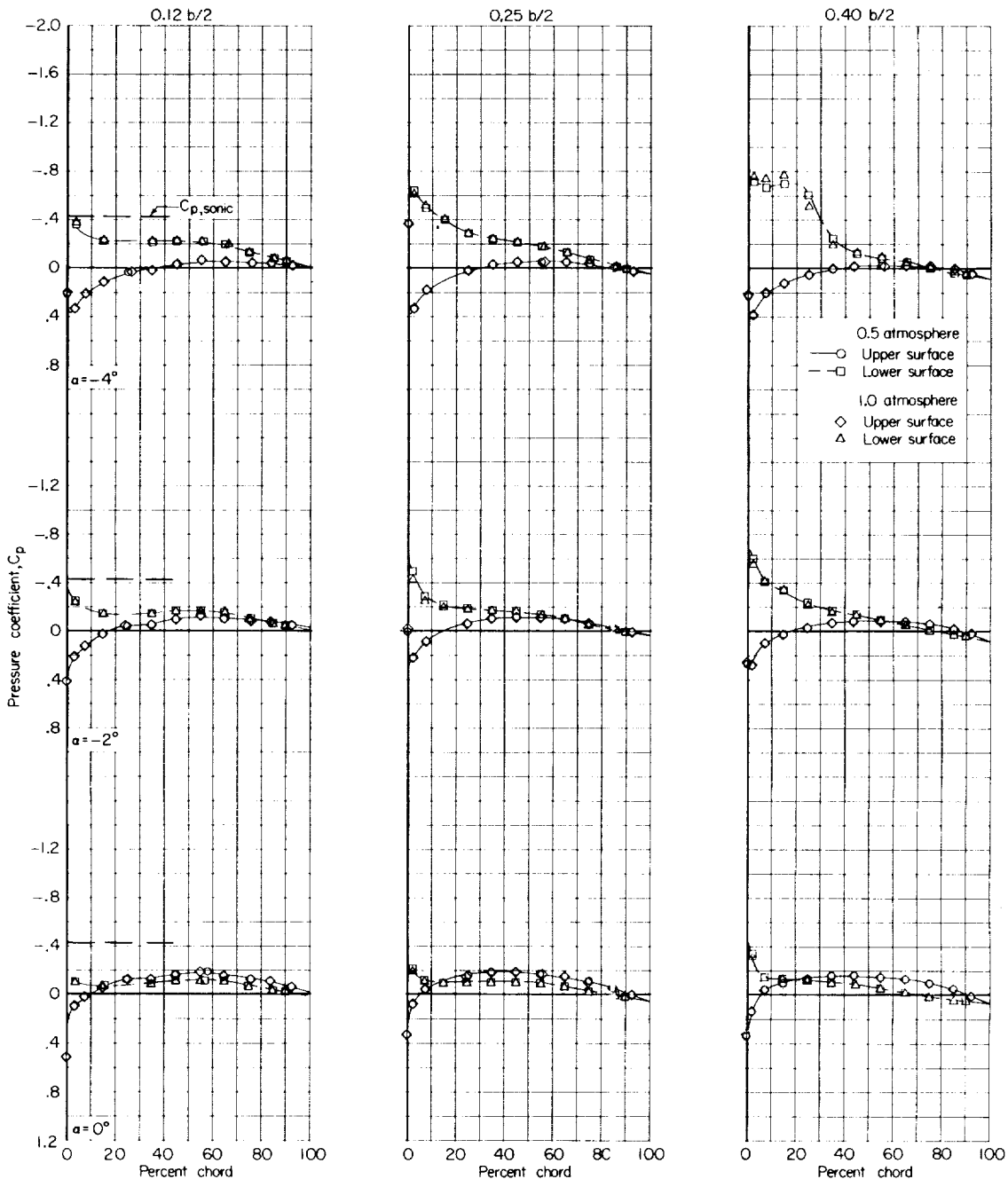
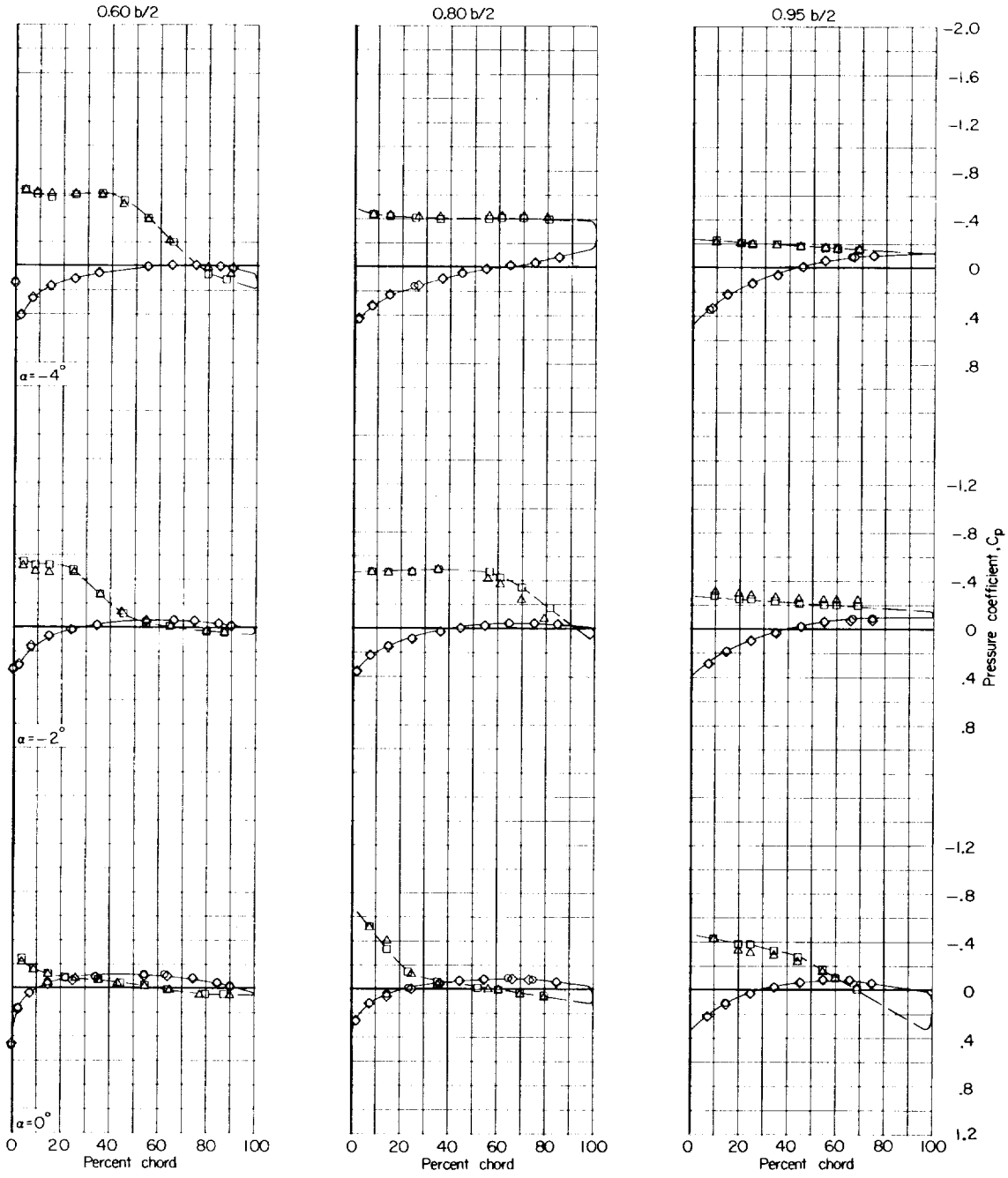


Figure 3.- A sketch of the location of pressure orifices on the wing and body.



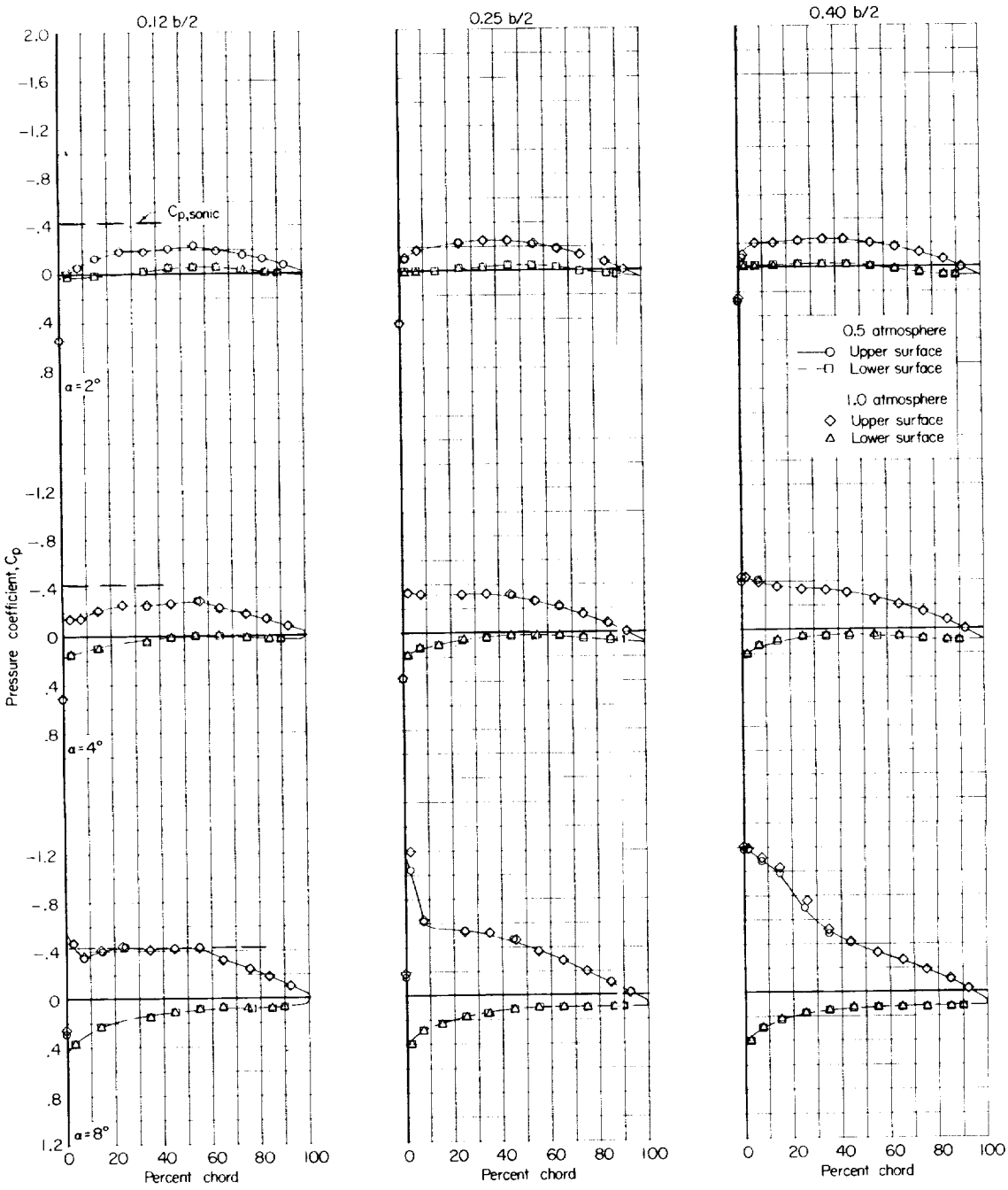
(a) M = 0.800; $\alpha = -4^{\circ}$, -2°, and 0°.

Figure 4.- Pressure measurements on the wing in the presence of the body.



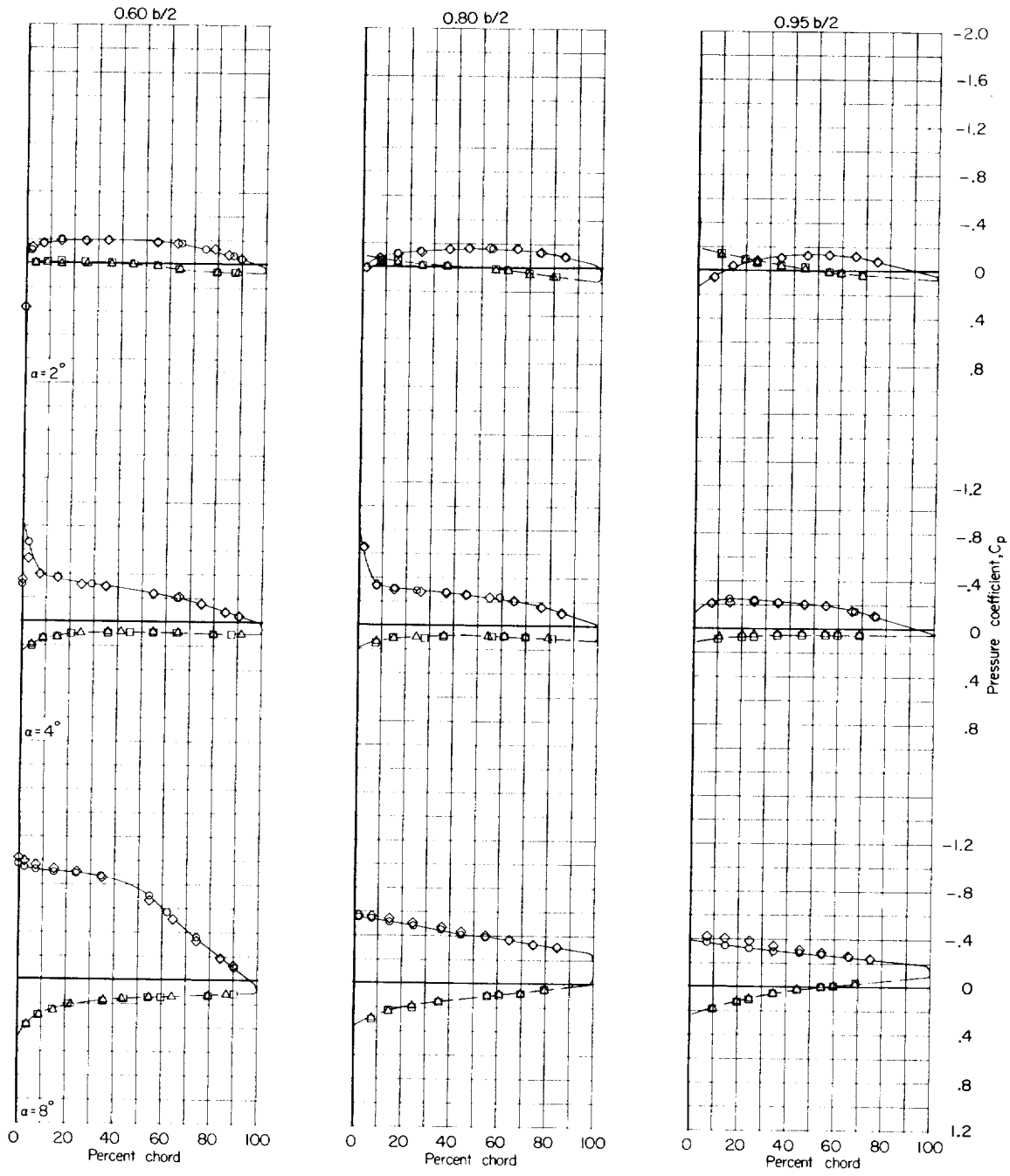
(a) Concluded.

Figure 4.- Continued.



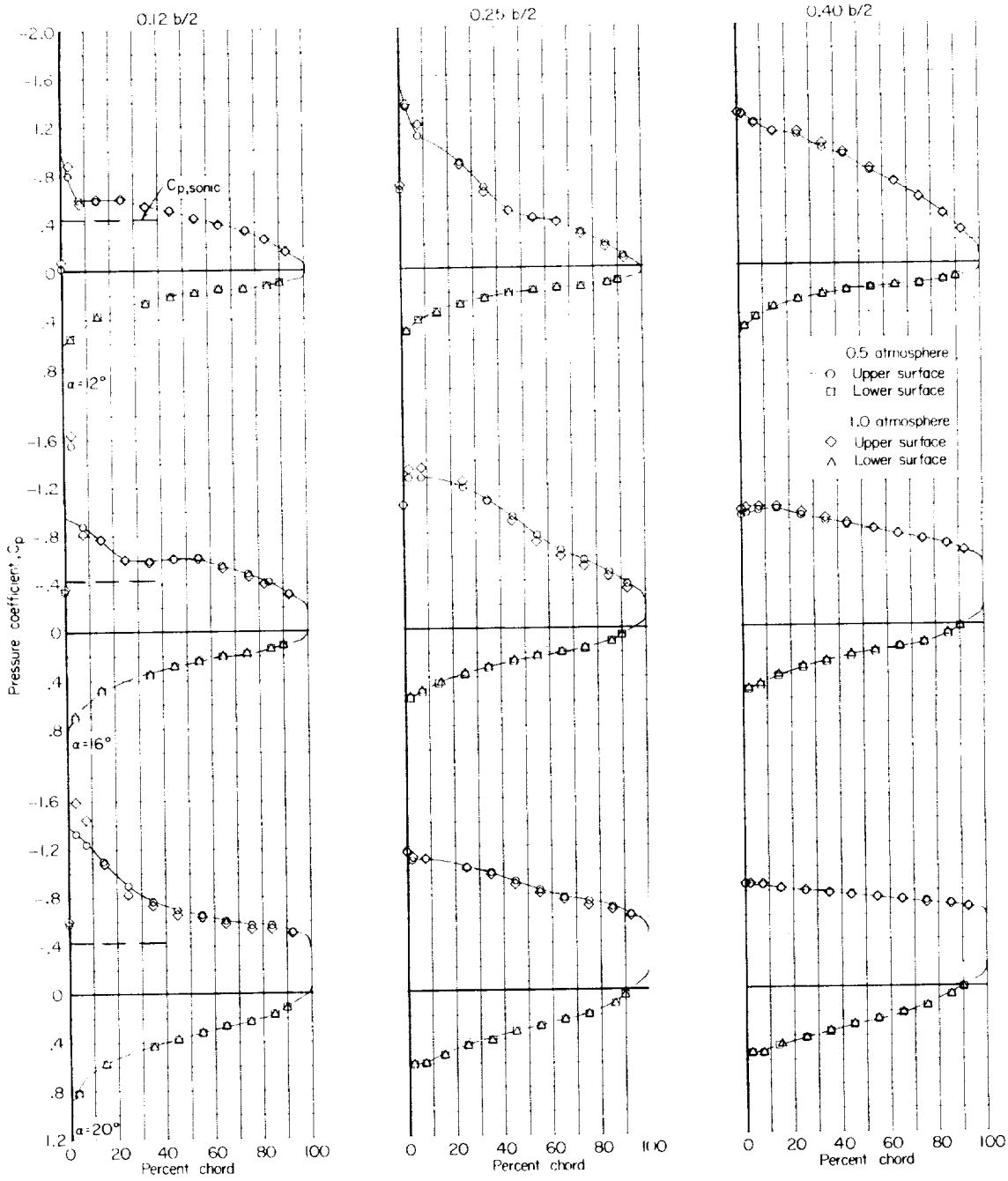
(b) M = 0.800; α = 2°, 4°, and 8°.

Figure 4.- Continued.



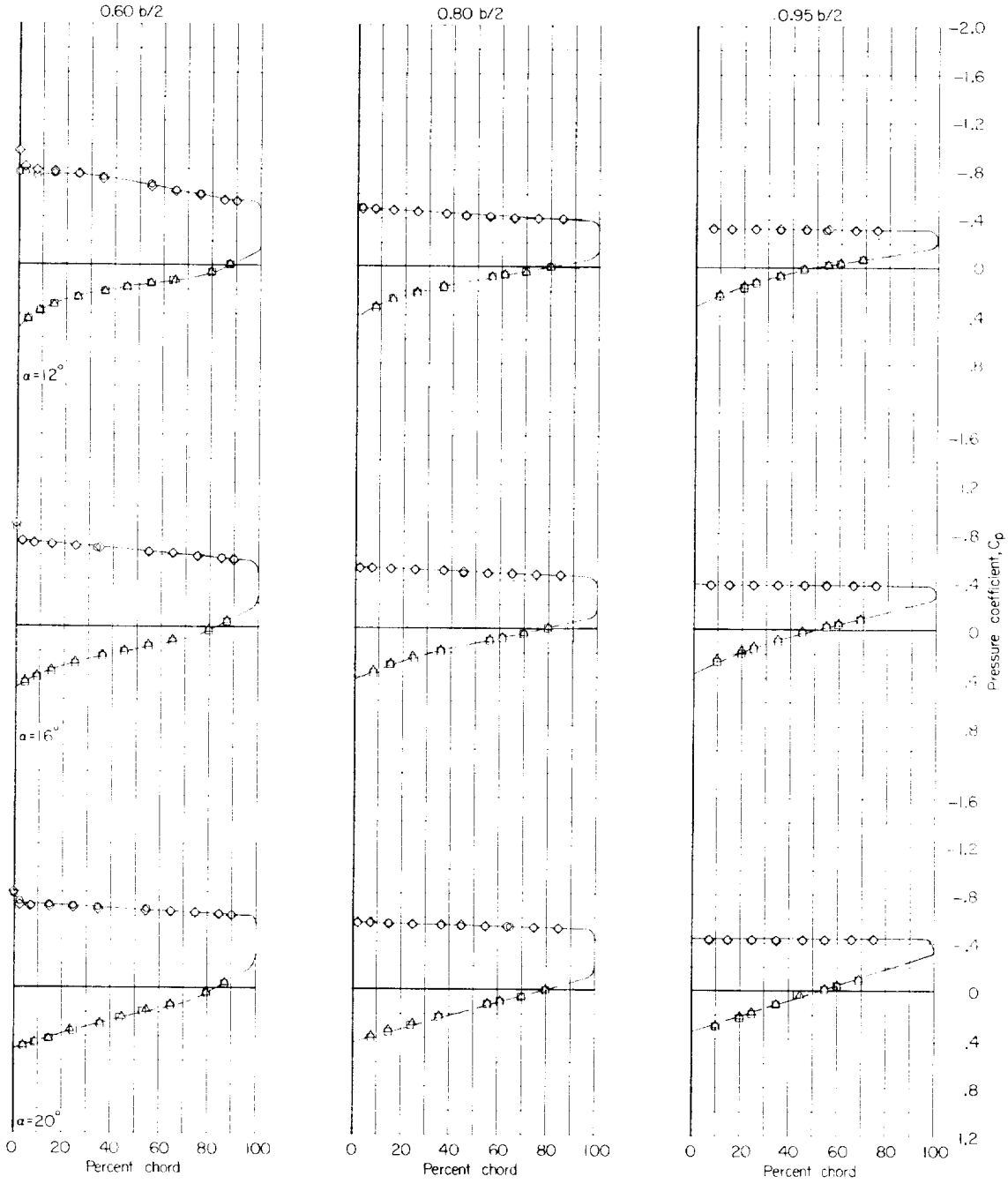
(b) Concluded.

Figure 4.- Continued.



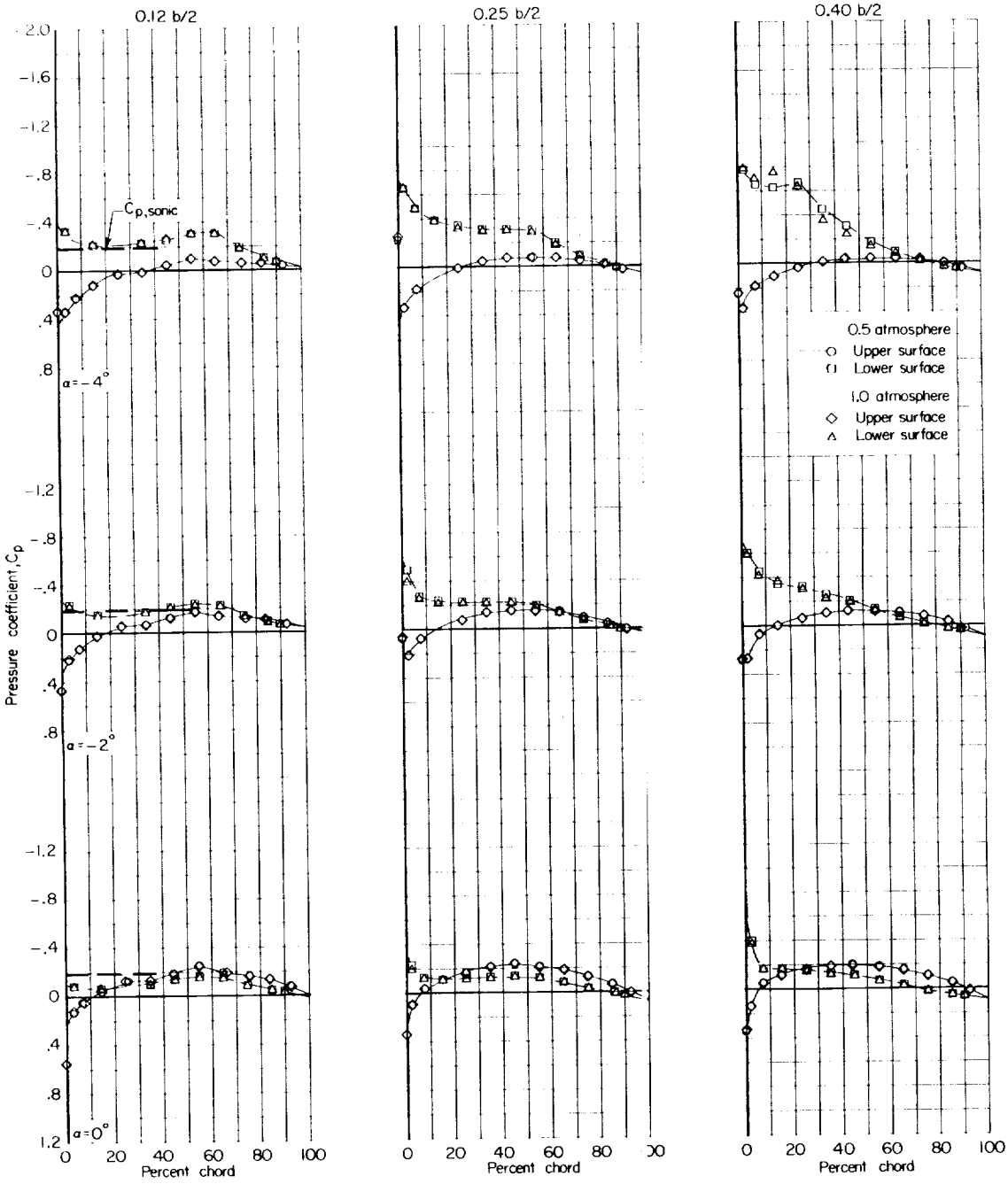
(c) M = 0.800; $\alpha = 12^{\circ}$, 16°, and 20°.

Figure 4.- Continued.



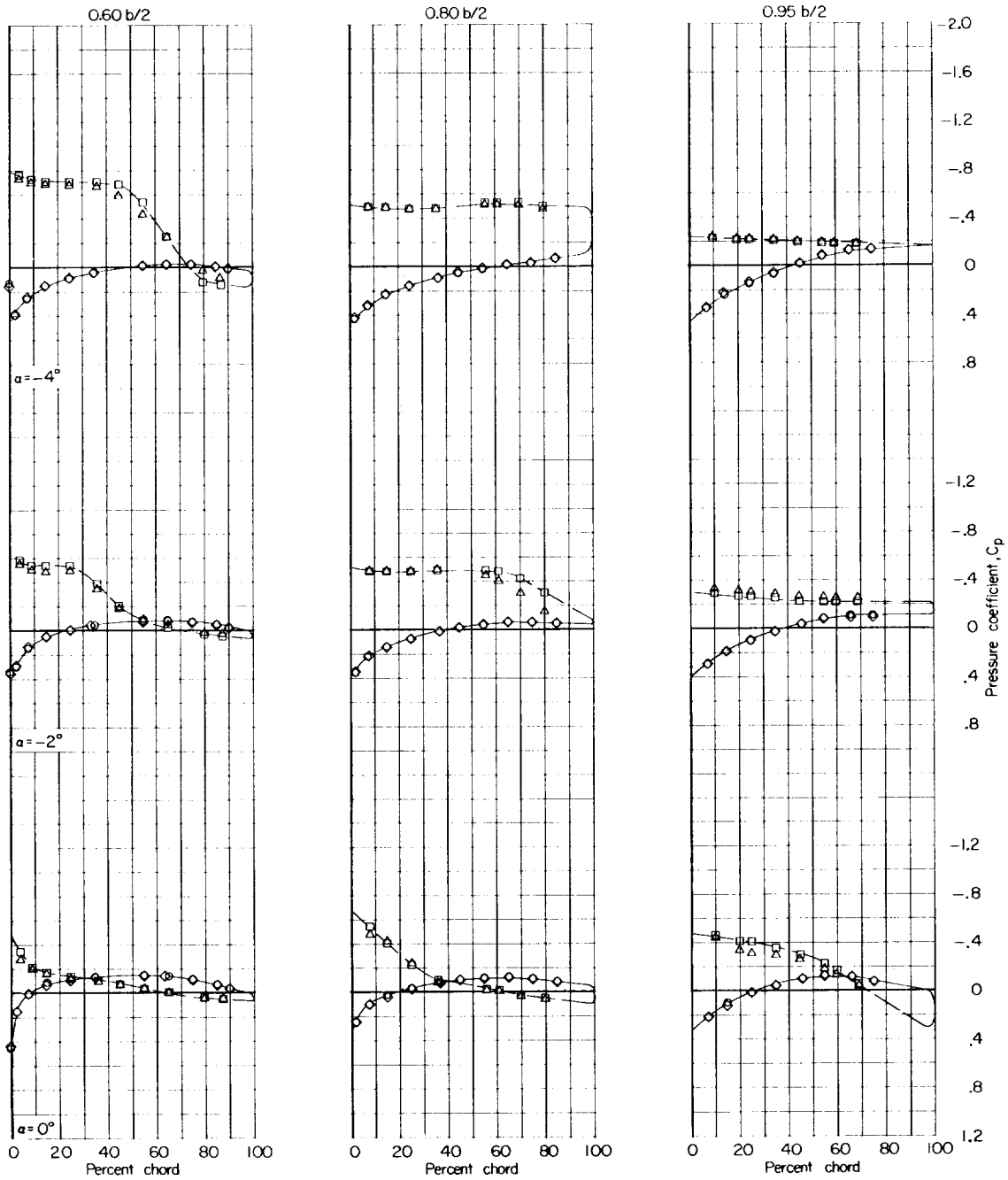
(c) Concluded.

Figure 4.- Continued.



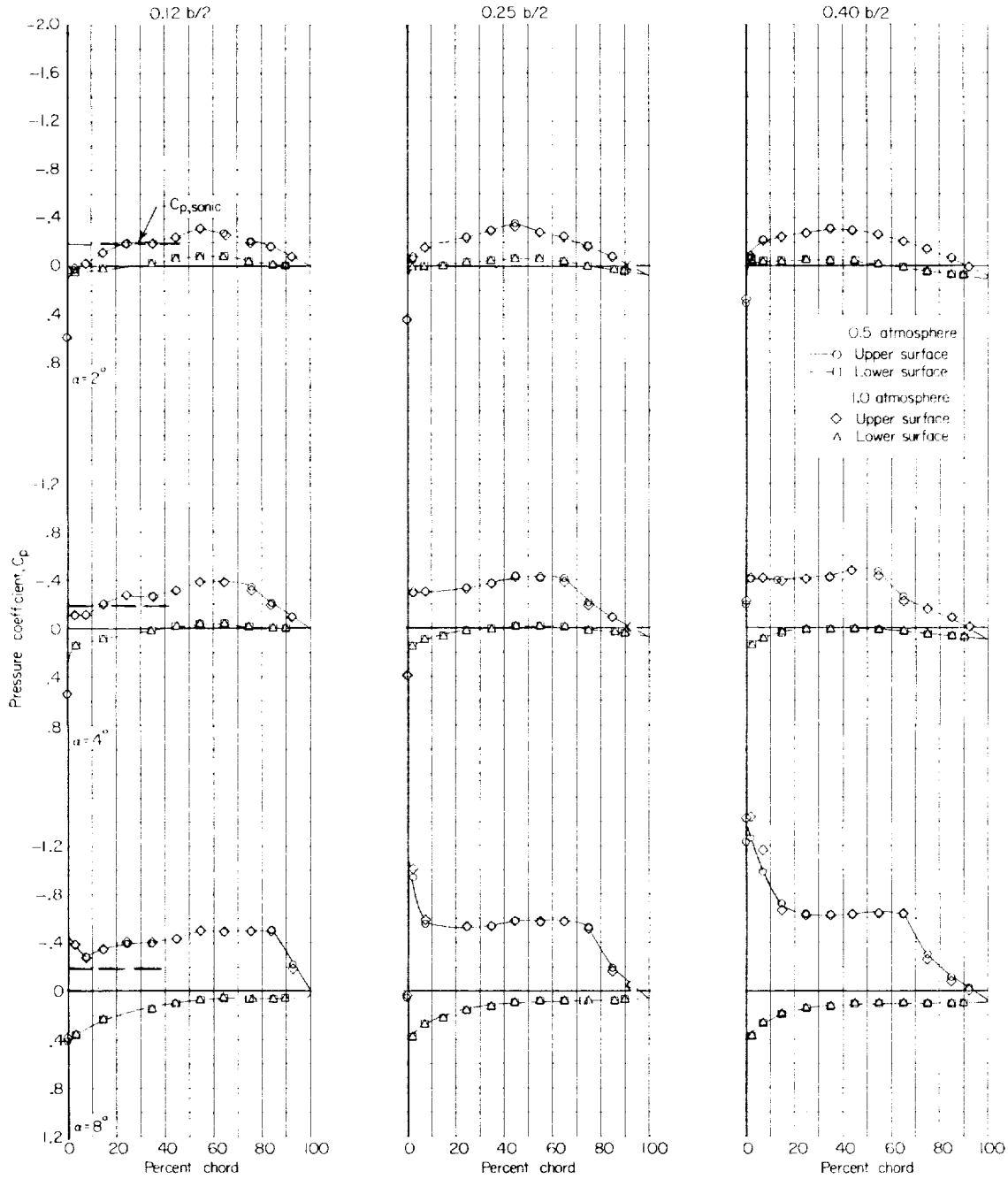
(d) M = 0.900; $\alpha = -4^{\circ}$, -2°, and 0°.

Figure 4.- Continued.



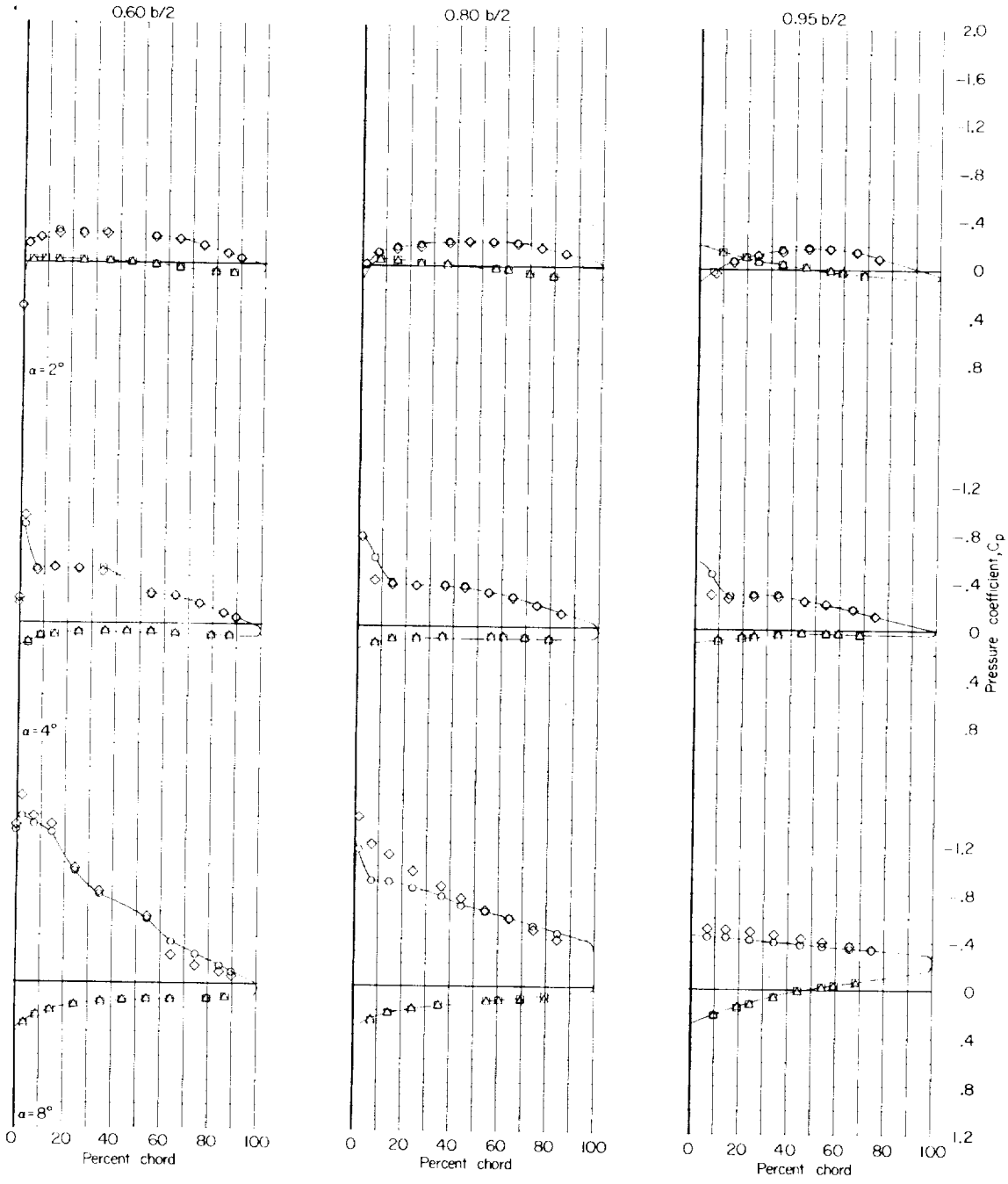
(d) Concluded.

Figure 4.- Continued.



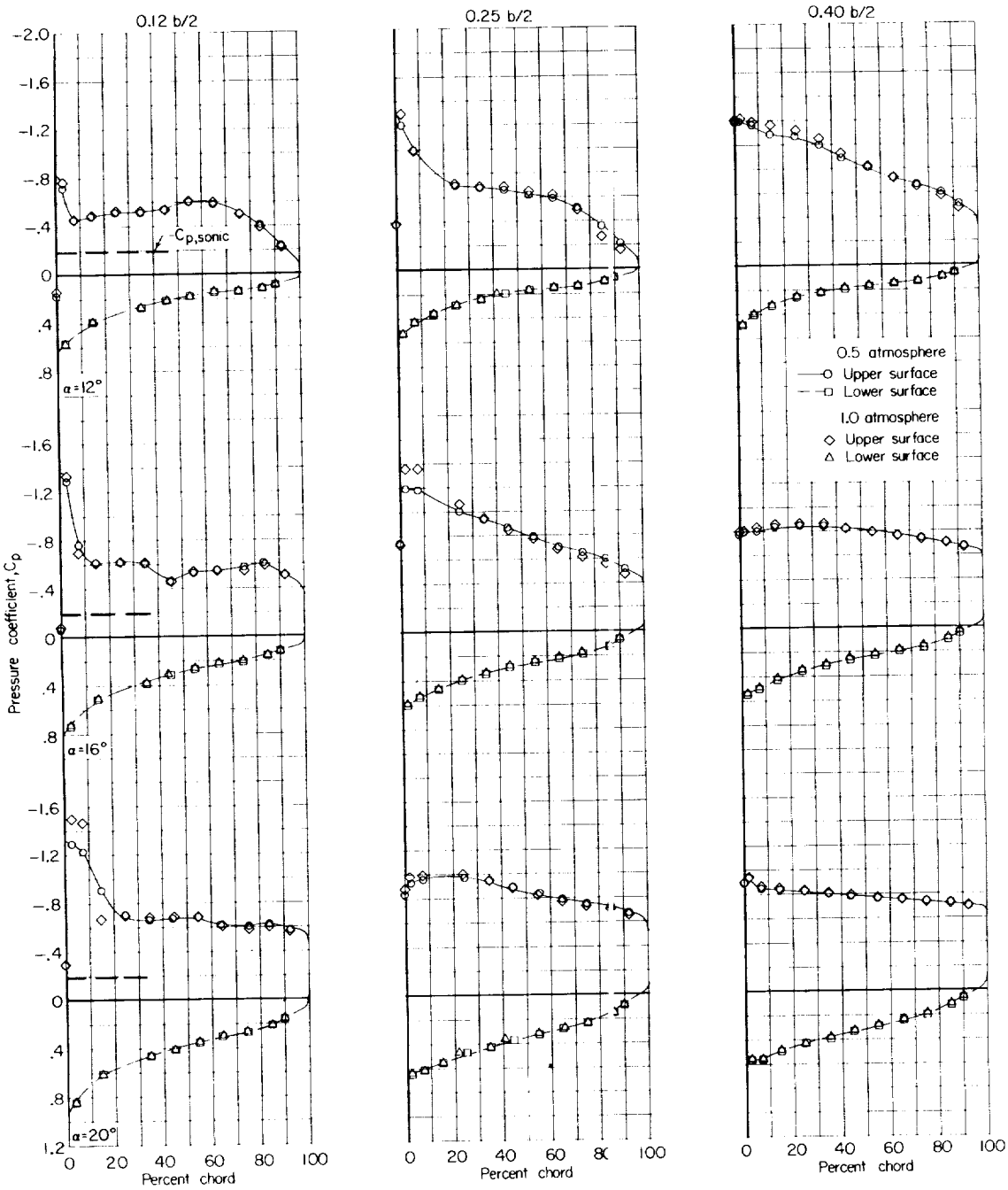
(e) M = 0.900; $\alpha = 2^{\circ}$, 4°, and 8°.

Figure 4.- Continued.



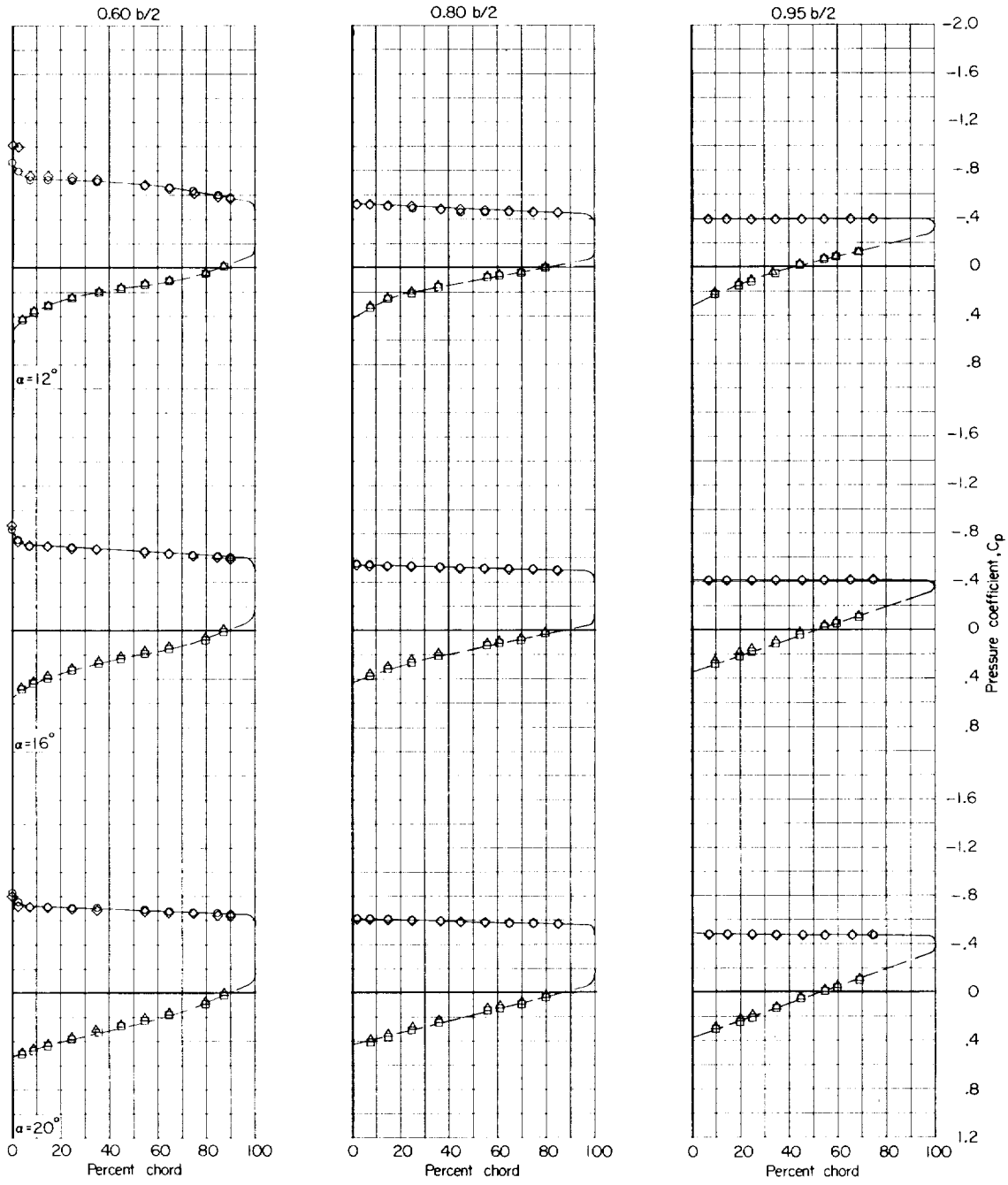
(e) Concluded.

Figure 4.- Continued.



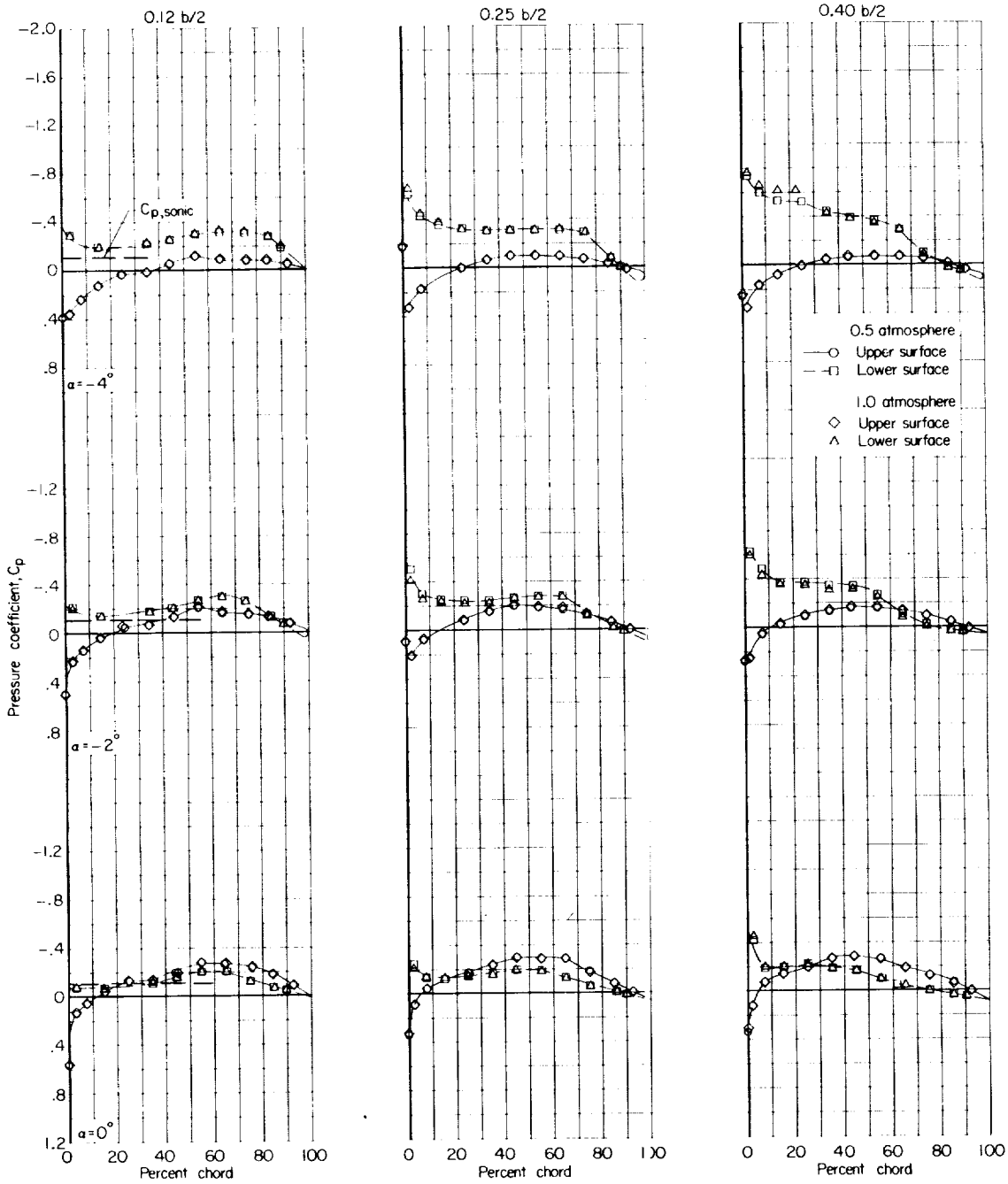
(f) M = 0.900; α = 12°, 16°, and 20°.

Figure 4.- Continued.



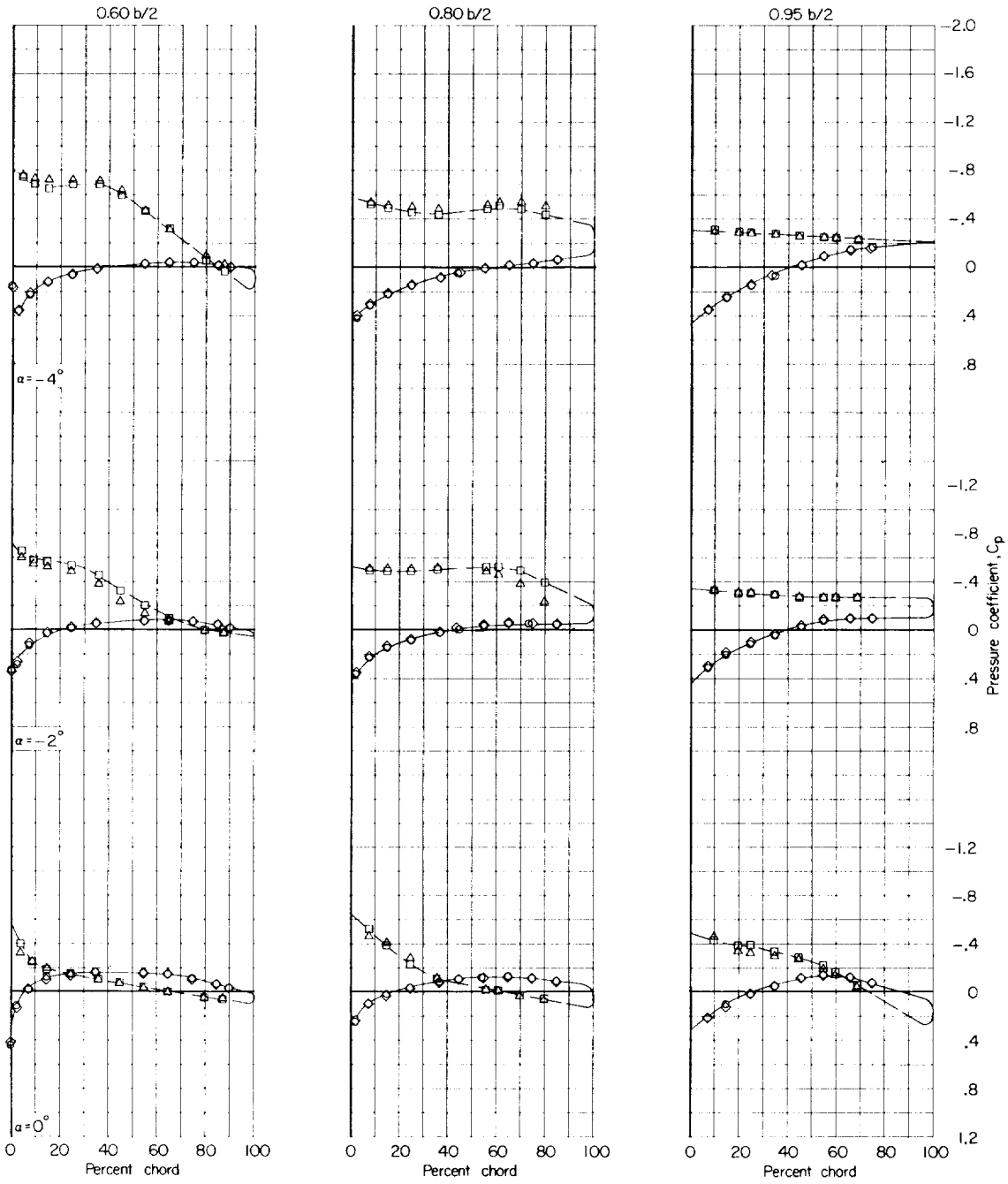
(f) Concluded.

Figure 4.- Continued.



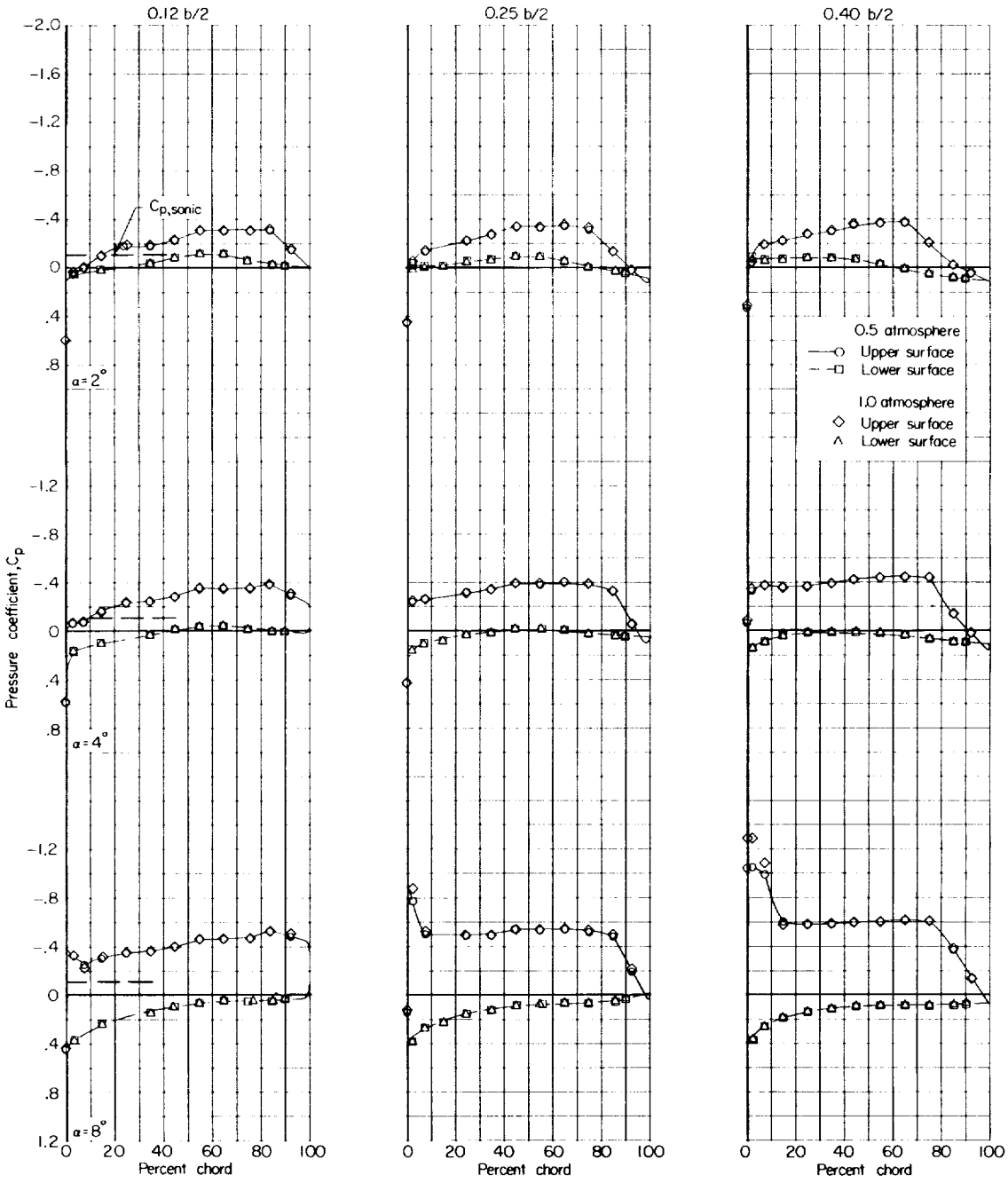
(g) M = 0.940; $\alpha = -4^{\circ}$, -2°, and 0°.

Figure 4.- Continued.



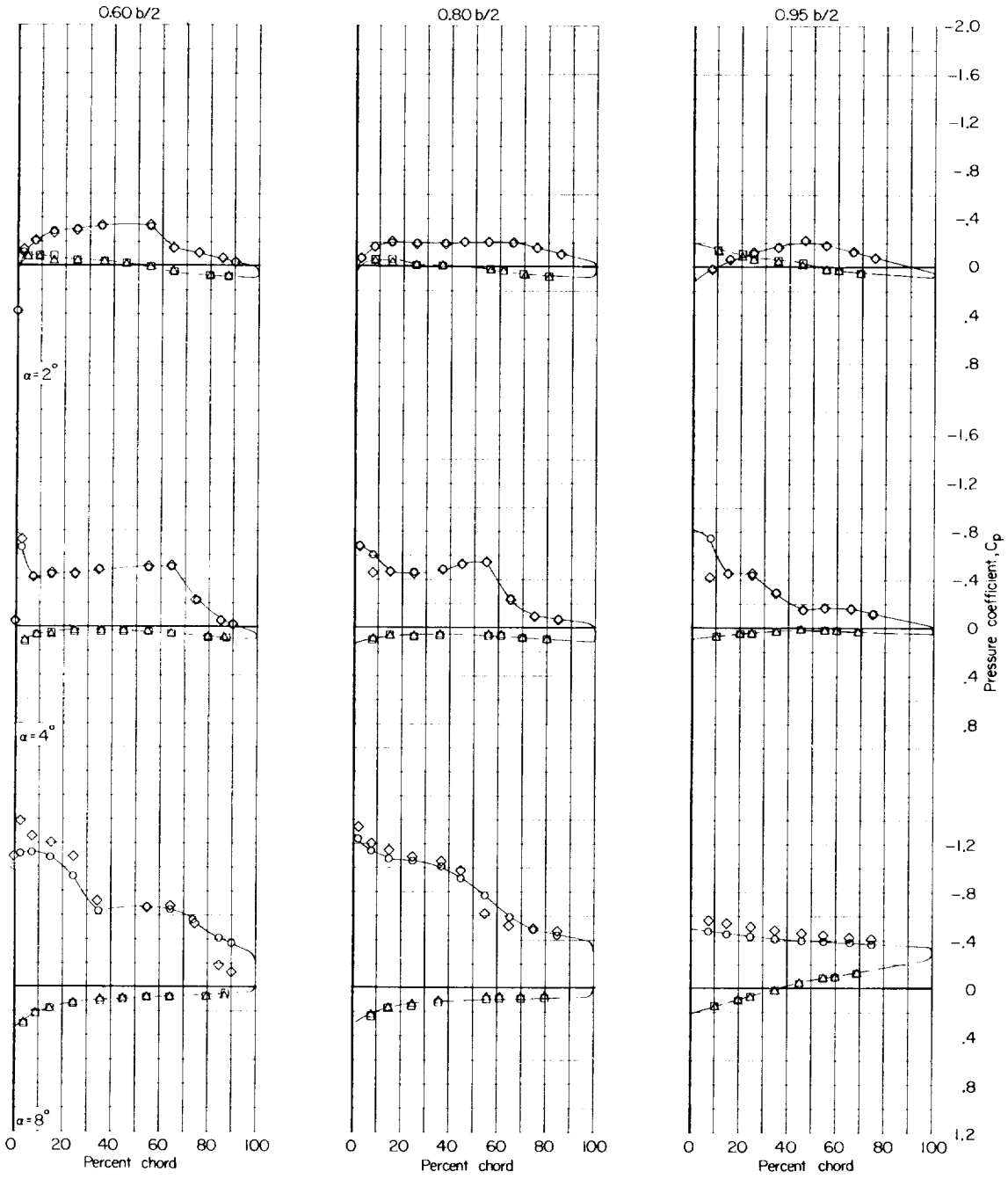
(g) Concluded.

Figure 4.- Continued.



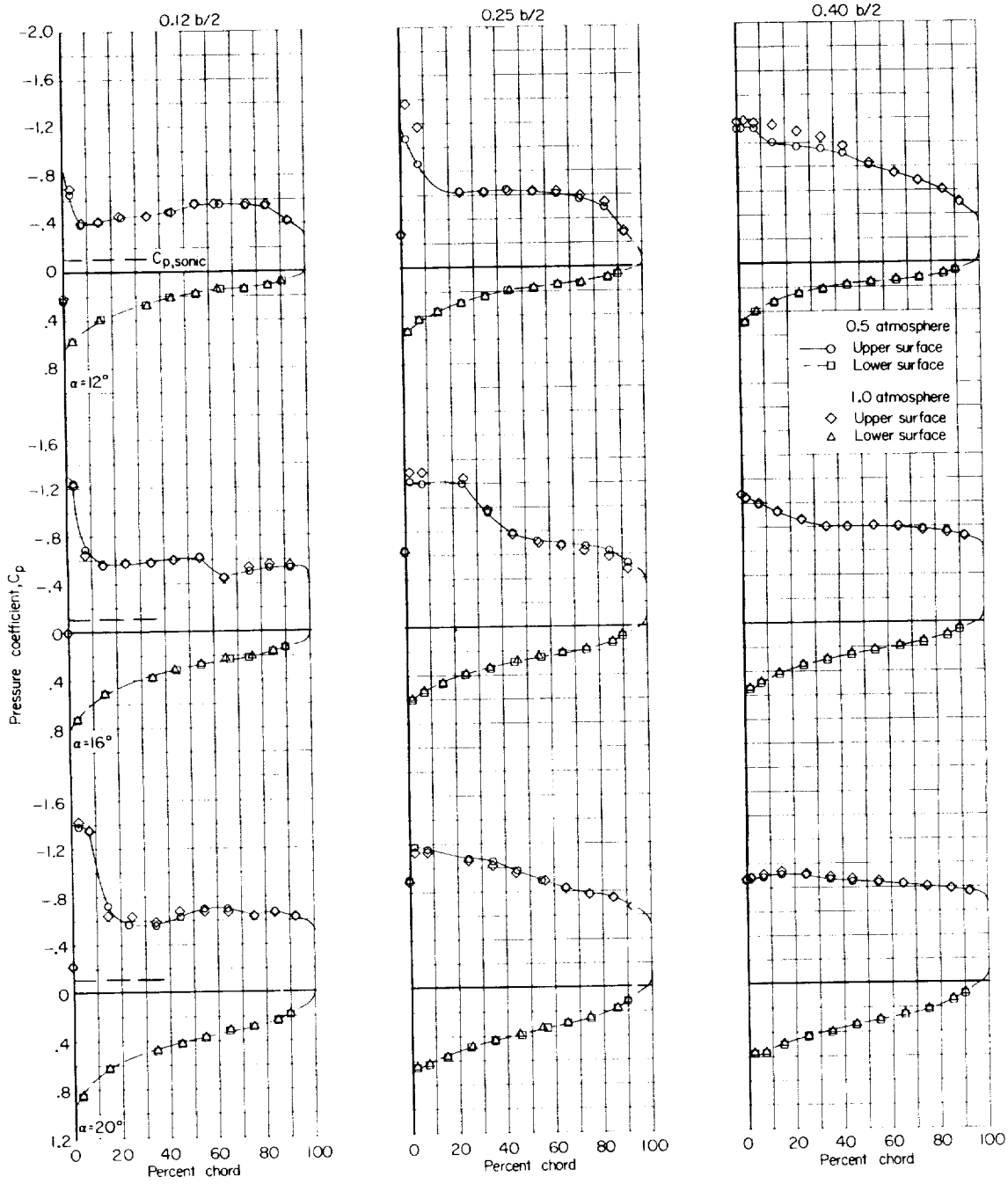
(h) M = 0.940; $\alpha = 2^{\circ}$, 4°, and 8°.

Figure 4.- Continued.



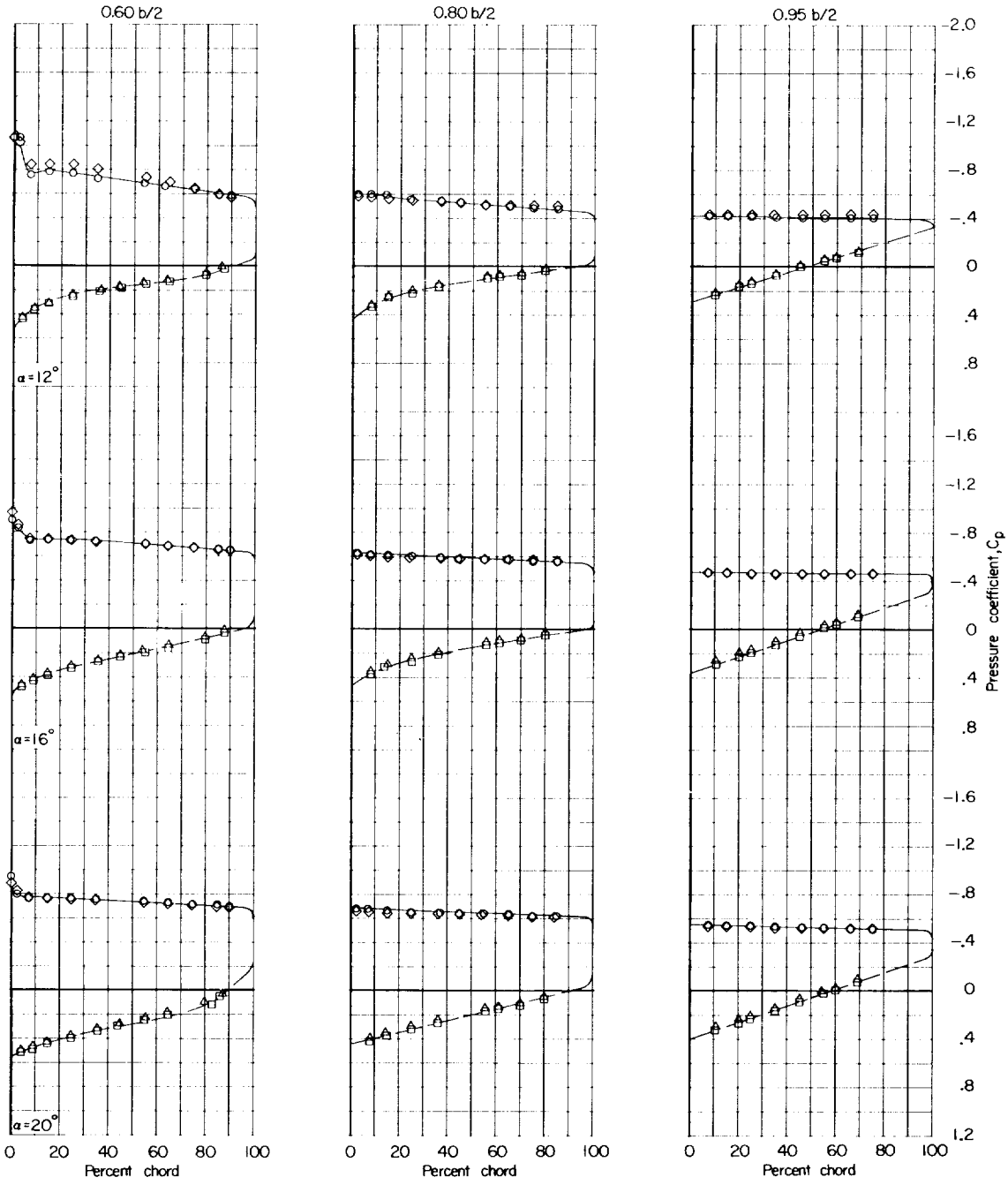
(h) Concluded.

Figure 4.- Continued.



(i) M = 0.940; $\alpha = 12^{o}$, 16^{c}, and 20^{o}.

Figure 4.- Continued.



(i) Concluded.

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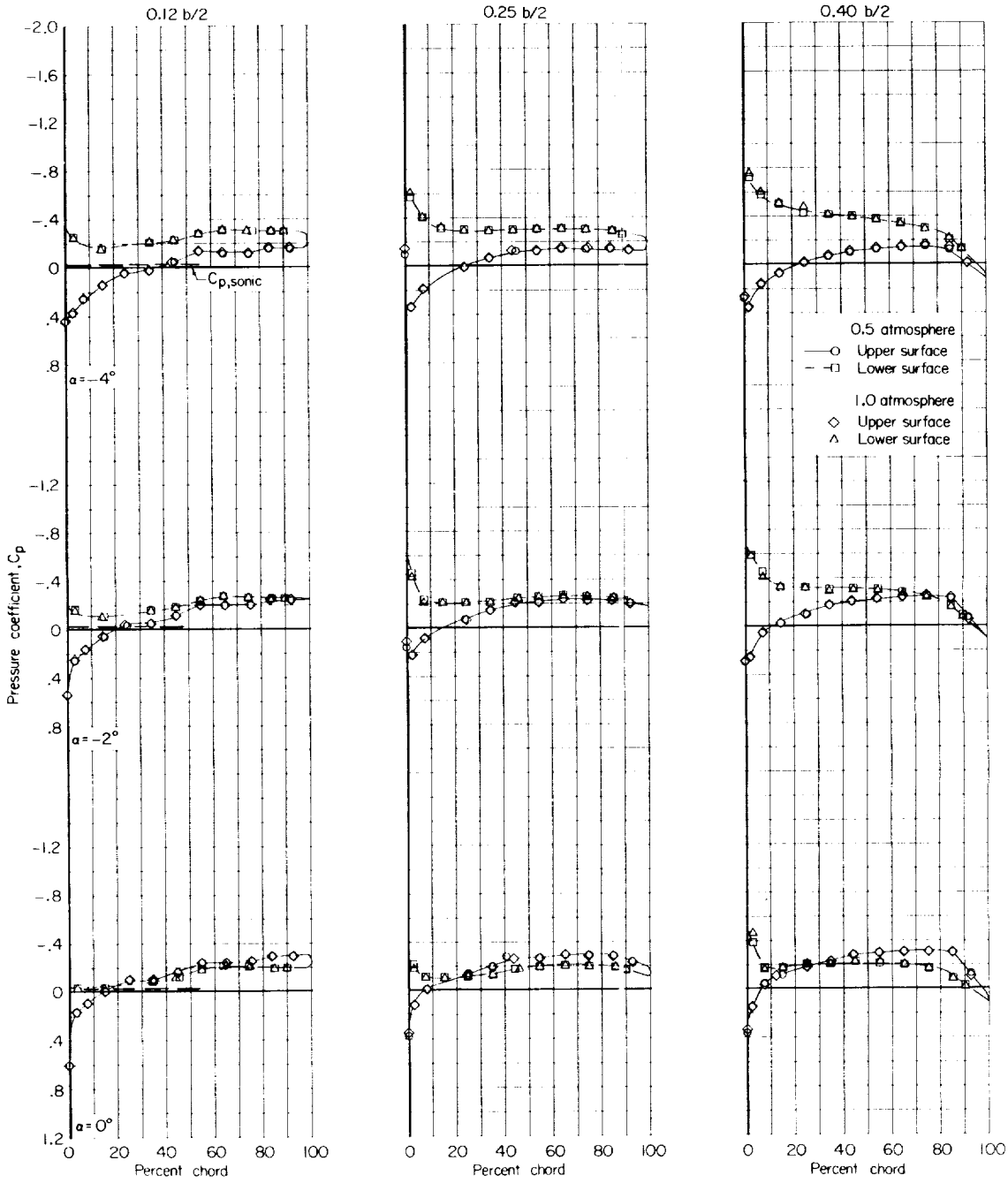
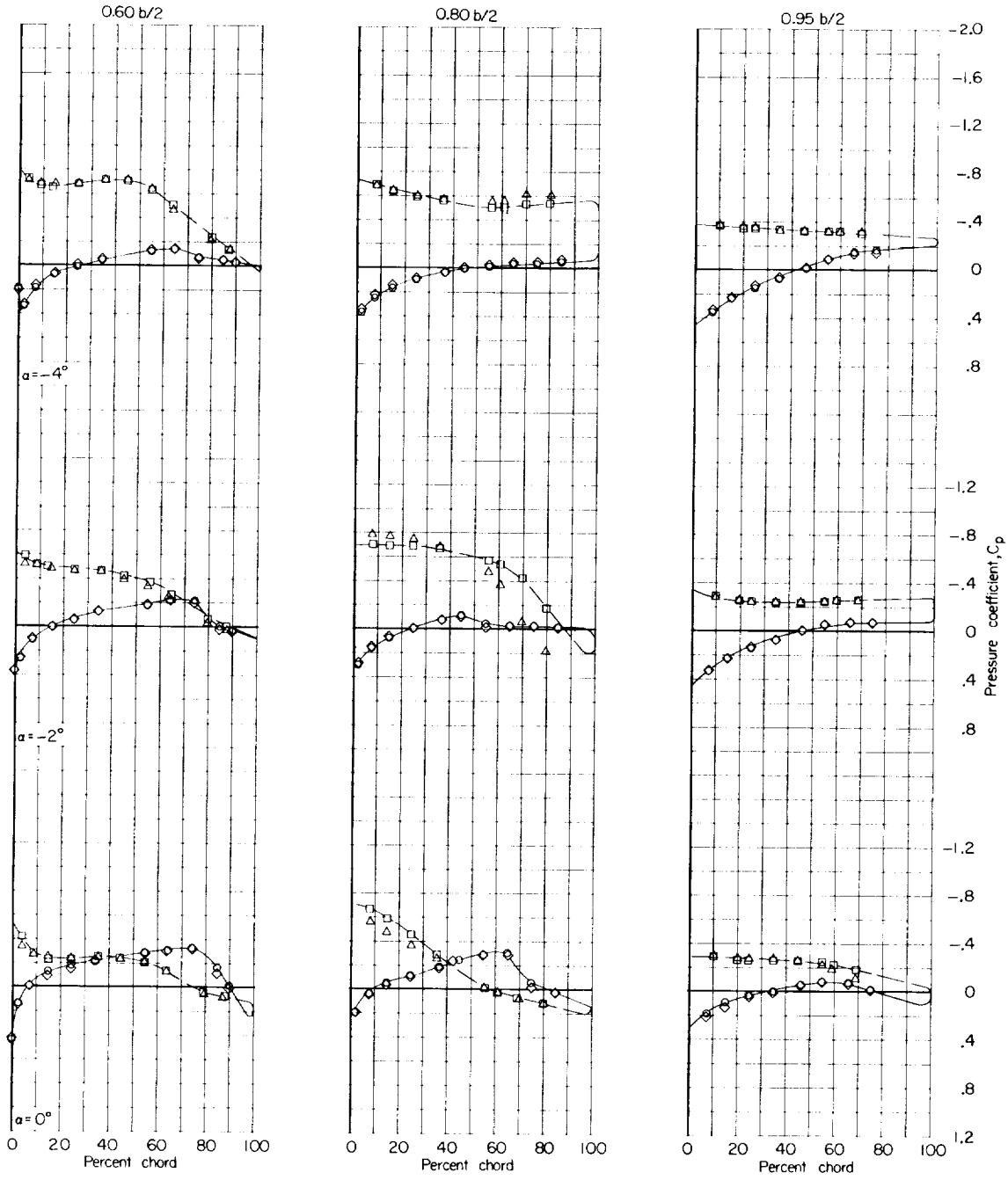


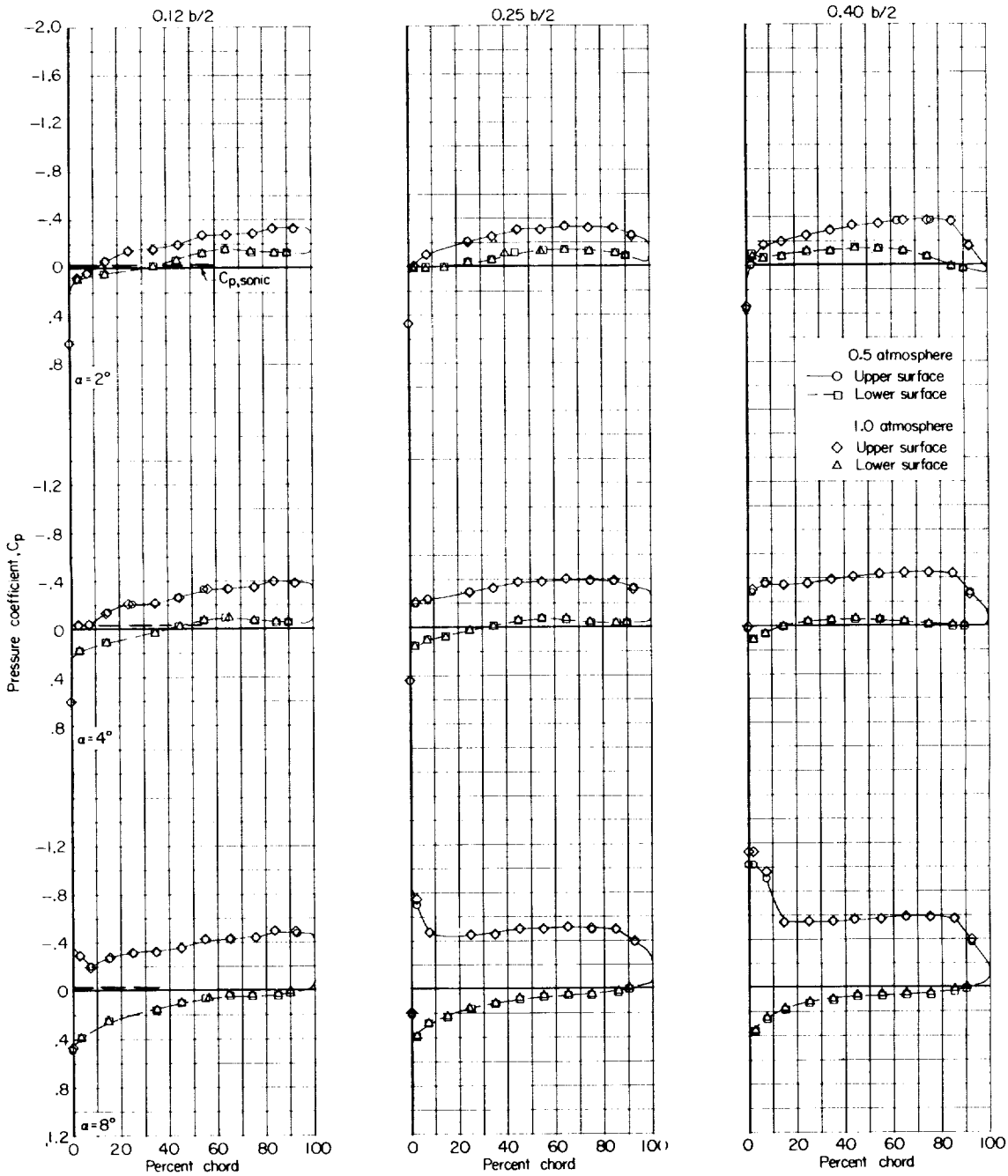
(j) M = 0.980; $\alpha = -4^{\circ}$, -2°, and 0°.

Figure 4.- Continued.



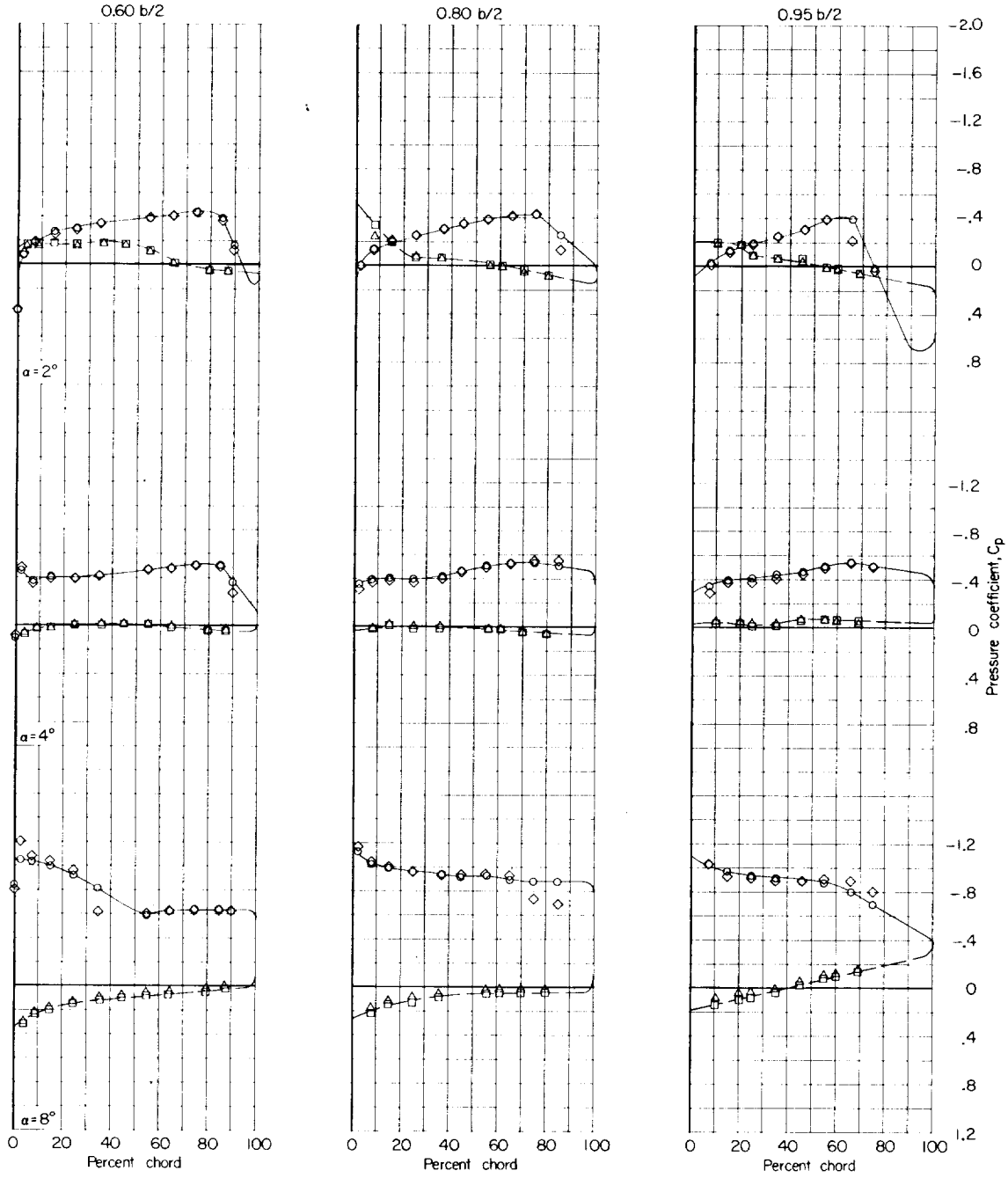
(j) Concluded.

Figure 4.- Continued.



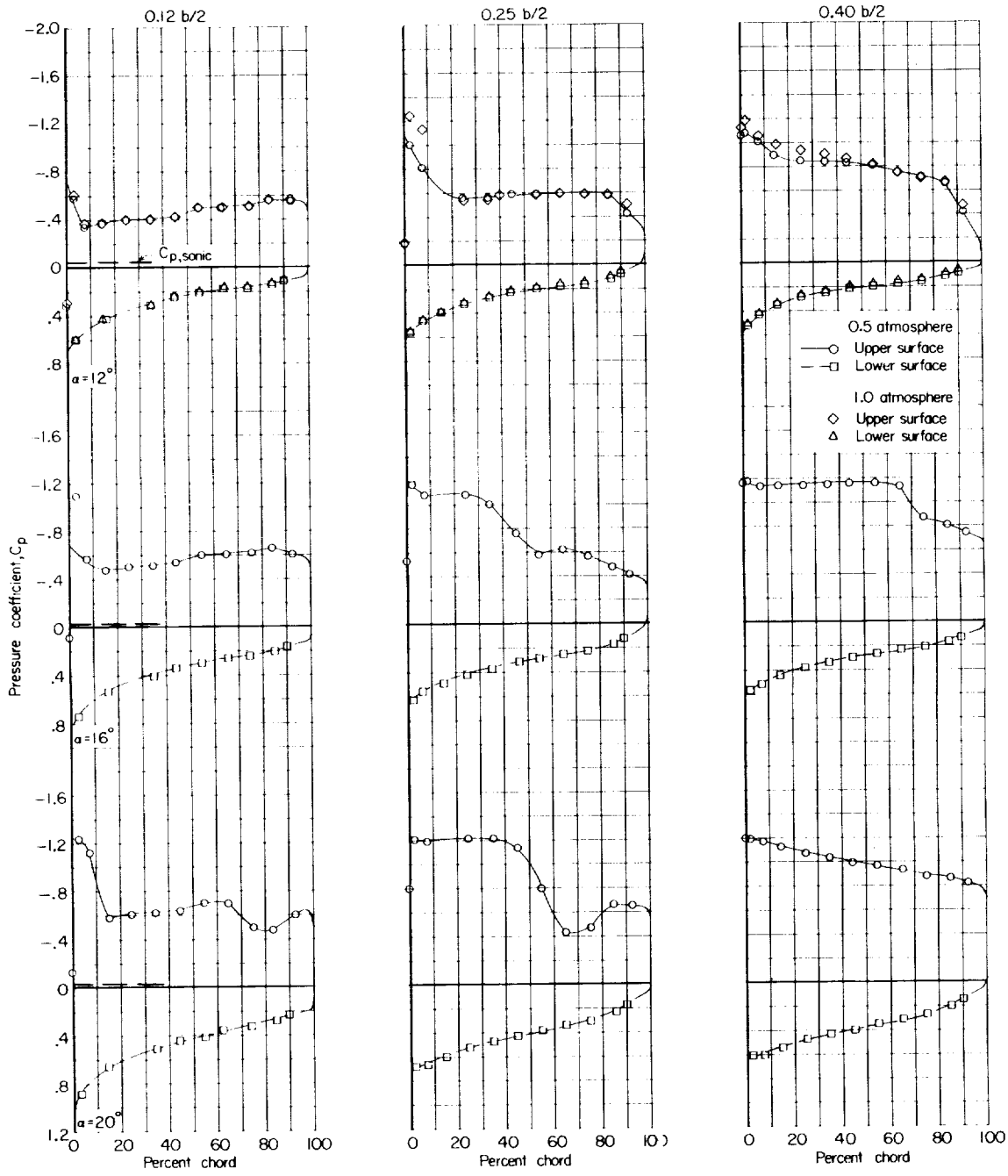
(k) M = 0.980; α = 2°, 4°, and 8°.

Figure 4.- Continued.



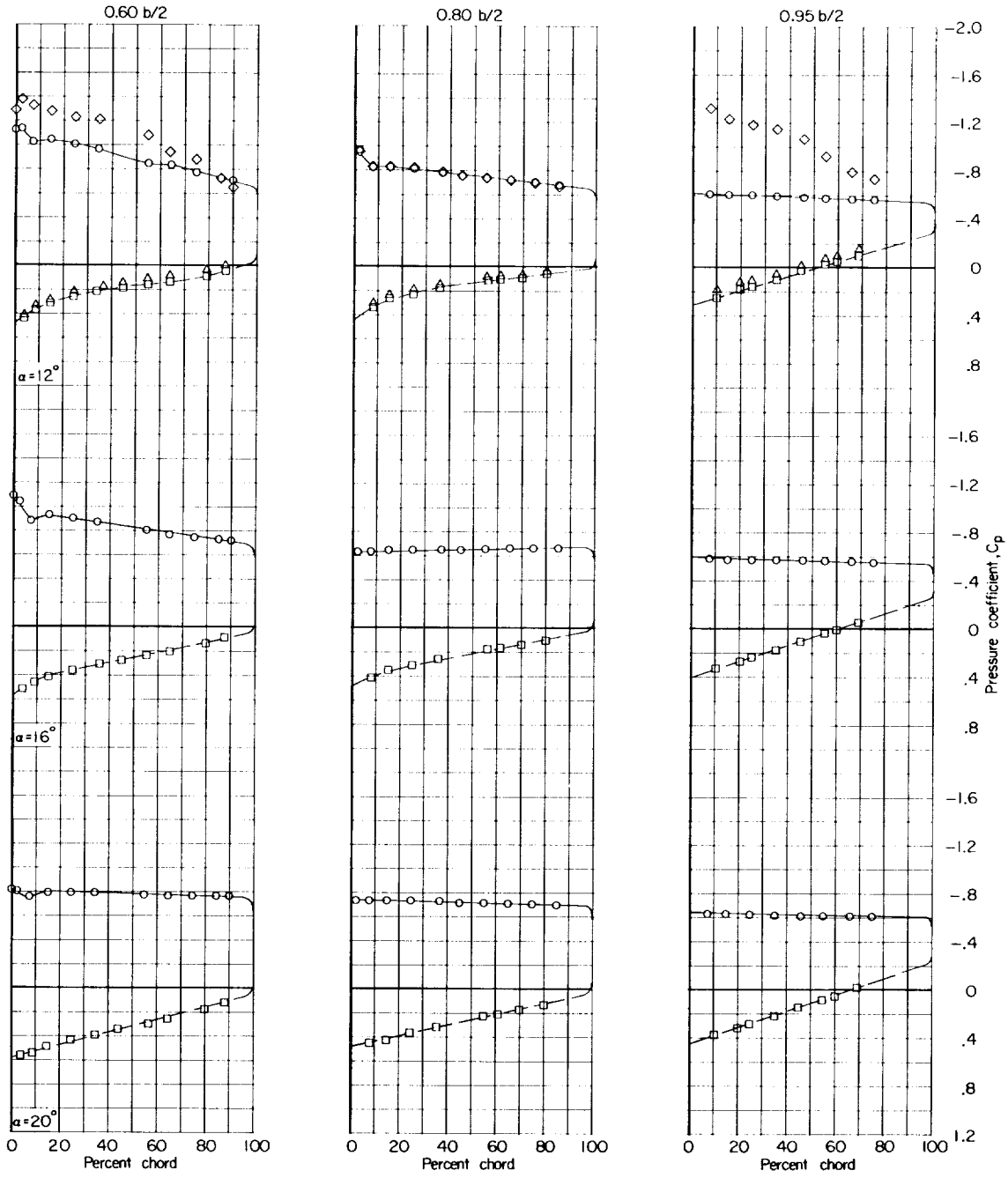
(k) Concluded.

Figure 4.- Continued.



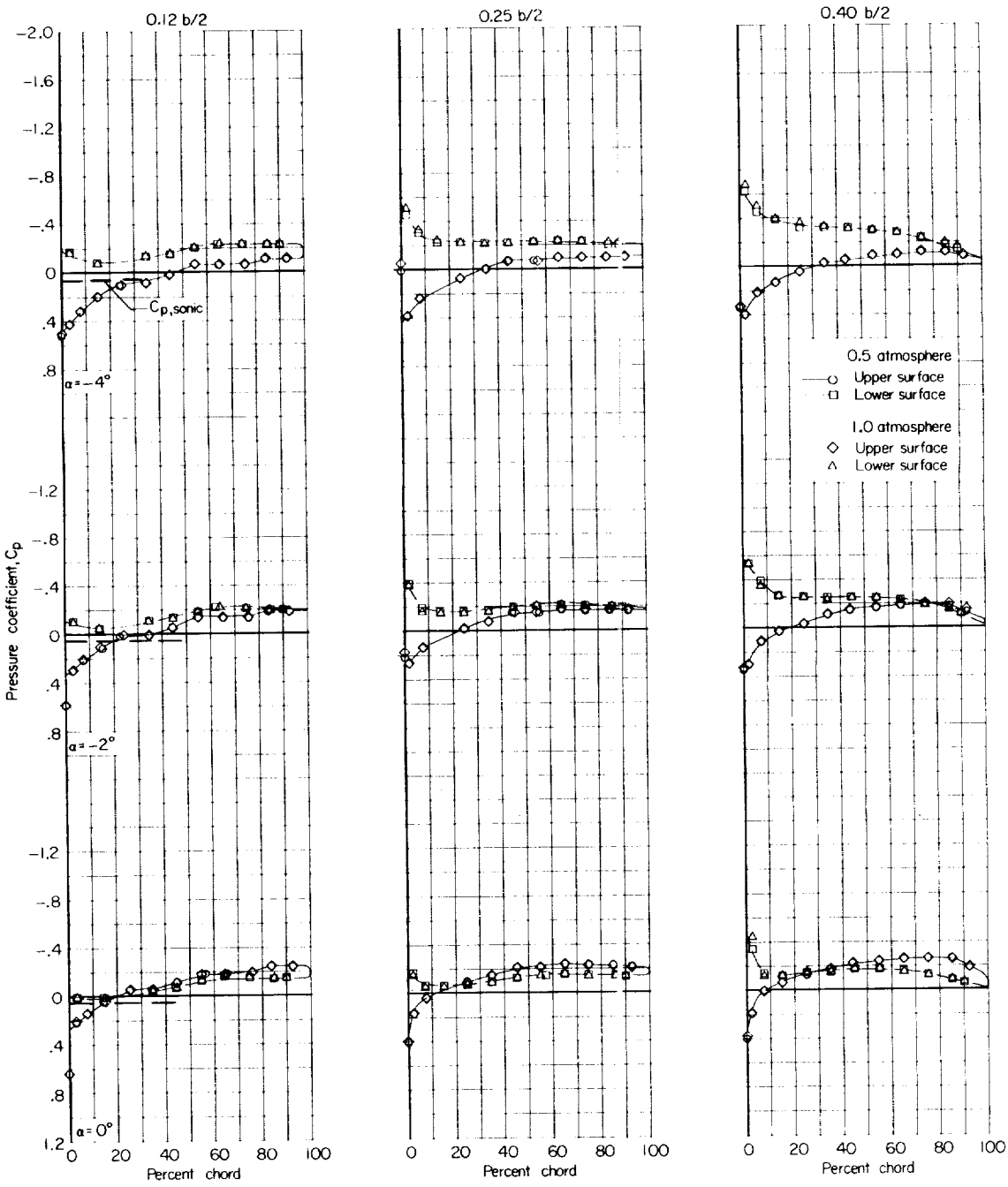
(l) M = 0.980; α = 12°, 16°, and 20°.

Figure 4.- Continued.



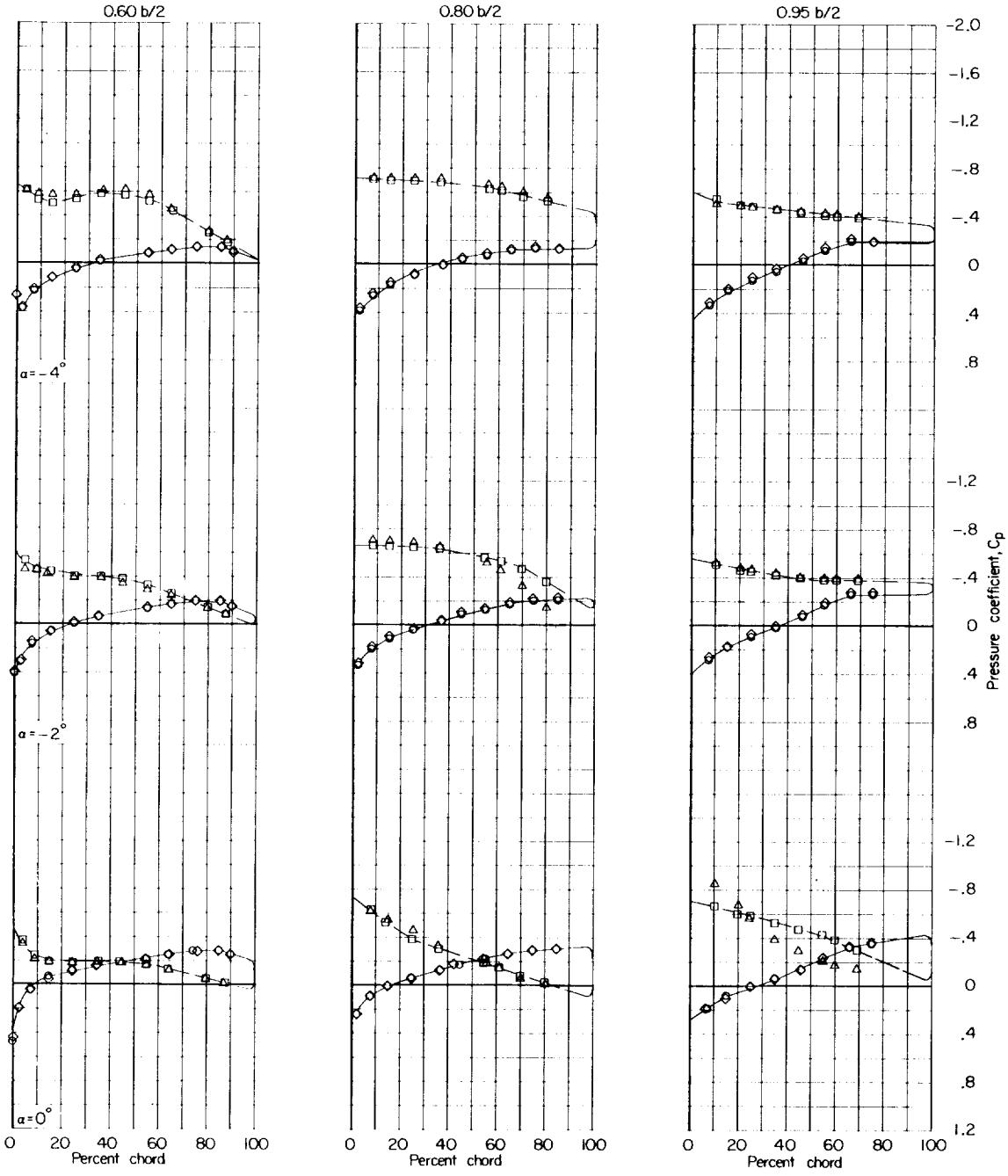
(l) Concluded.

Figure 4.- Continued.



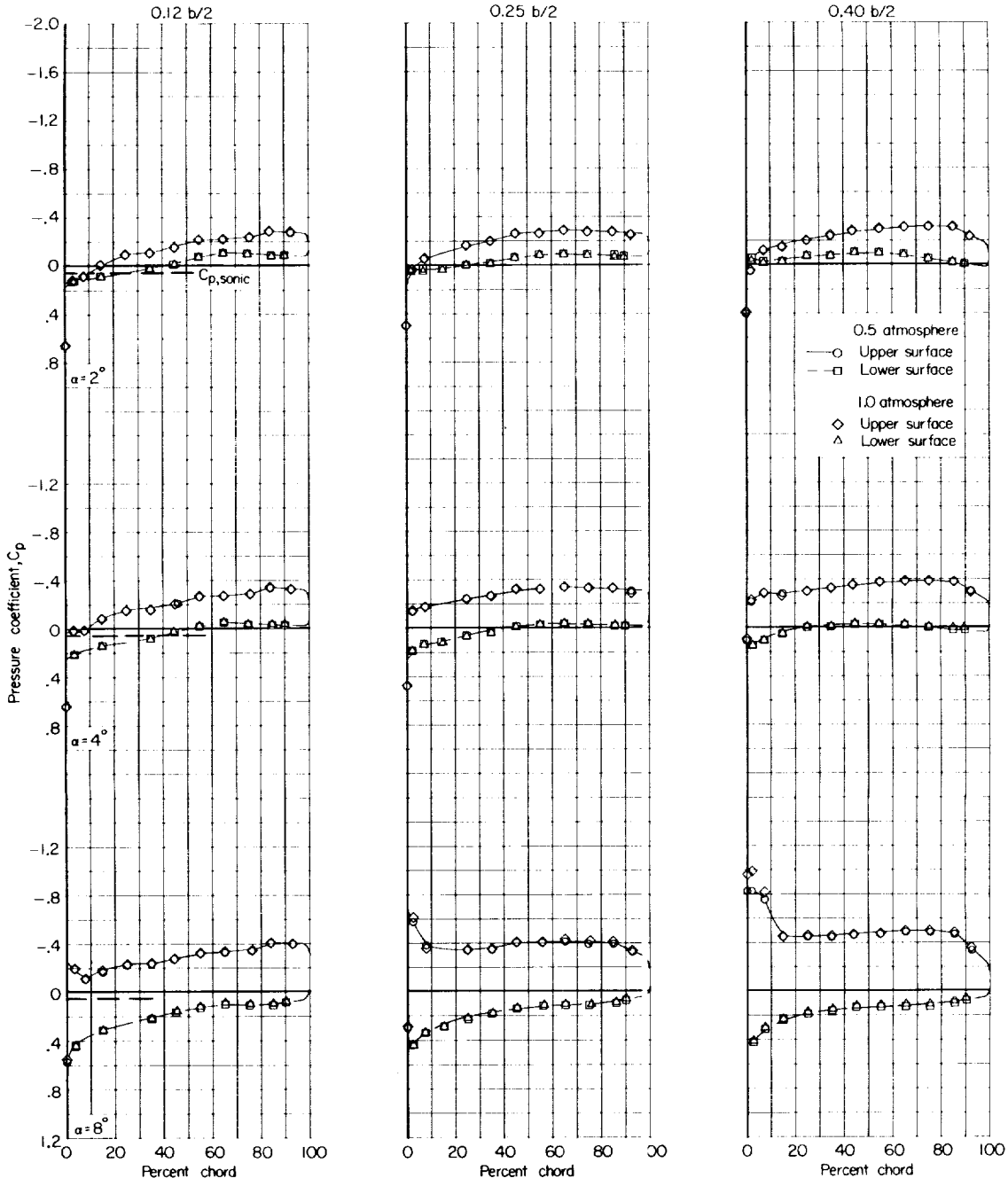
(m) M = 1.030; $\alpha = -4^{\circ}$, -2°, and 0°.

Figure 4.- Continued.



(m) Concluded.

Figure 4.- Continued.



(n) M = 1.030; $\alpha = 2^\circ$, 4°, and 8°.

Figure 4.- Continued.

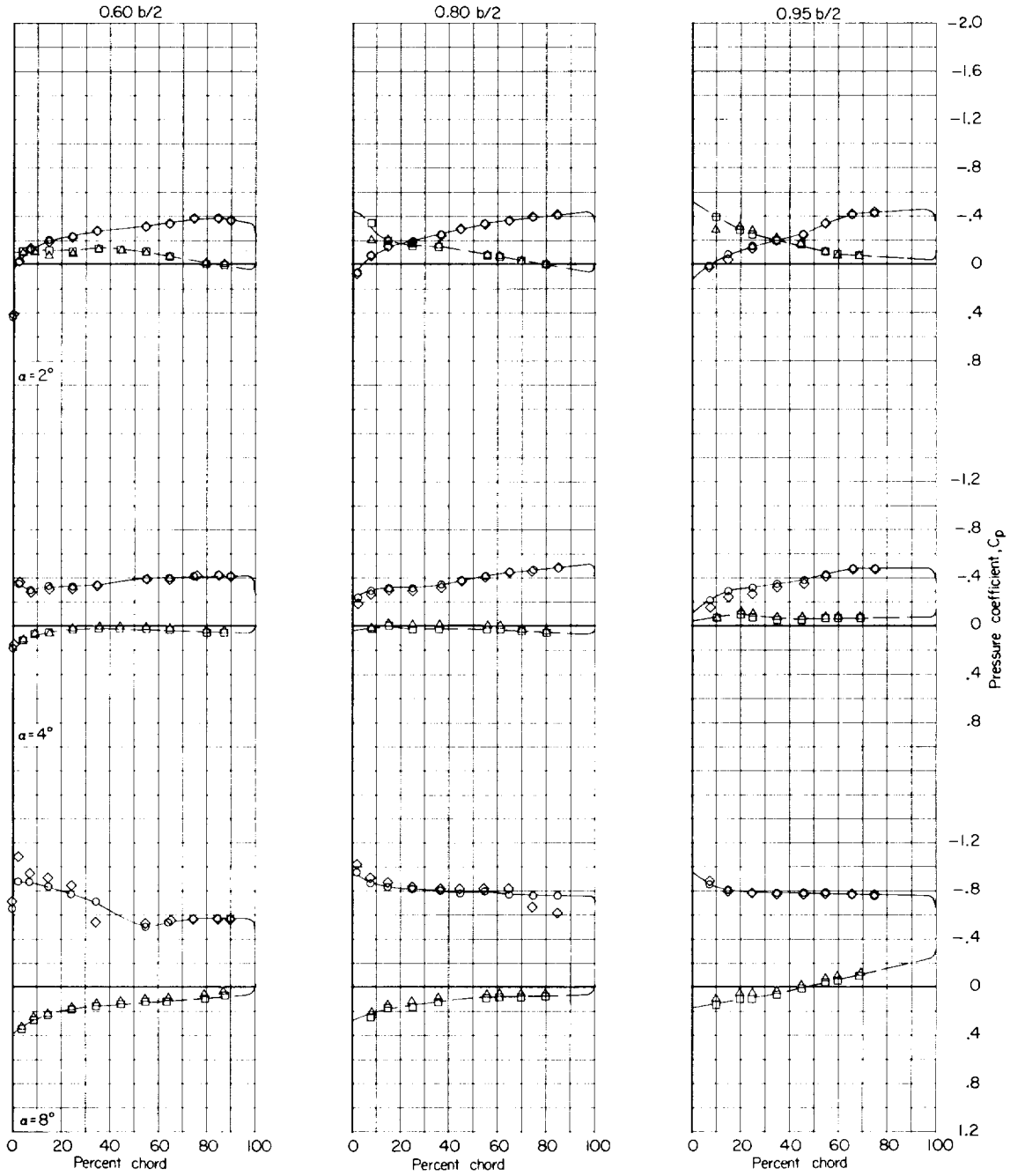


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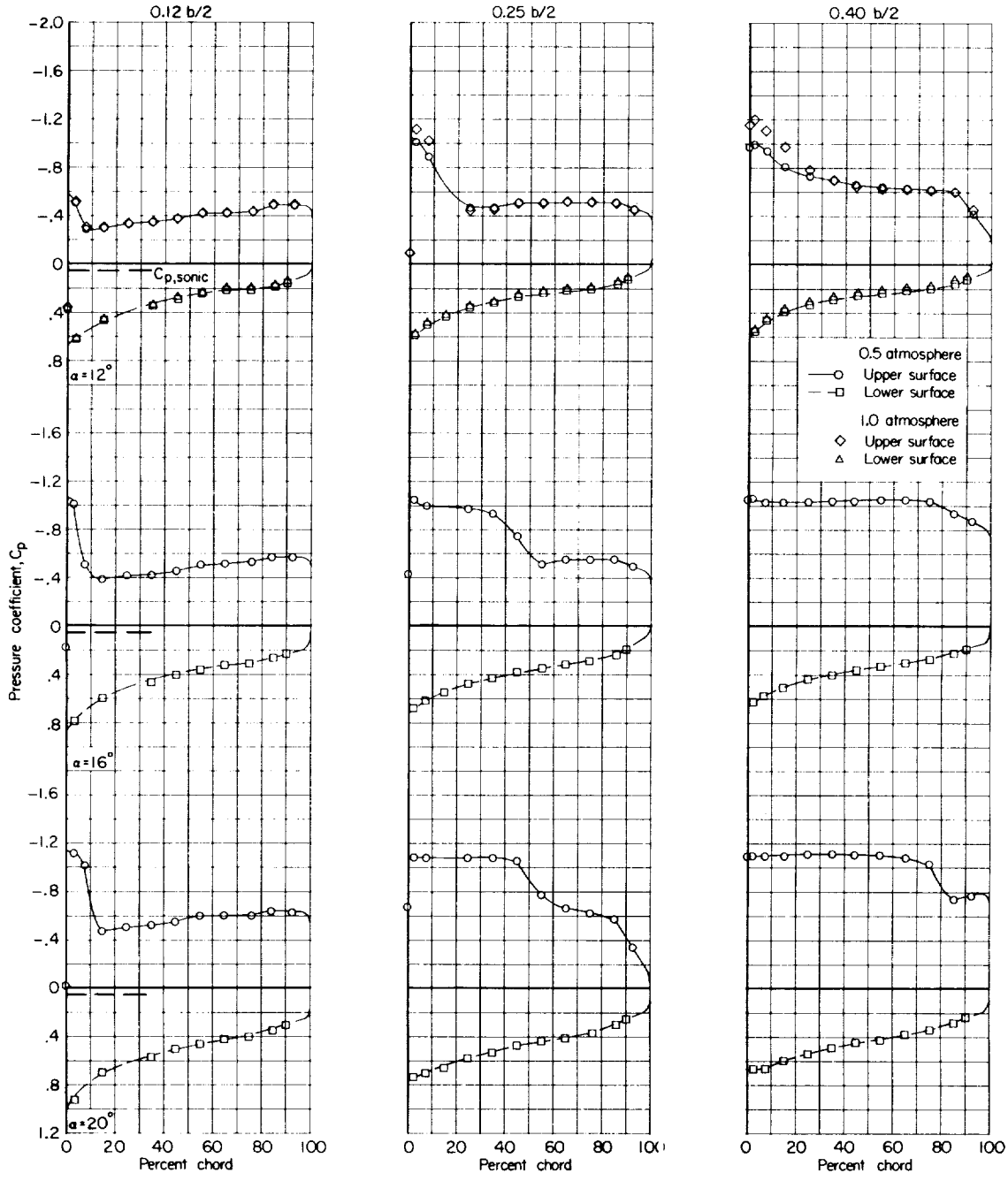
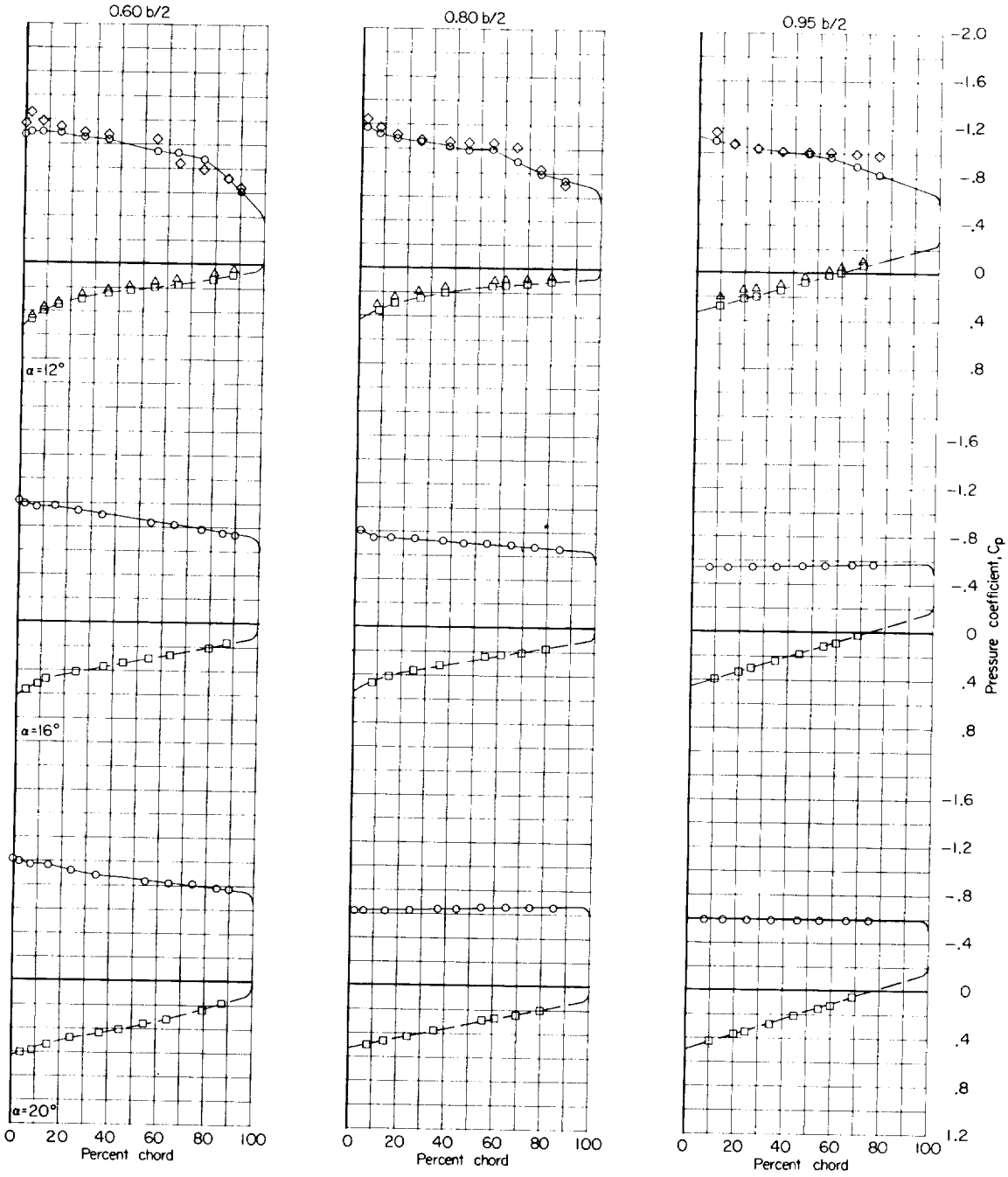
Figure 4.- Continued.



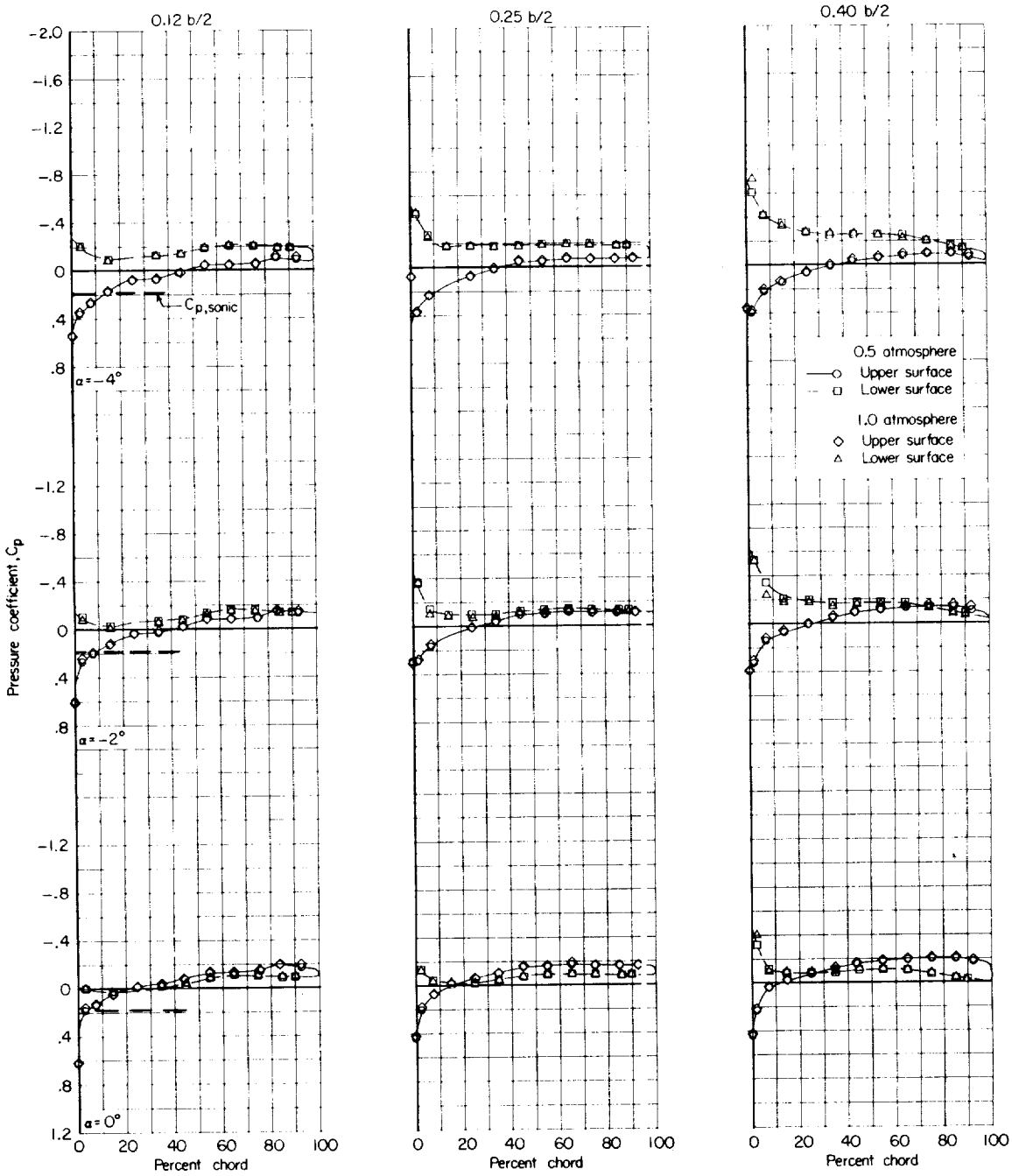
(o) M = 1.030; α = 12°, 16°, and 20°.

Figure 4.- Continued.



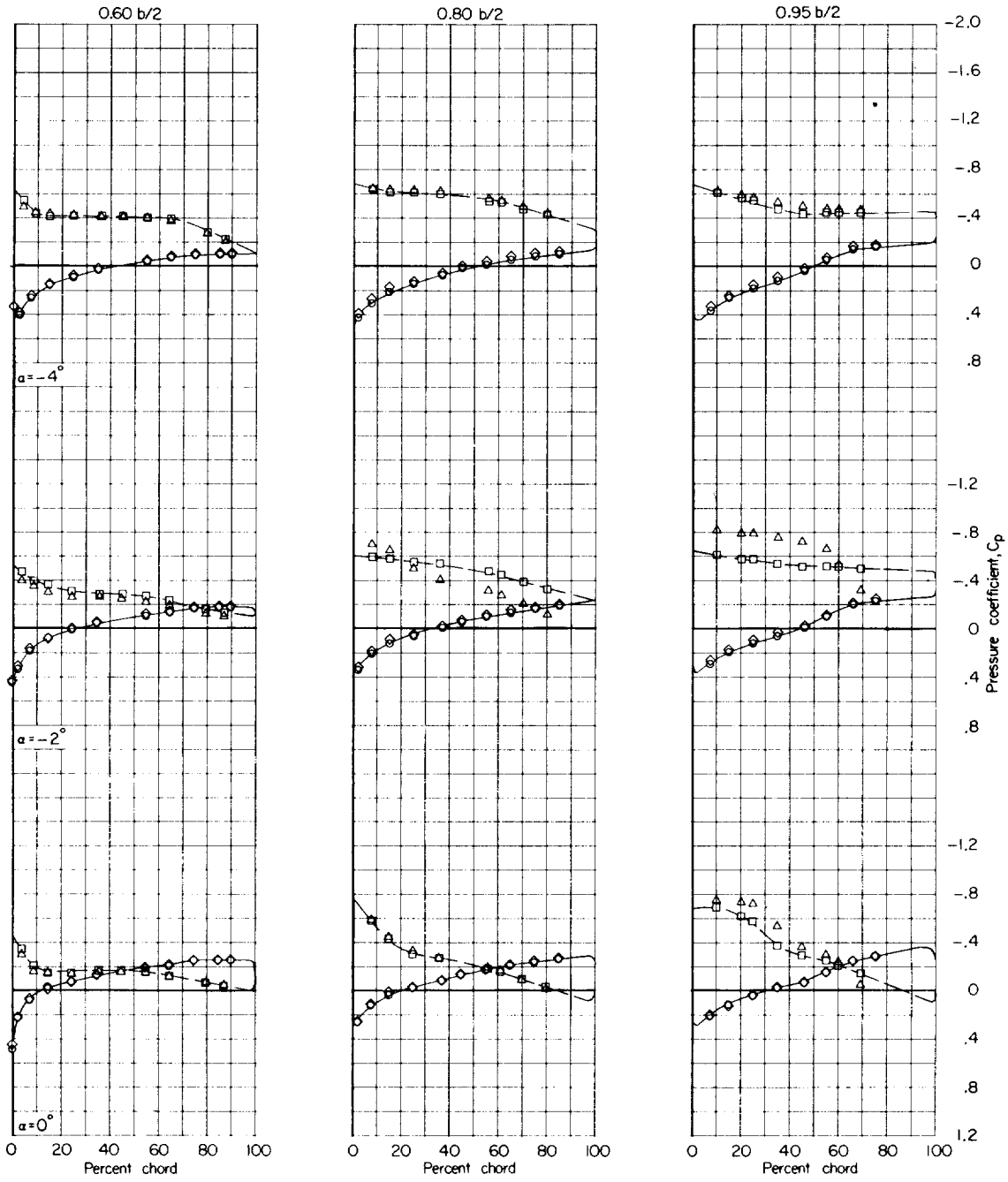
(o) Concluded.

Figure 4.- Continued.



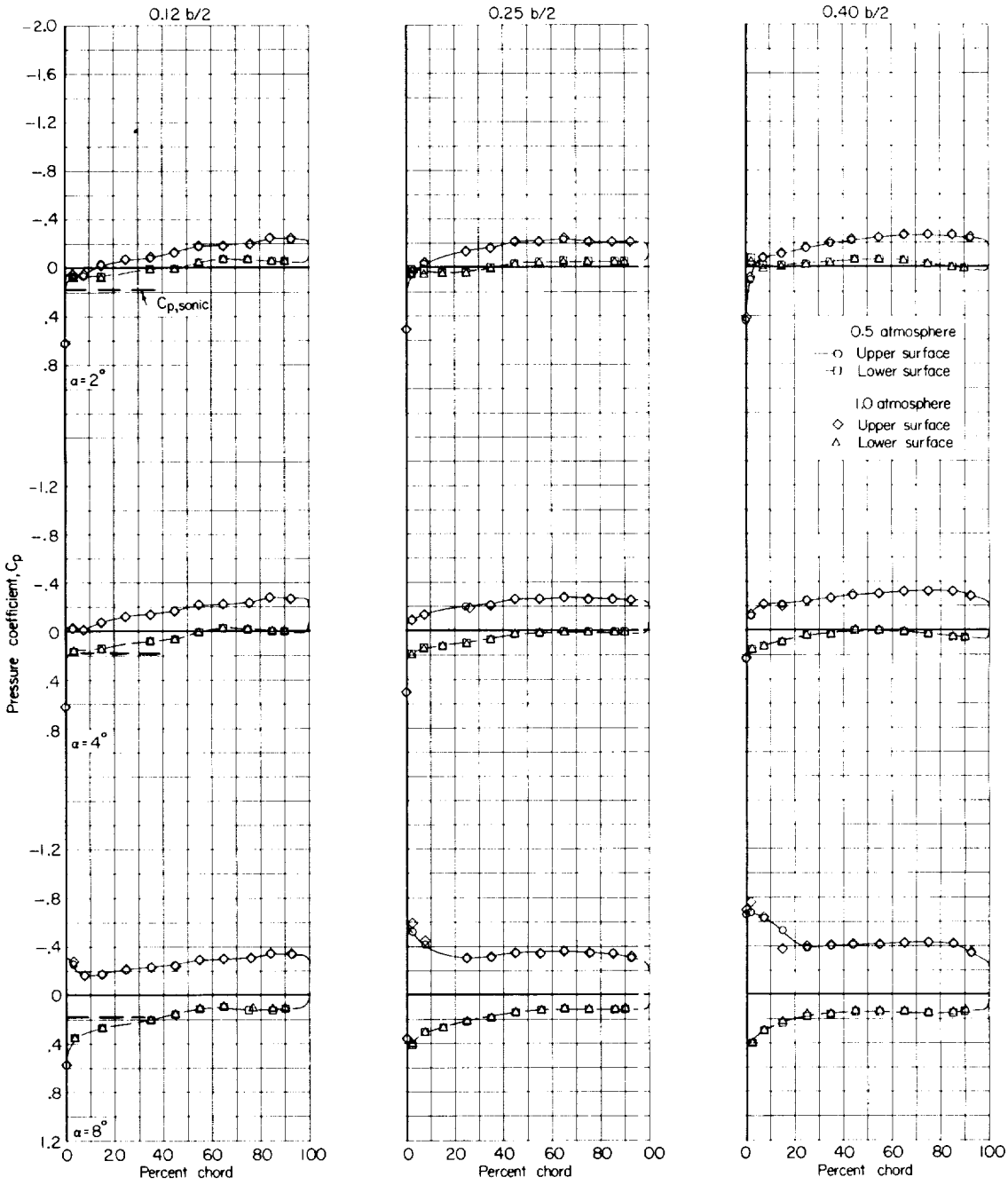
(p) M = 1.125; α = -4°, -2°, and 0°.

Figure 4.- Continued.



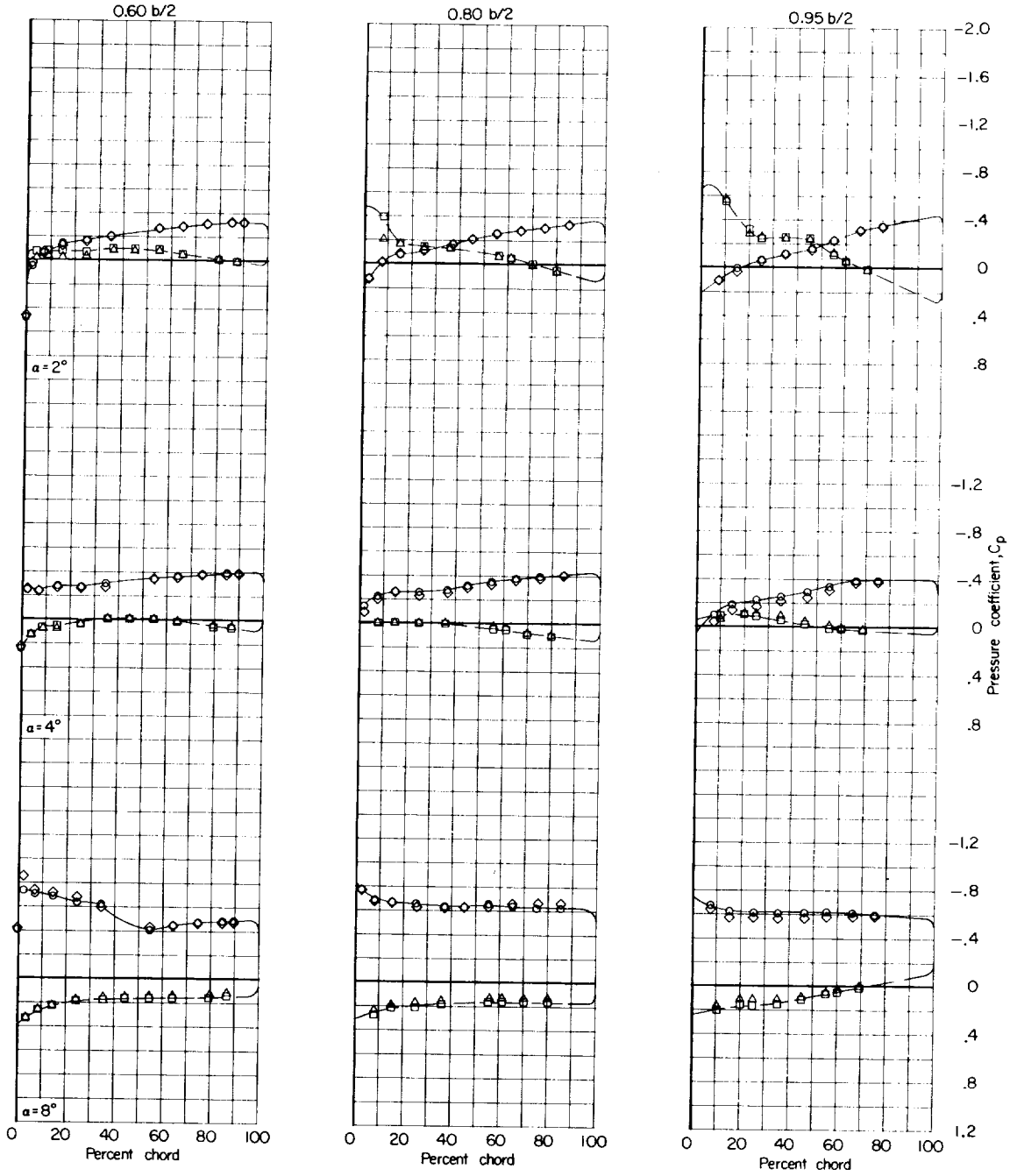
(p) Concluded.

Figure 4.- Continued.



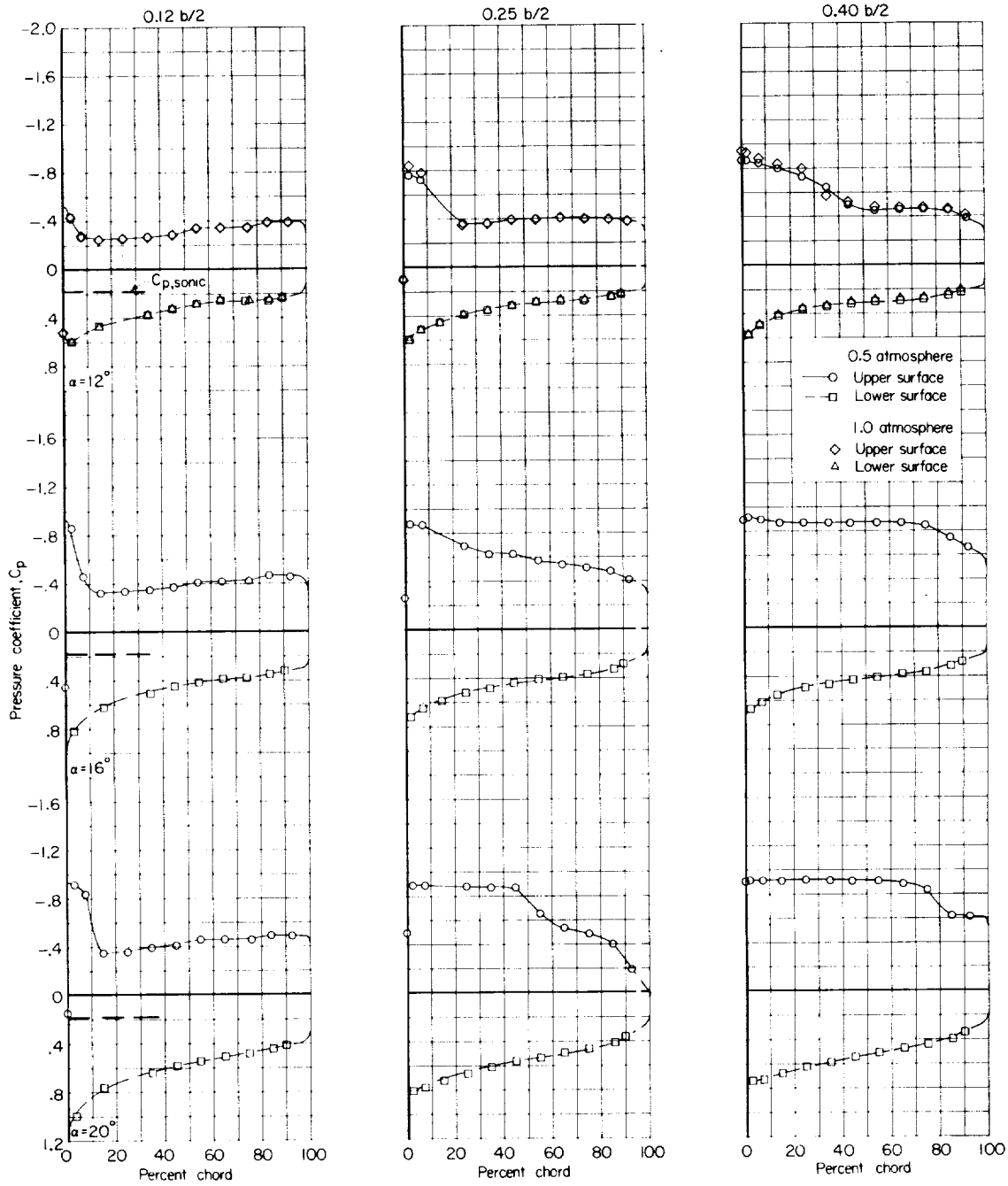
(q) M = 1.125; α = 2°, 4°, and 8°.

Figure 4.- Continued.



(q) Concluded.

Figure 4.- Continued.



(r) M = 1.125; α = 12°, 16°, and 20°.

Figure 4.- Continued.

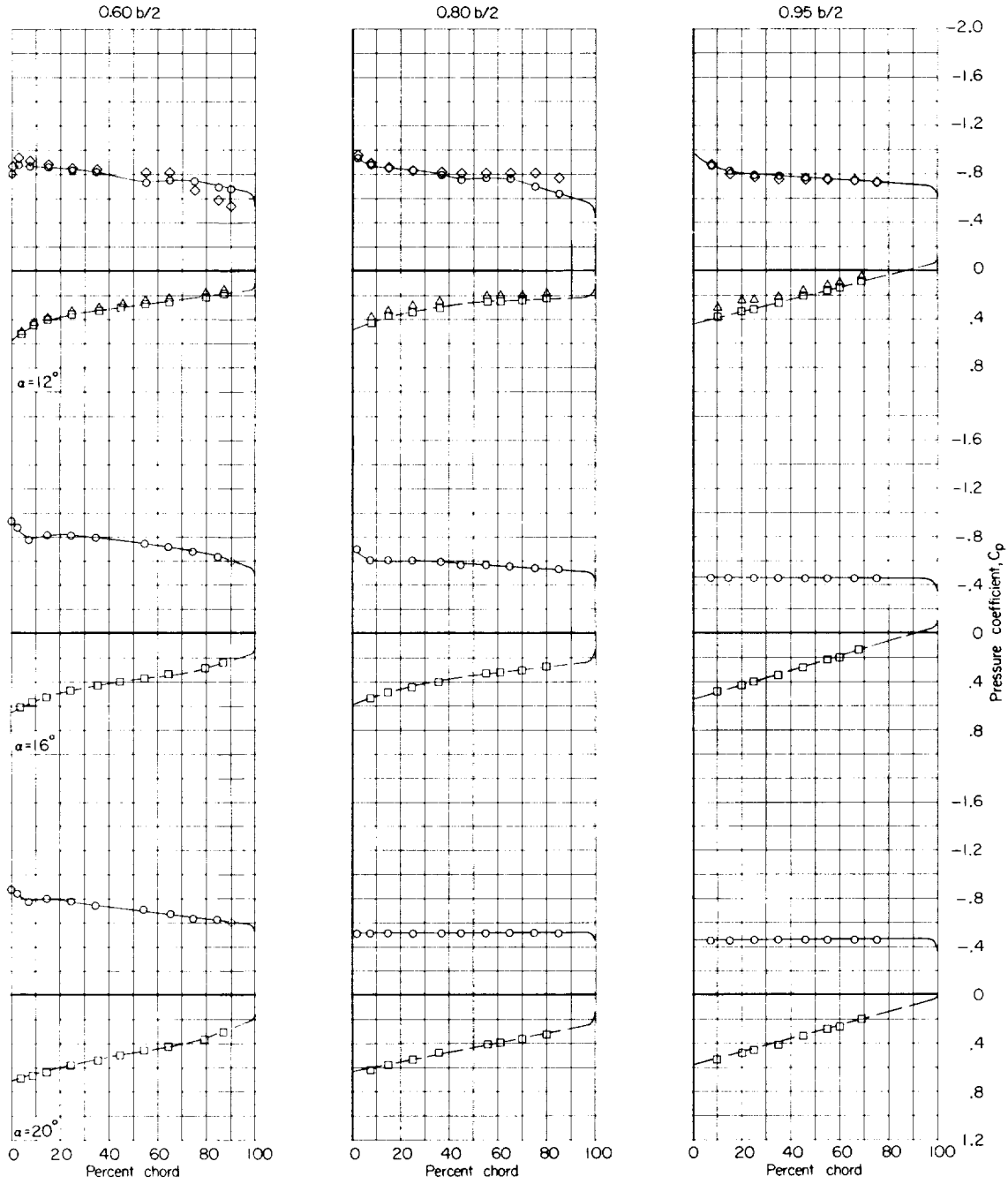


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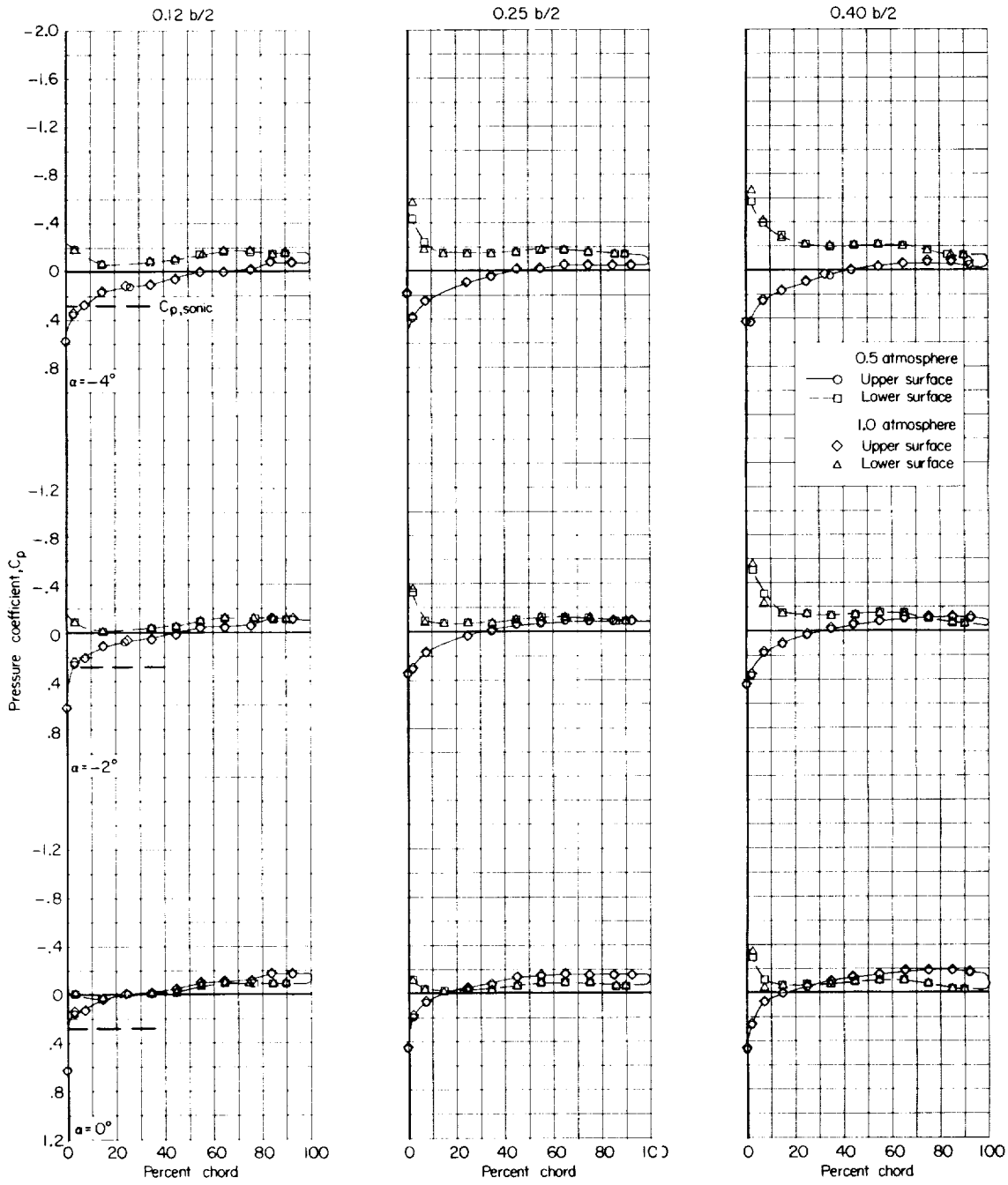
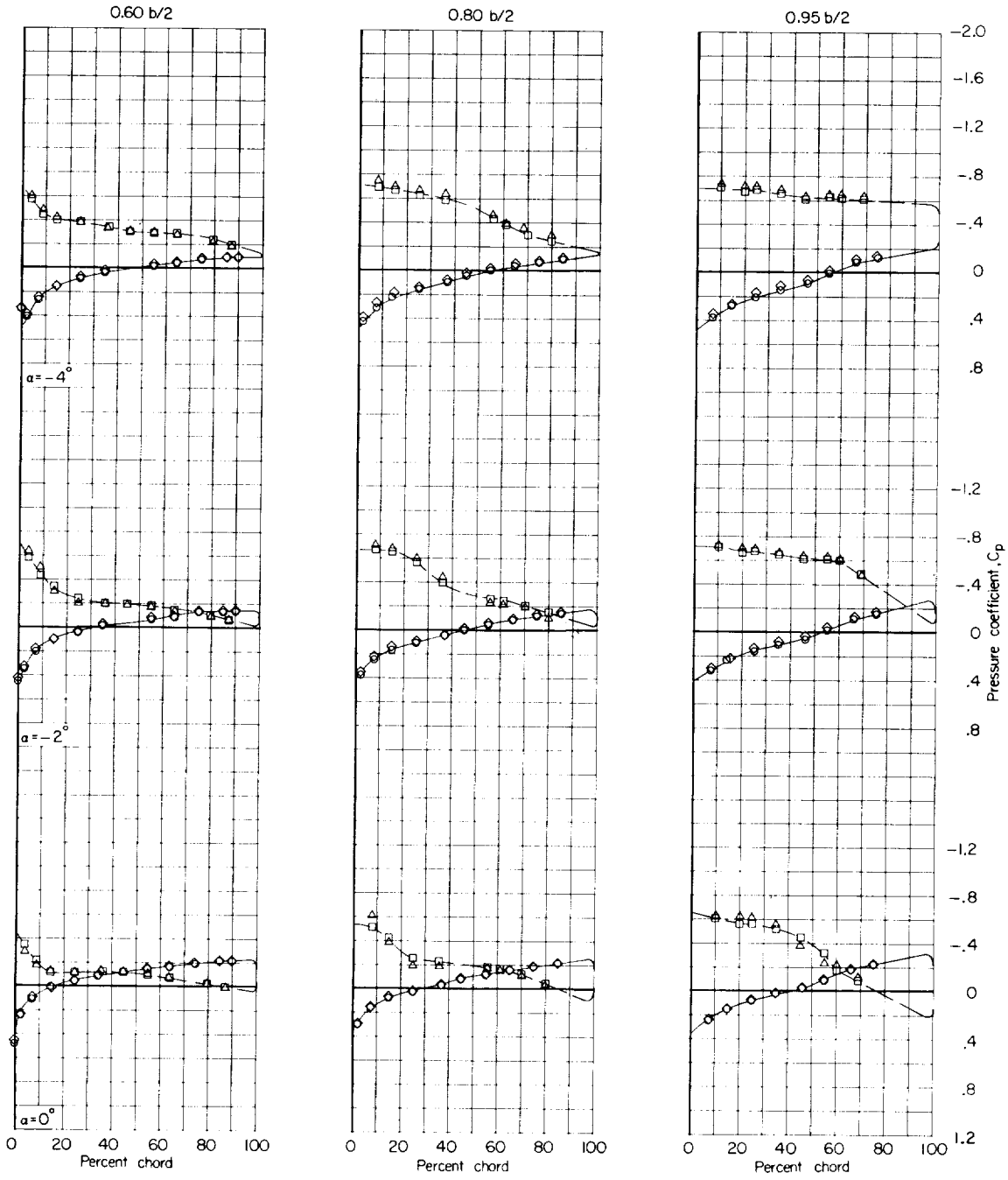
Figure 4.- Continued.



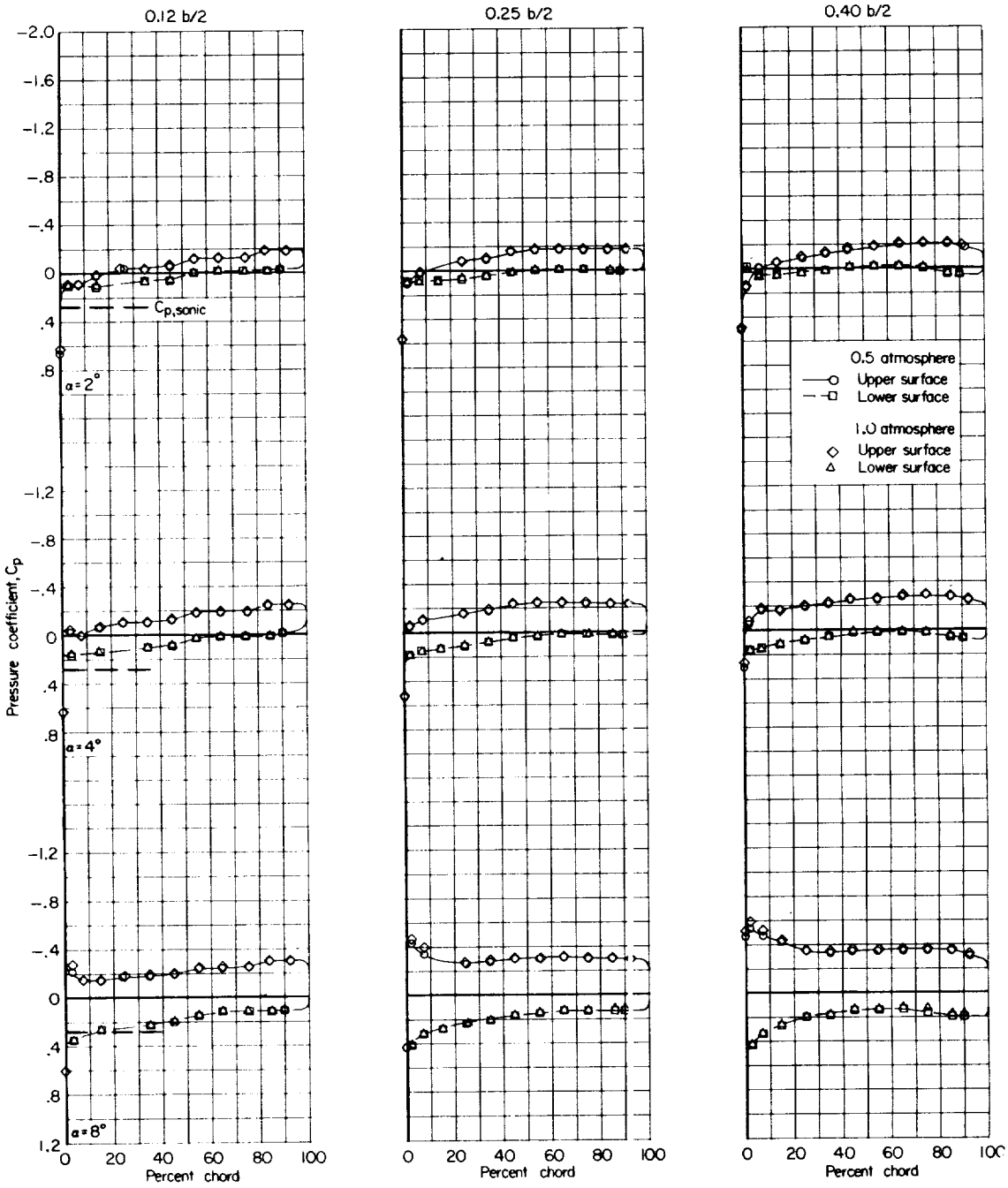
(s) M = 1.200; $\alpha = -4^{\circ}$, -2°, and 0°.

Figure 4.- Continued.



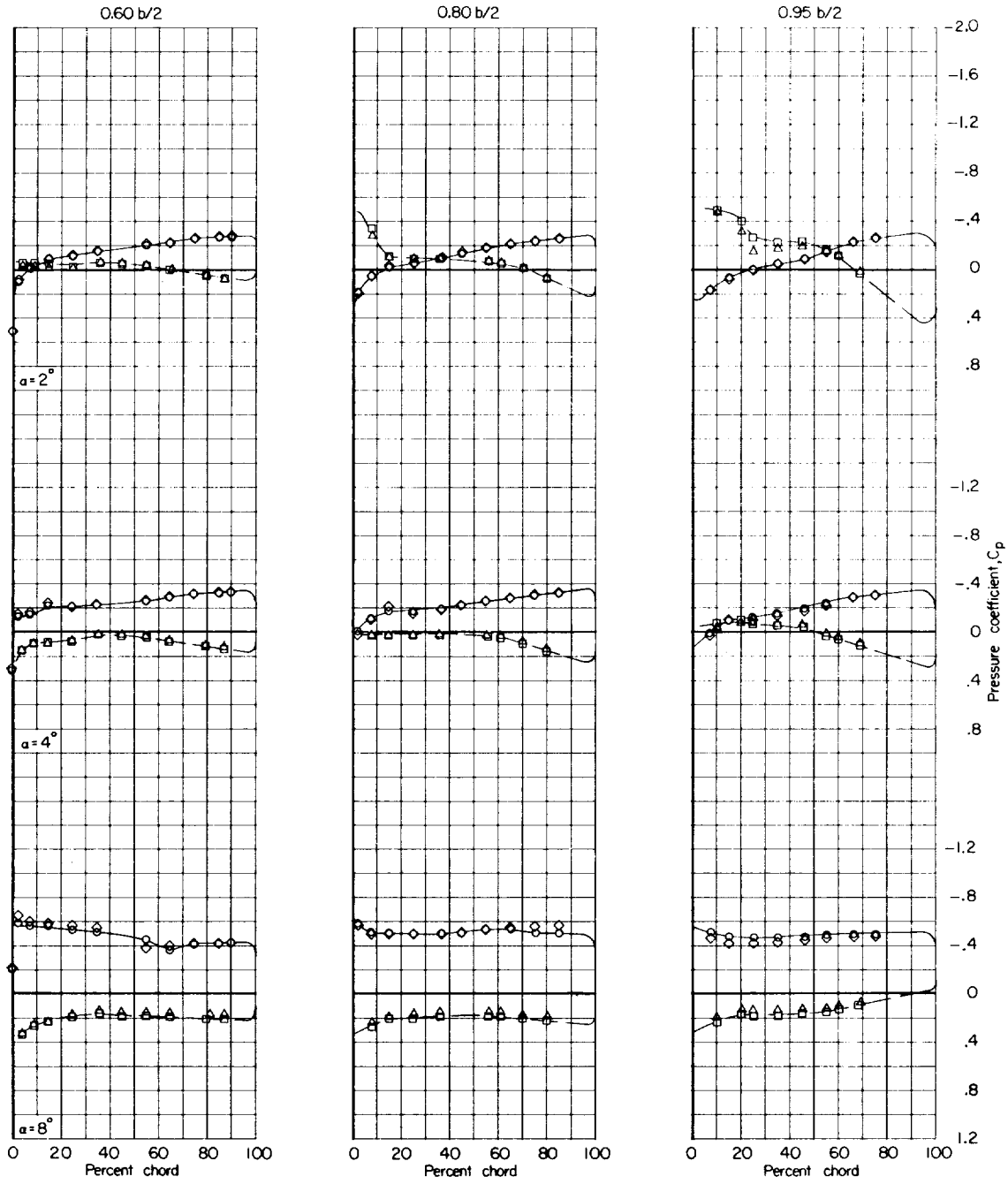
(s) Concluded.

Figure 4.- Continued.



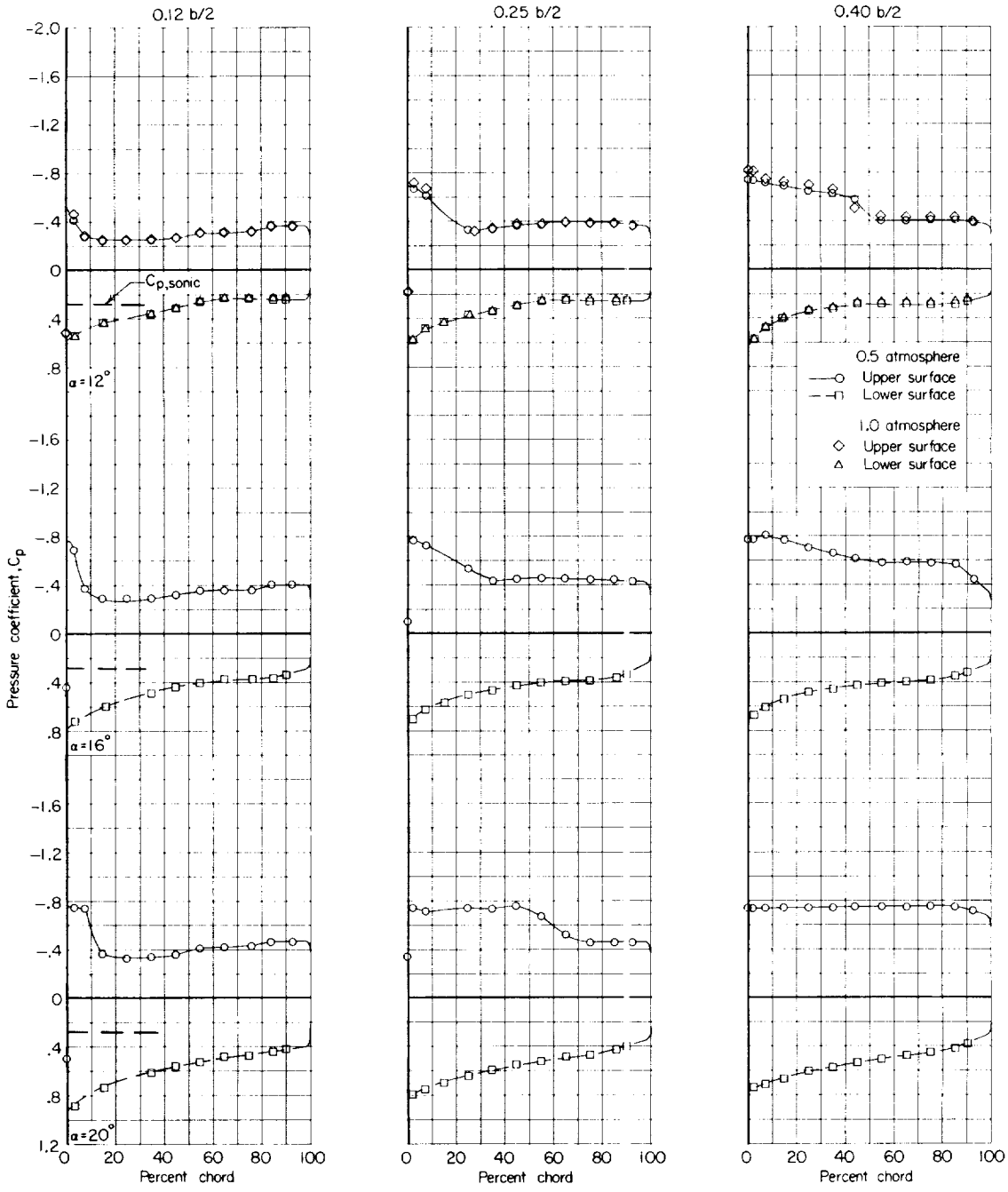
(t) M = 1.200; $\alpha = 2^{\circ}$, 4°, and 8°.

Figure 4.- Continued.



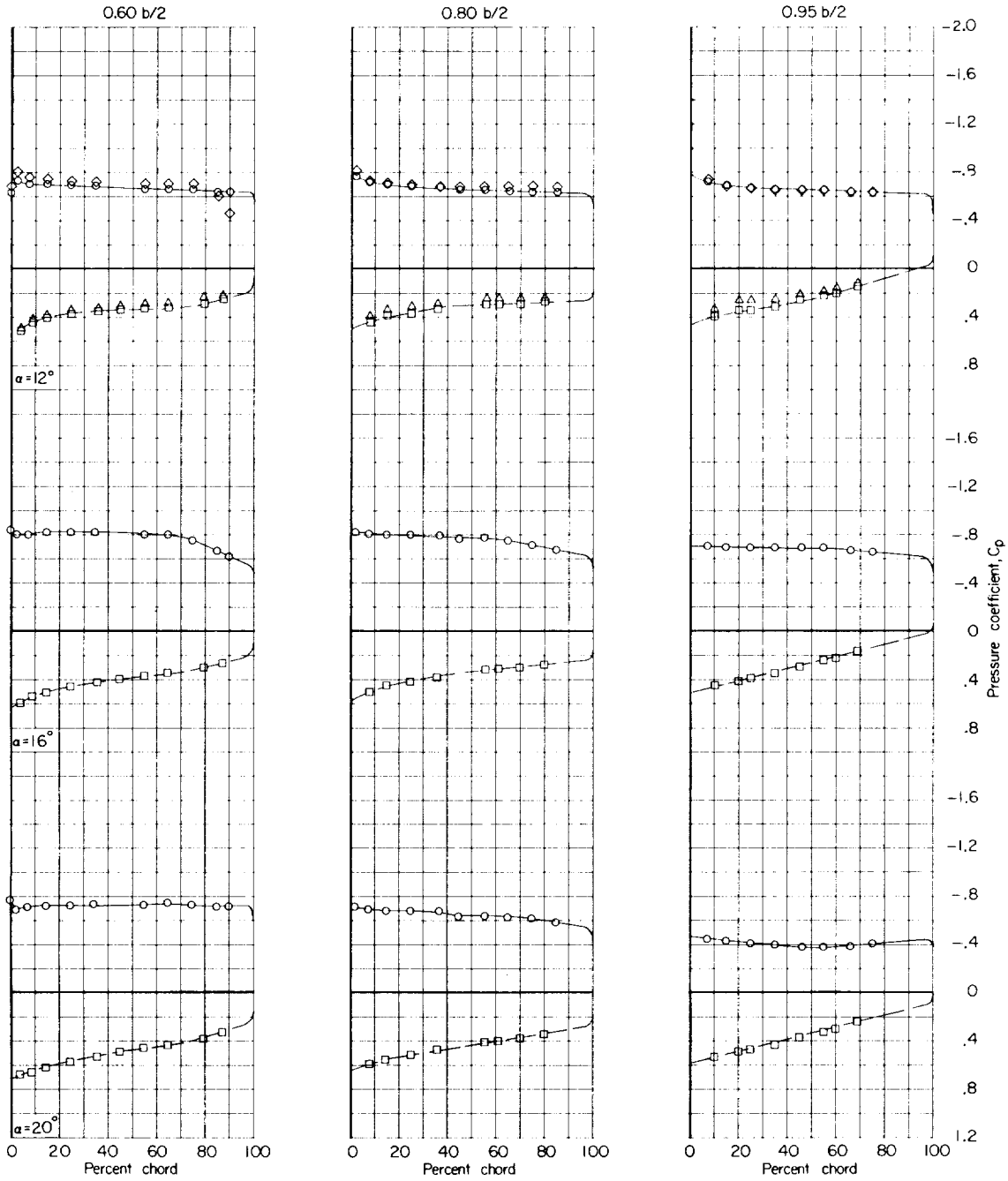
(t) Concluded.

Figure 4.- Continued.



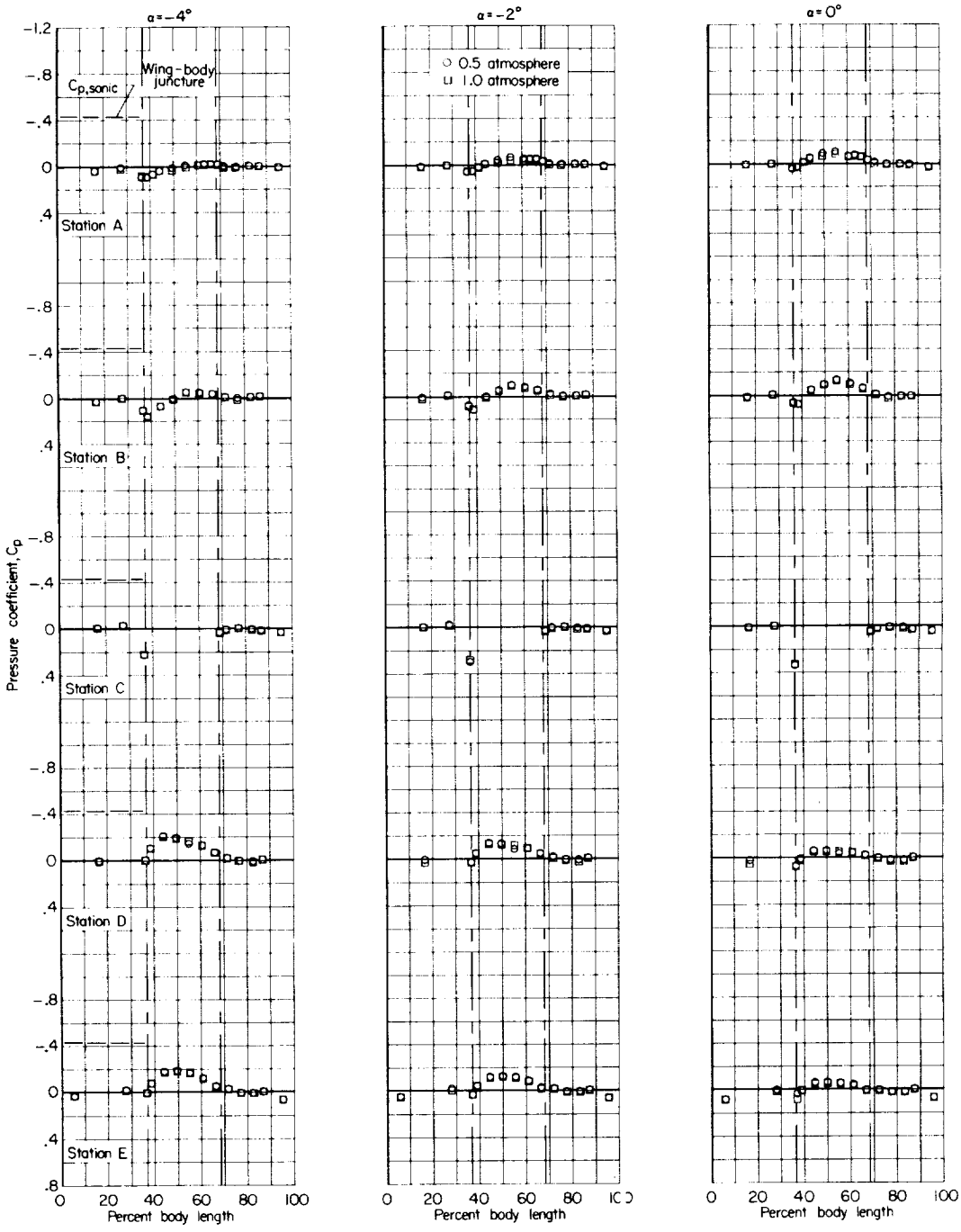
(u) M = 1.200; α = 12°, 16°, and 20°.

Figure 4.- Continued.



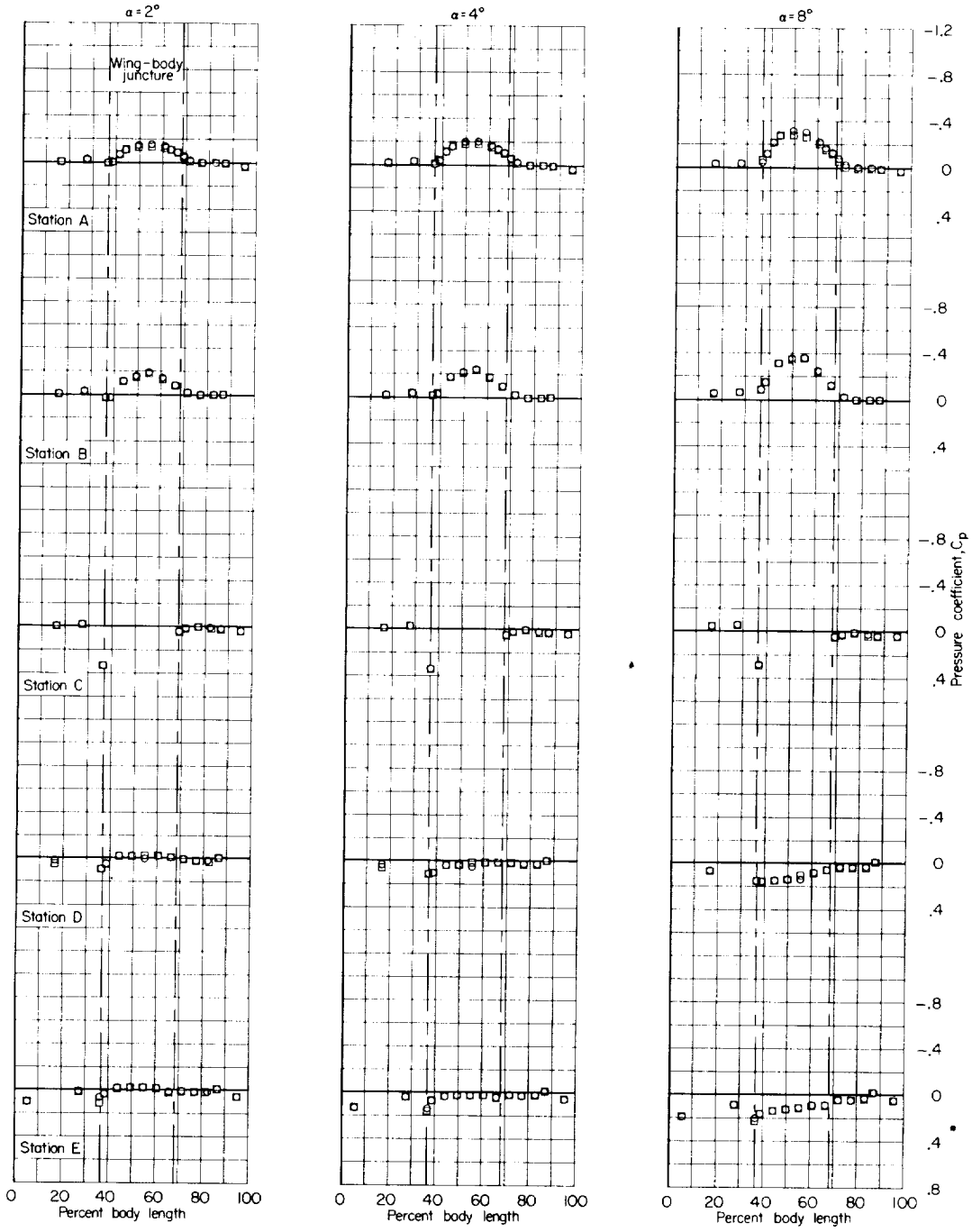
(u) Concluded.

Figure 4.- Concluded.



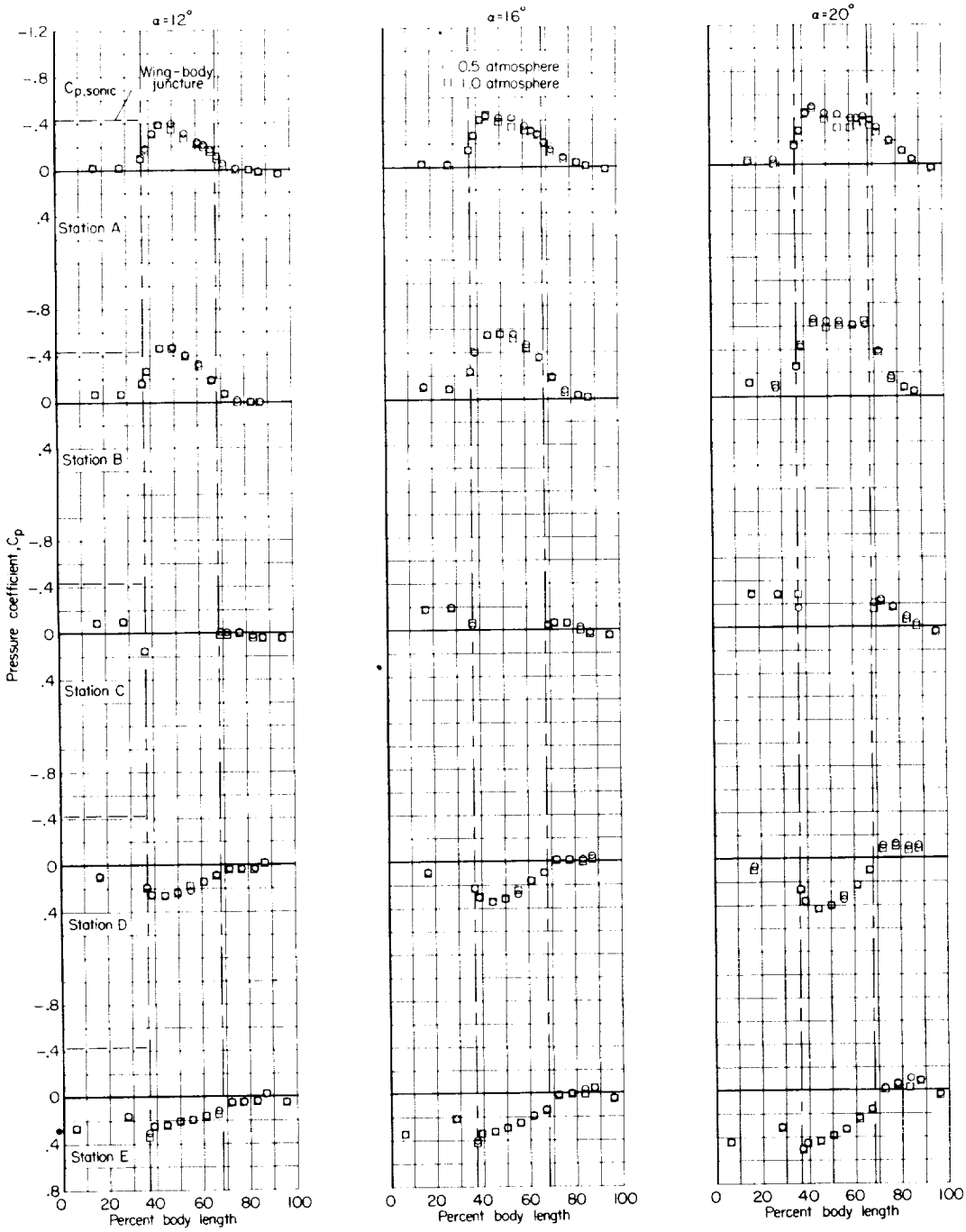
(a) M = 0.800.

Figure 5.- Pressure measurements on the body in the presence of the wing.



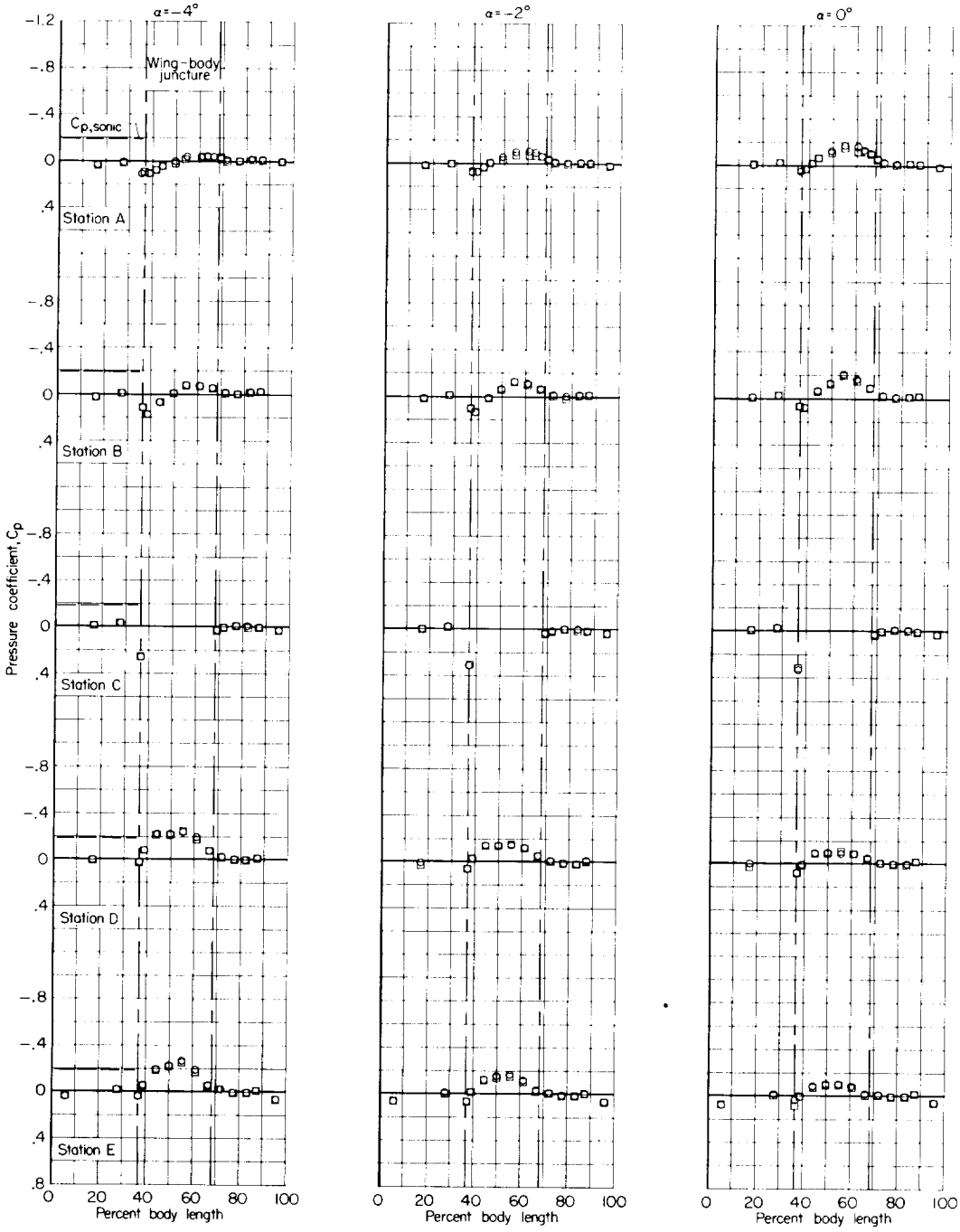
(a) Continued.

Figure 5.- Continued.



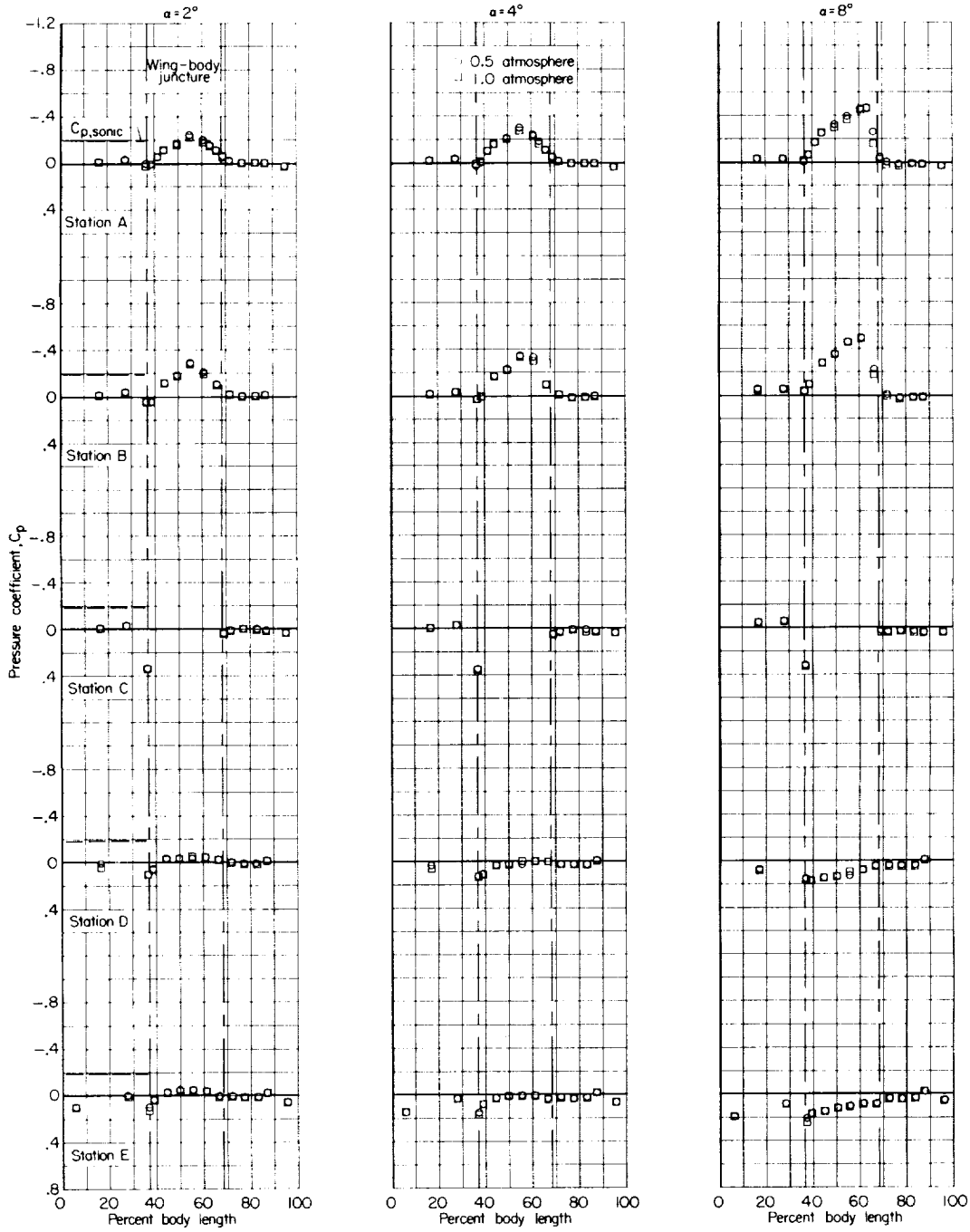
(a) Concluded.

Figure 5.- Continued.



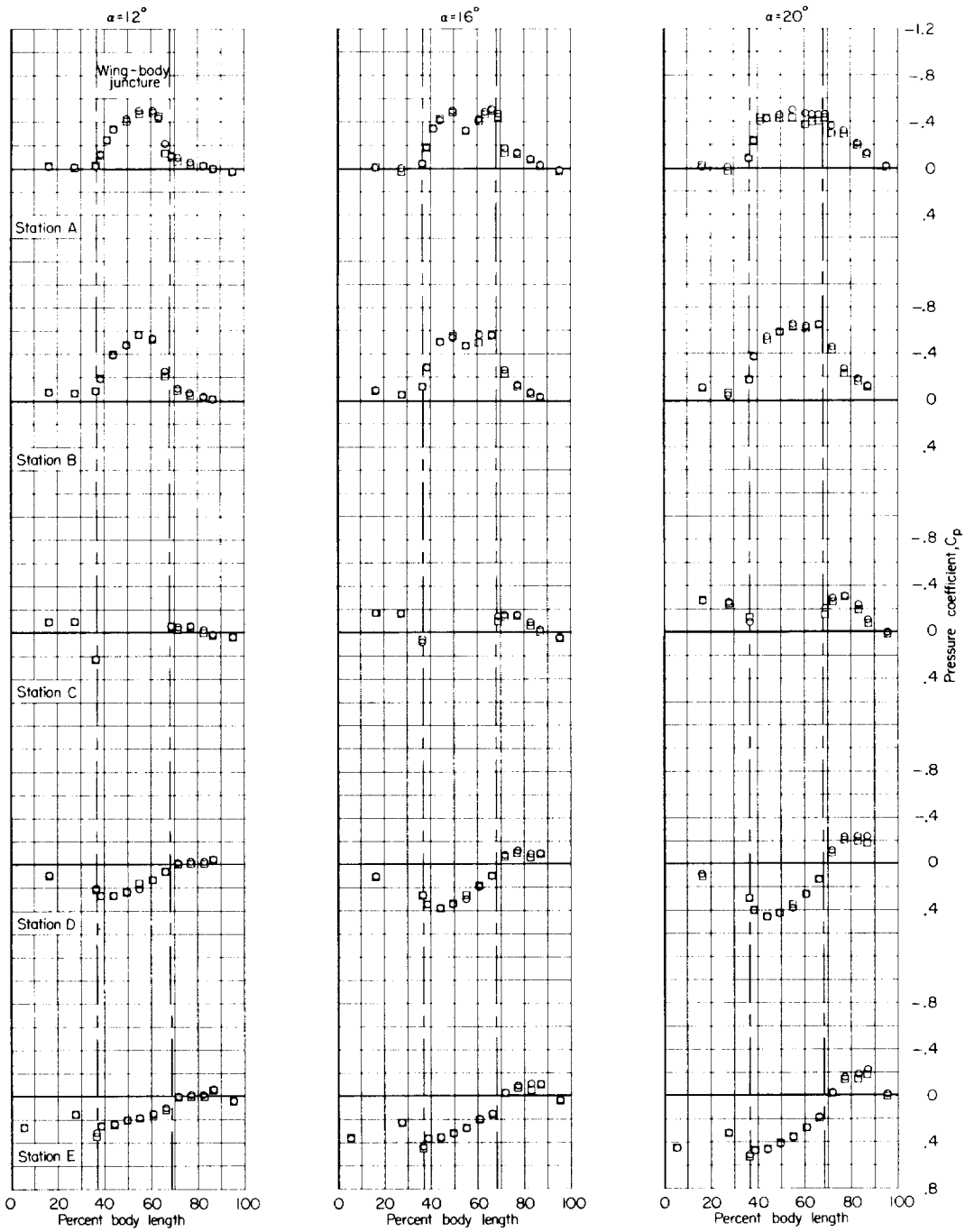
(b) M = 0.900.

Figure 5.- Continued.



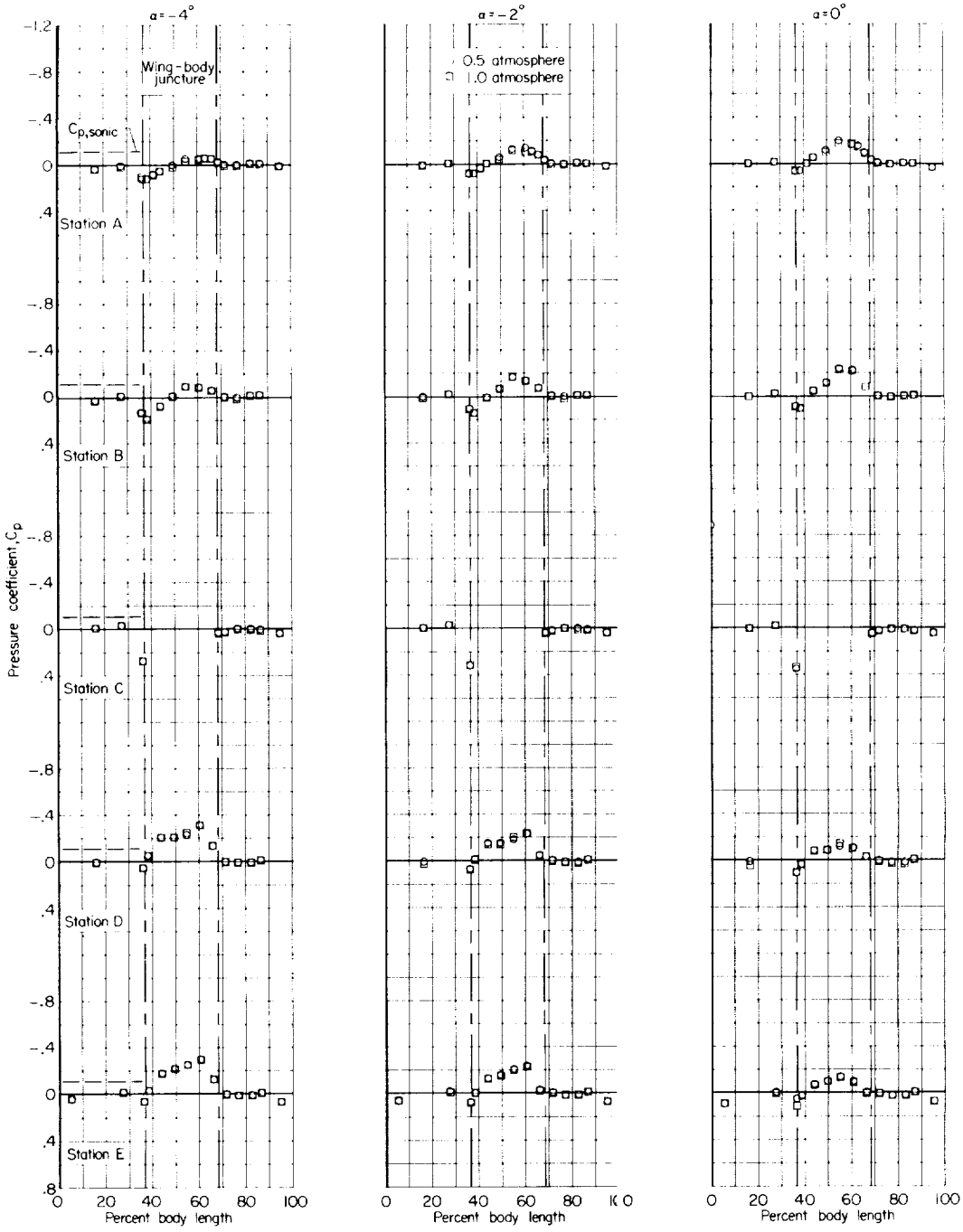
(b) Continued.

Figure 5.- Continued.



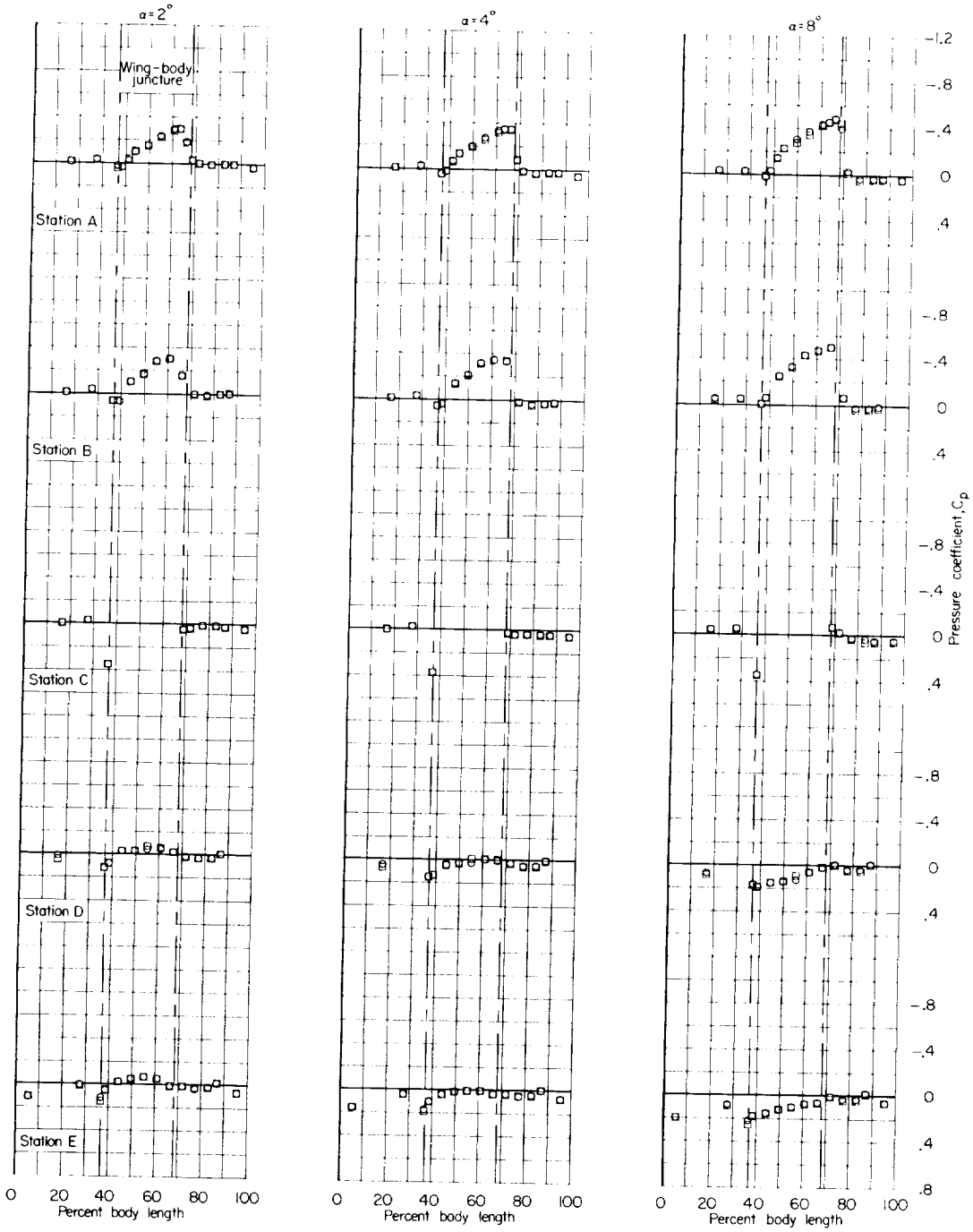
(b) Concluded.

Figure 5.- Continued.



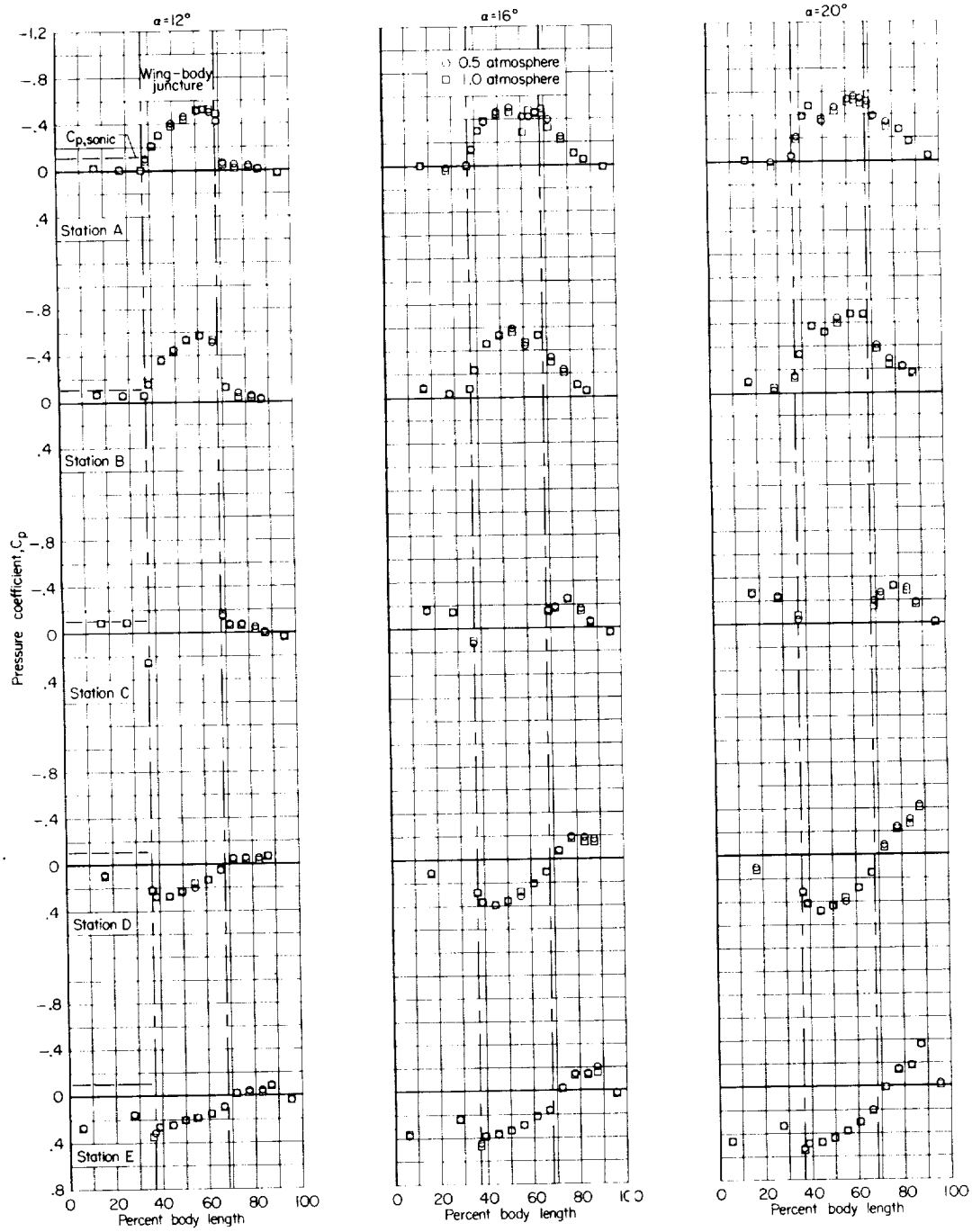
(c) M = 0.940

Figure 5.- Continued.



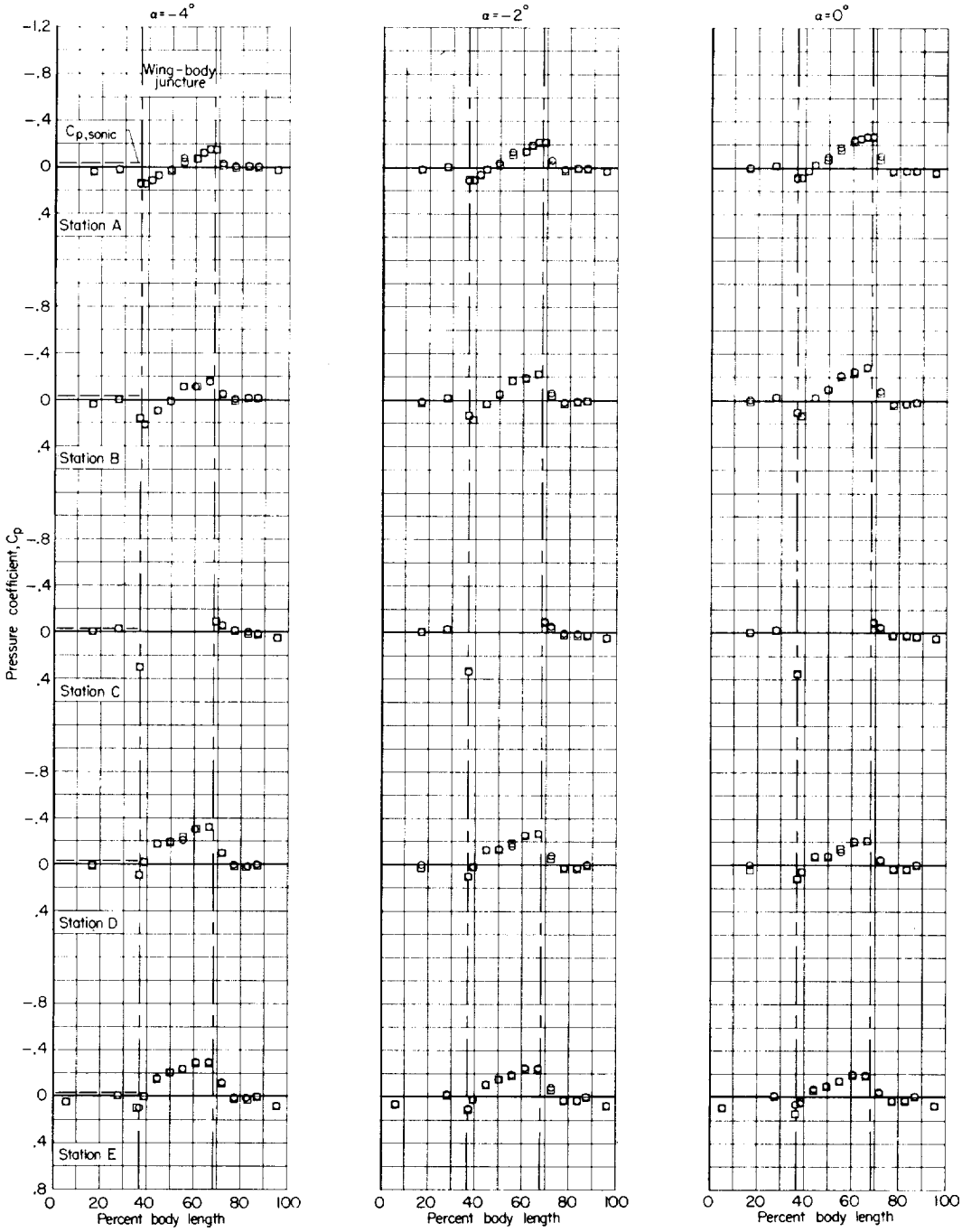
(c) Continued.

Figure 5.- Continued.



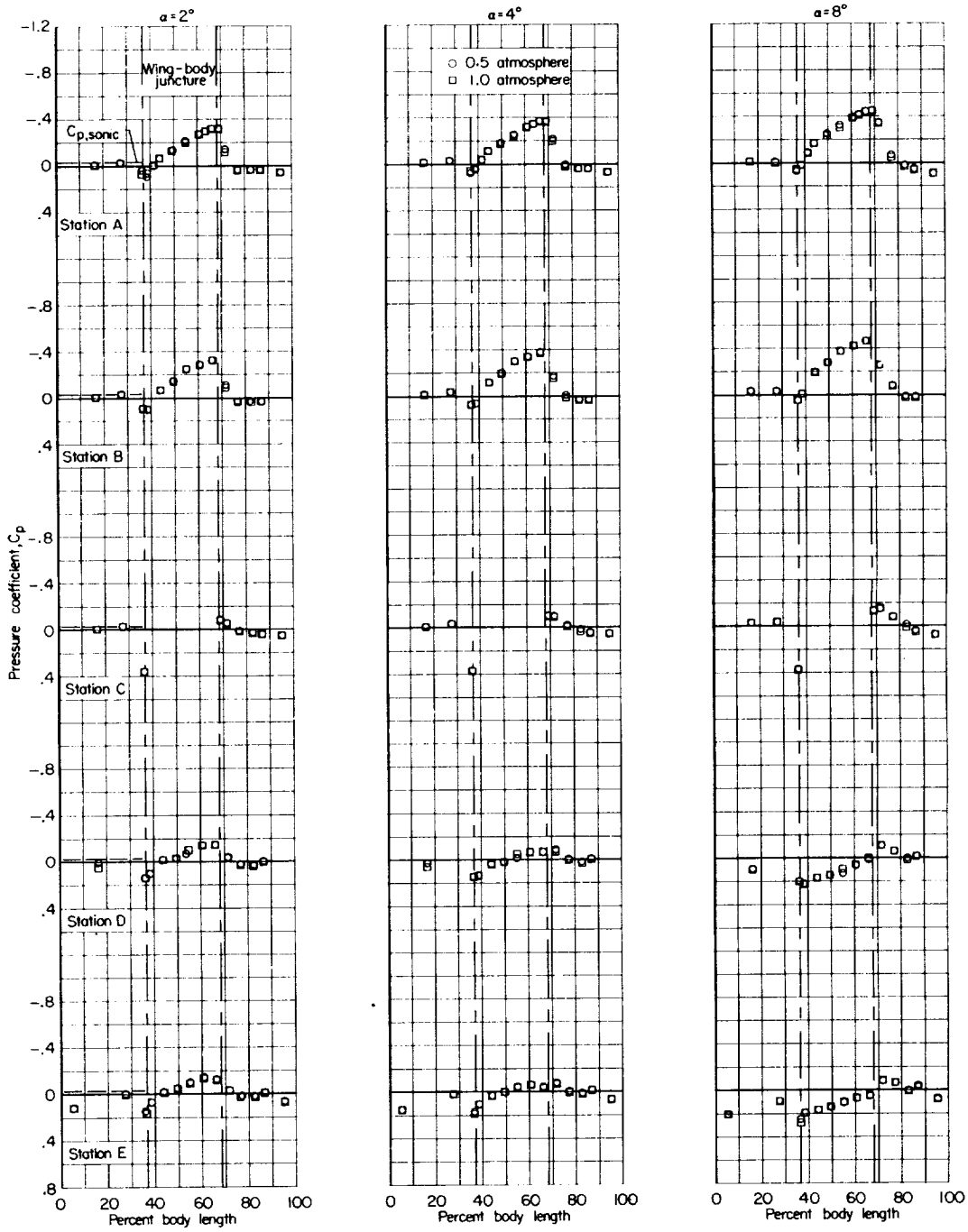
(c) Concluded.

Figure 5.- Continued.



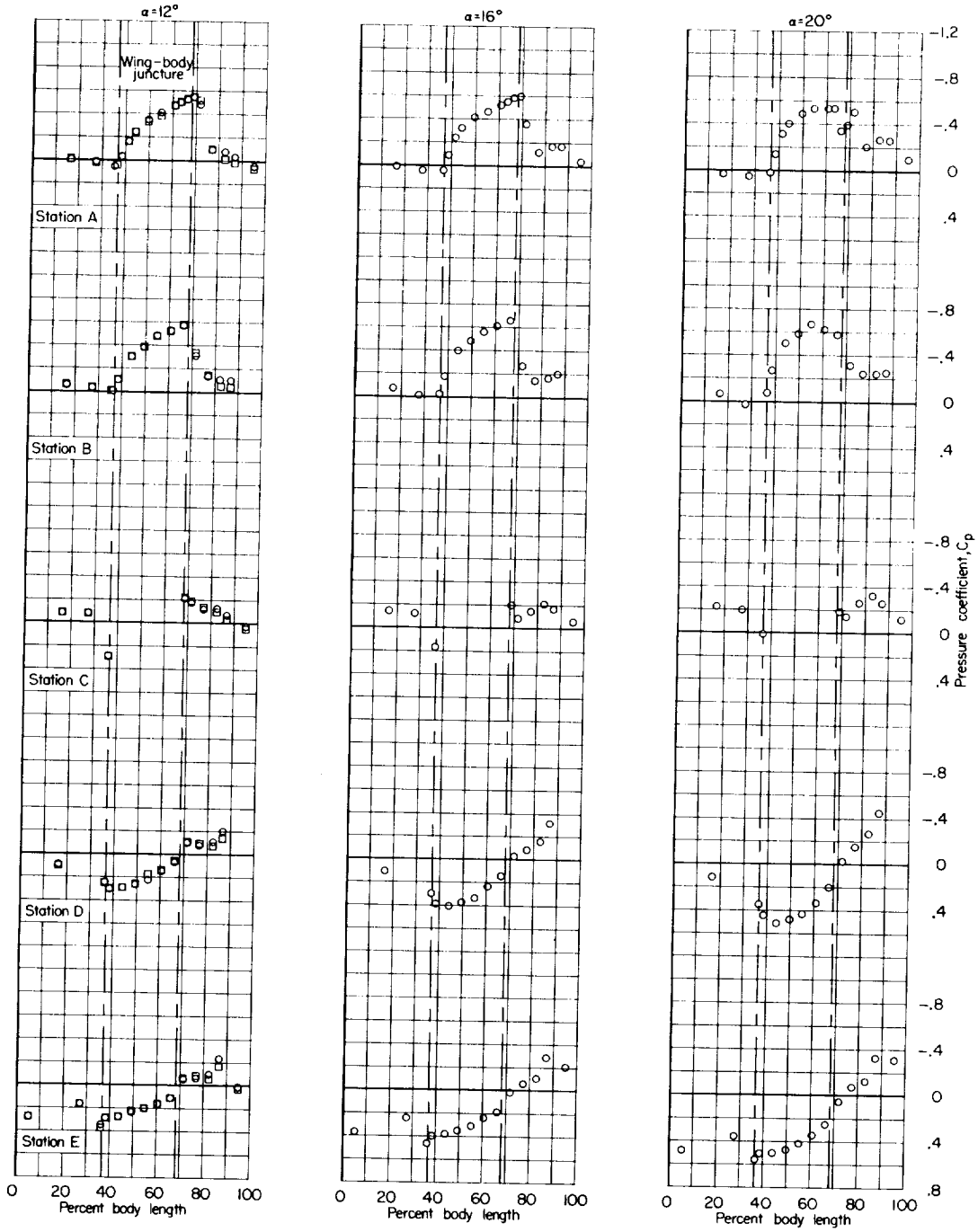
(d) M = 0.980.

Figure 5.- Continued.



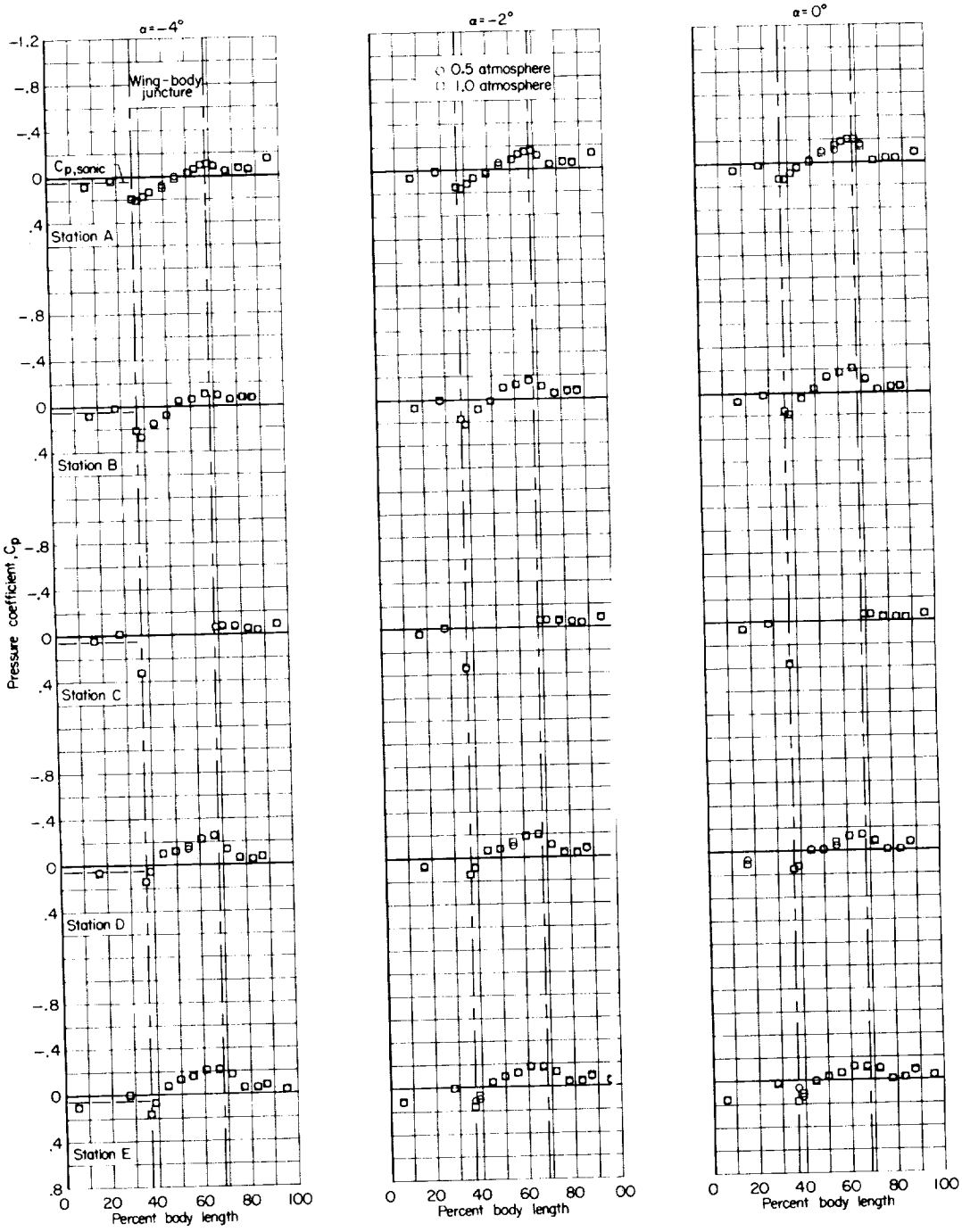
(d) Continued.

Figure 5.- Continued.



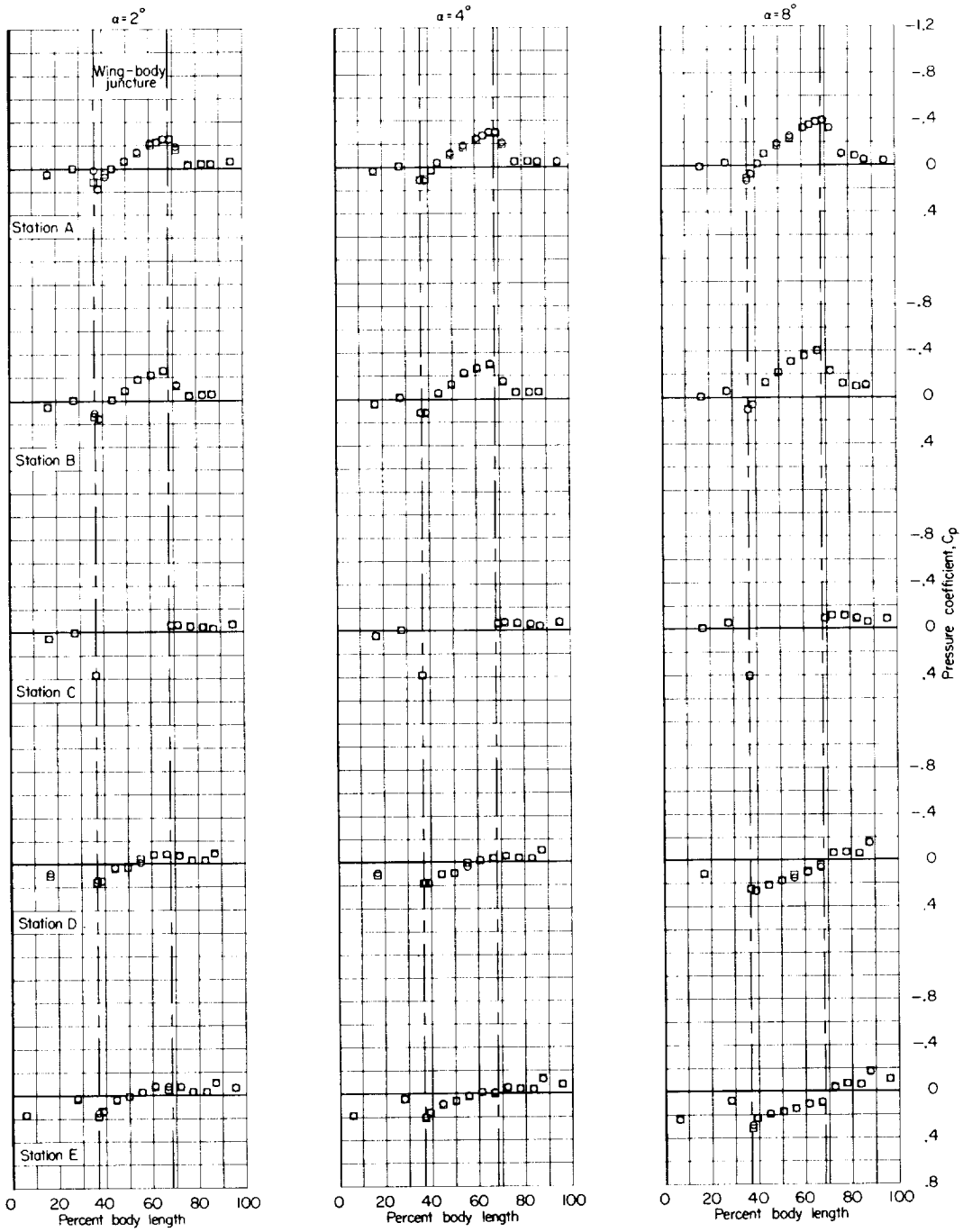
(d) Concluded.

Figure 5.- Continued.



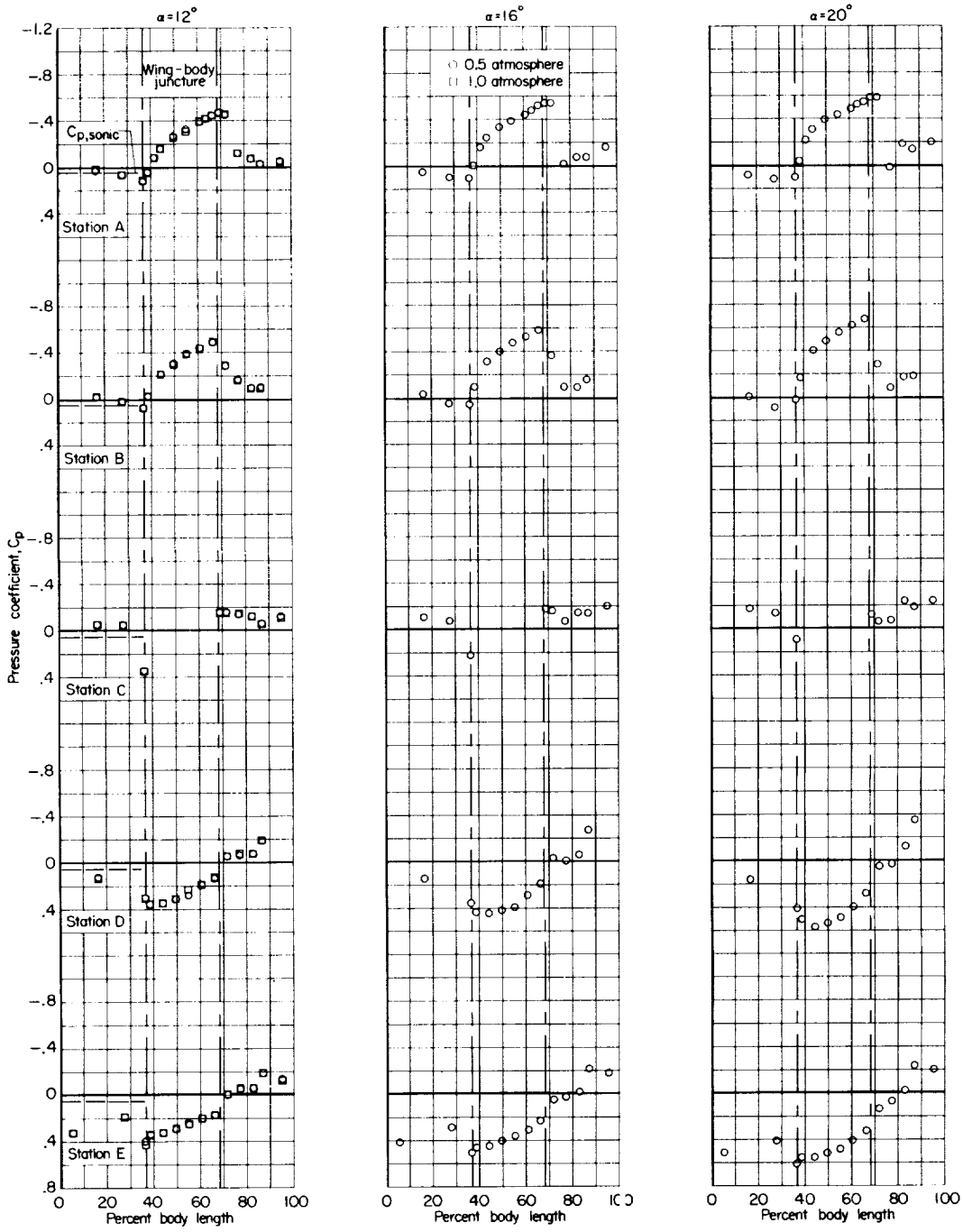
(e) M = 1.030.

Figure 5.- Continued.



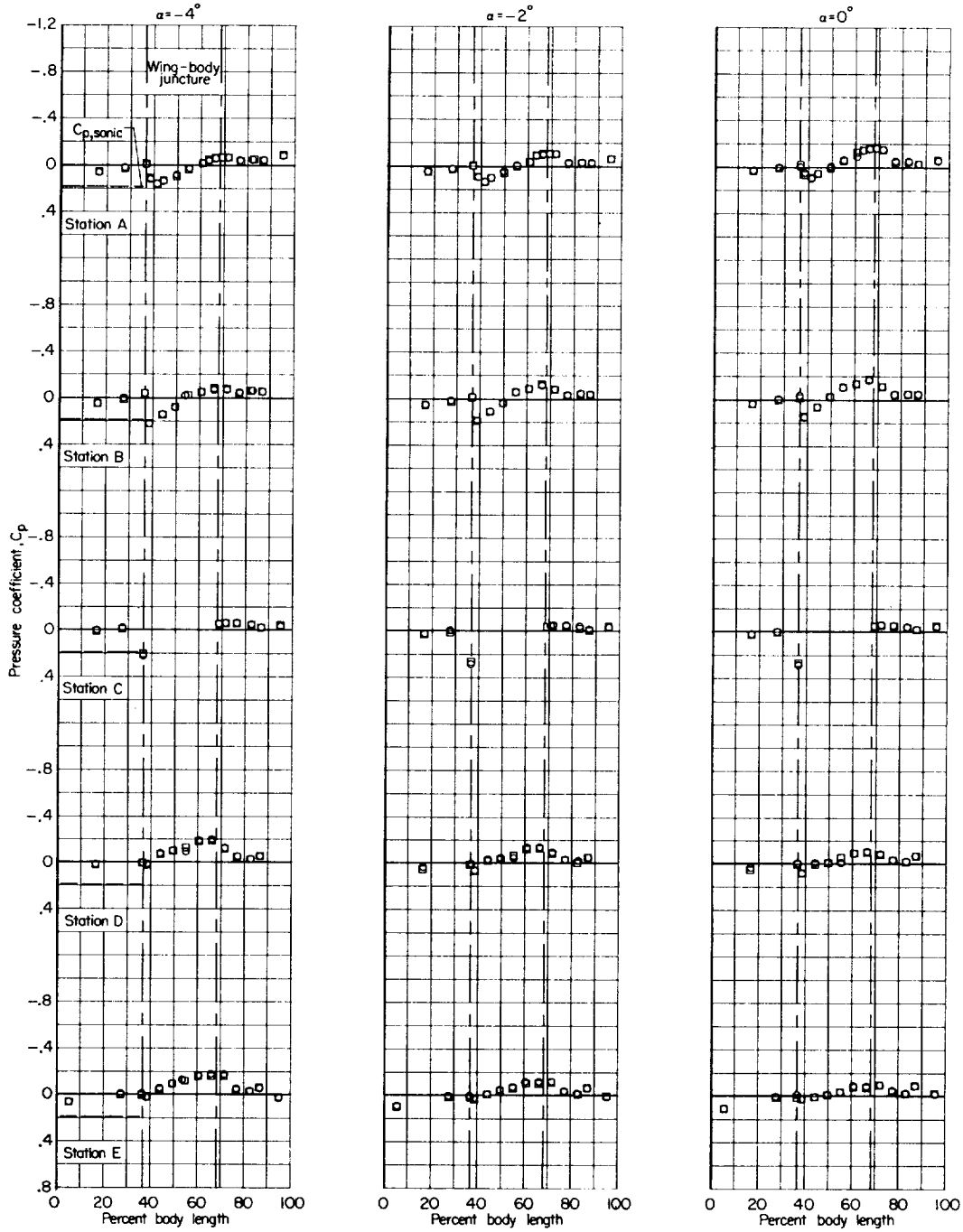
(e) Continued.

Figure 5.- Continued.



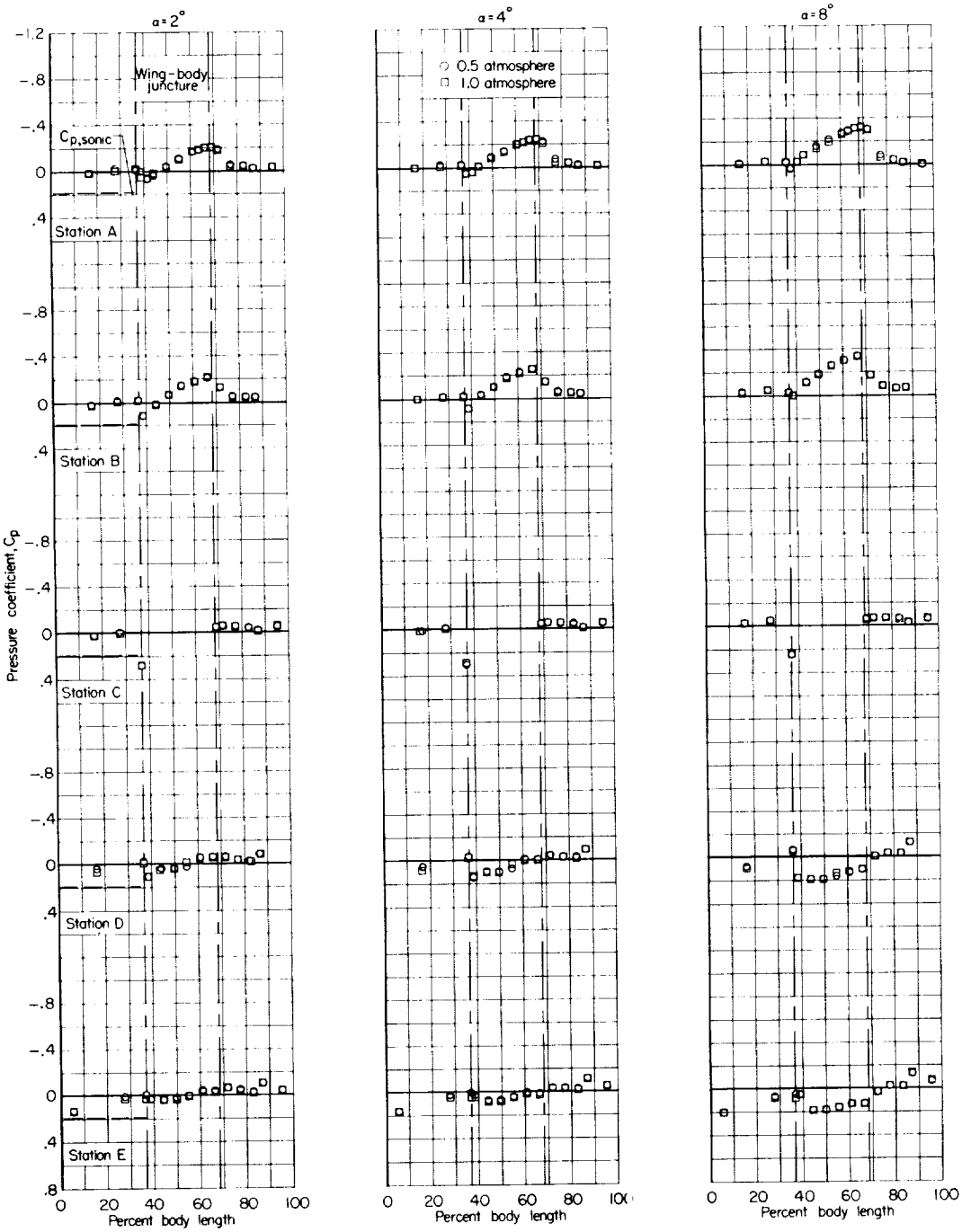
(e) Concluded.

Figure 5.- Continued.



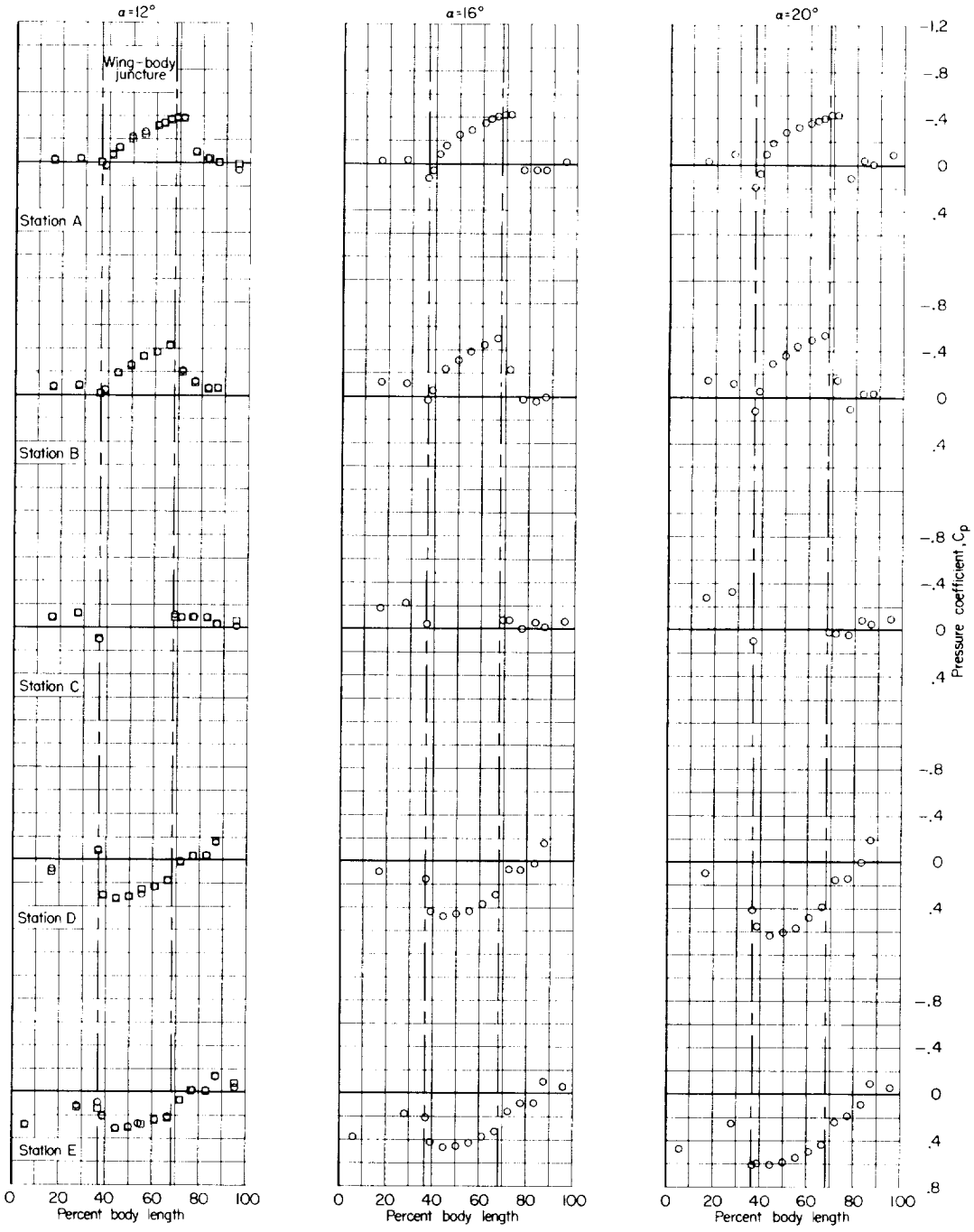
(f) M = 1.125.

Figure 5.- Continued.



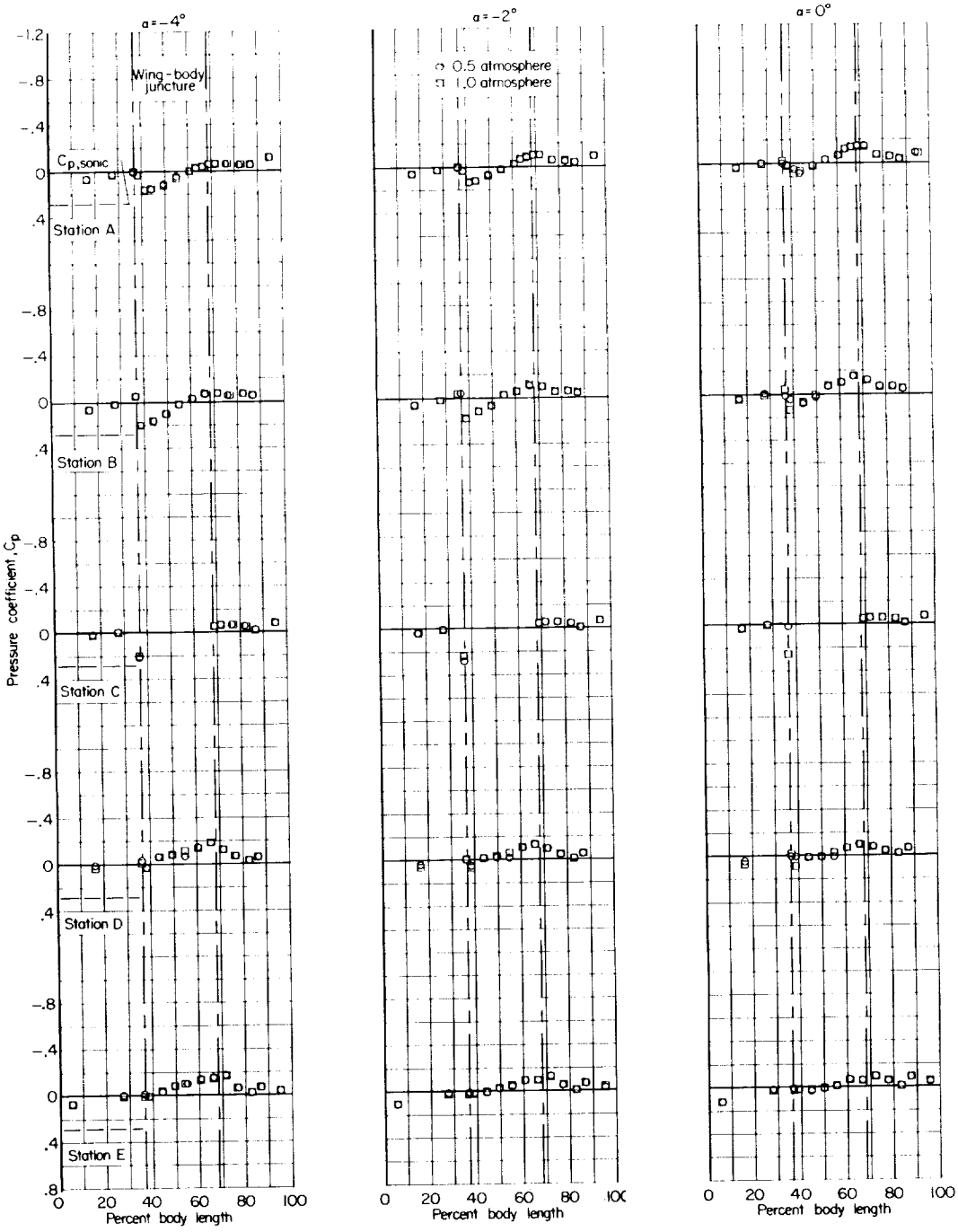
(f) Continued.

Figure 5.- Continued.



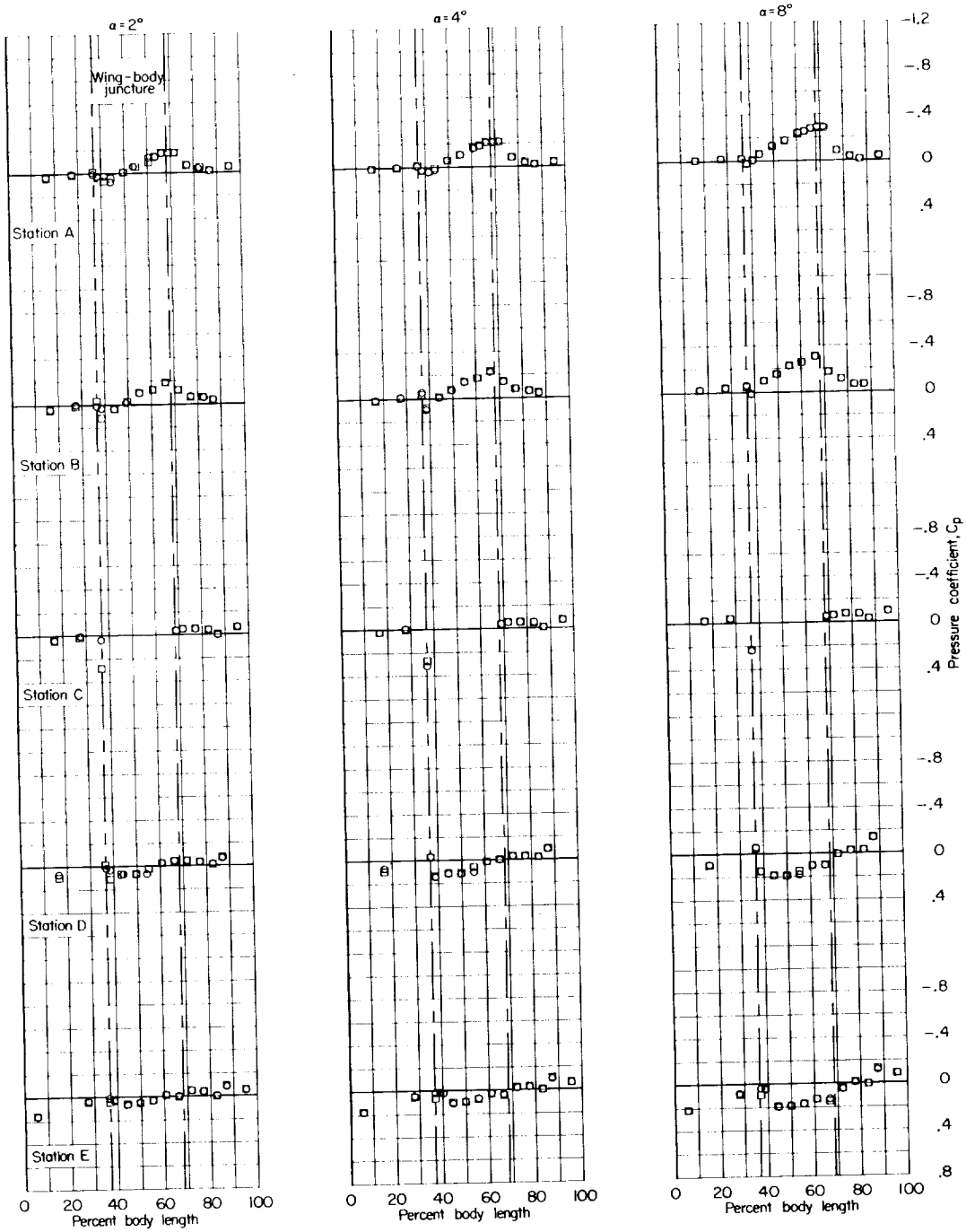
(f) Concluded.

Figure 5.- Continued.



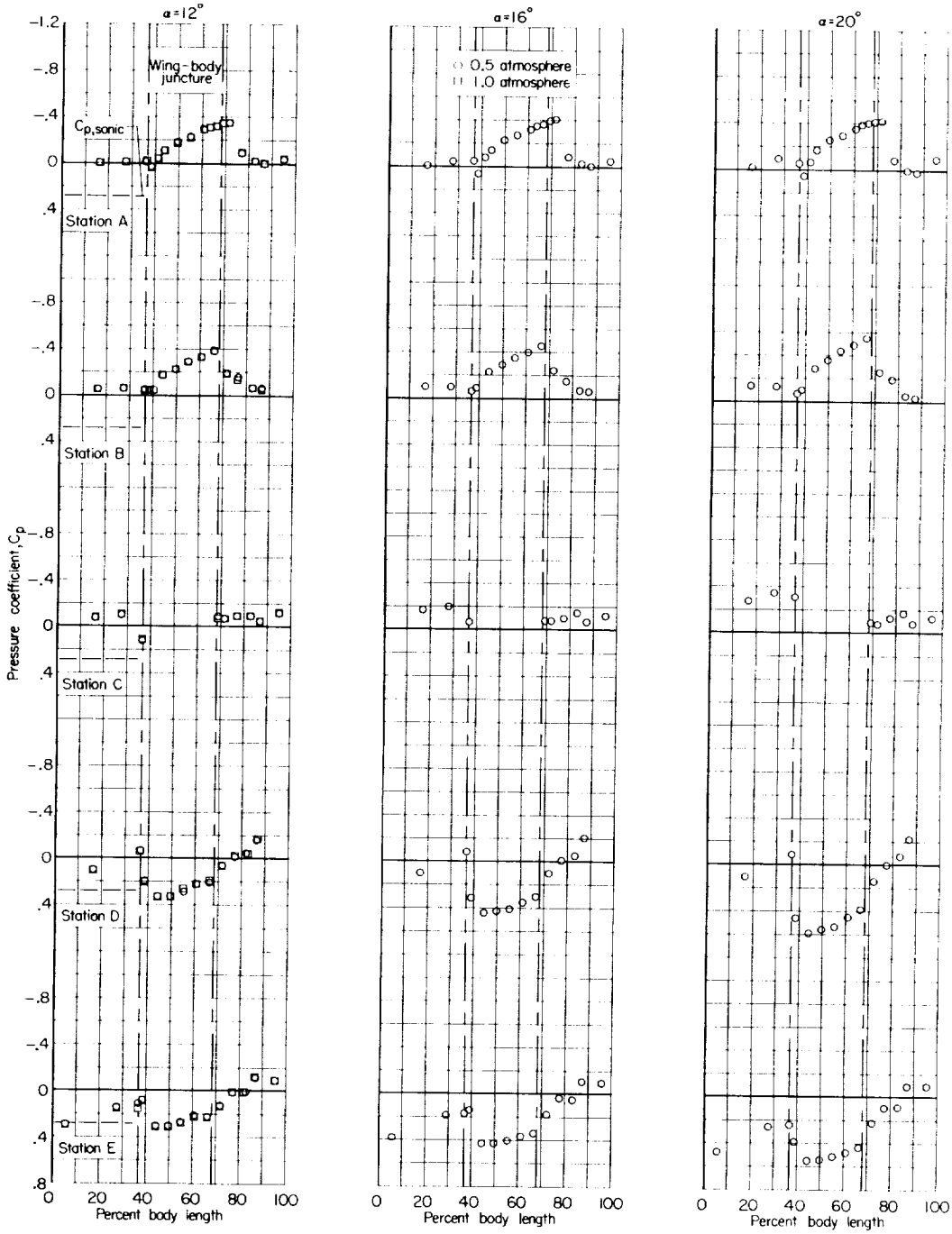
(g) M = 1.200.

Figure 5.- Continued.



(g) Continued.

Figure 5.- Continued.



(g) Concluded.

Figure 5.- Concluded.

NASA - Langley Field, Va. L-207